DUPLICATE

PILOT'S OPERATING HANDBOOK

PIPER CHEROKEE WARRIOR II

FAA APPROVED IN NORMAL AND UTILITY CATEGORIES BASED ON CAR 3 AND FAR PART 21, SUBPART J. THIS HANDBOOK INCLUDES THE MATERIAL REQUIRED TO BE FURNISHED TO THE PILOT BY CAR 3 AND FAR PART 21, SUBPART J AND CONSTITUTES THE APPROVED AIRPLANE FLIGHT MANUAL AND MUST BE CARRIED IN THE AIRPLANE AT ALL TIMES.

PA-28-161
REPORT: VB-880

FAA APPROVED BY: Ward Evans
WARD EVANS
D.O.A. NO. SO-1
PIPER AIRCRAFT CORPORATION
VERO BEACH, FLORIDA

AIRPLANE SERIAL NO. 28-7916139

AIRPLANE REGISTRATION NO. N405SB

DATE OF APPROVAL: DECEMBER 16, 1976

PIPER

WARNING

EXTREME CARE MUST BE EXERCISED TO LIMIT THE USE OF THIS MANUAL TO APPLICABLE AIRCRAFT. THIS MANUAL REVISED AS INDICATED BELOW OR SUBSEQUENTLY REVISED IS VALID FOR USE WITH THE AIRPLANE IDENTIFIED ON THE FACE OF THE TITLE PAGE WHEN OFFICIALLY APPROVED. SUBSEQUENT REVISIONS SUPPLIED BY PIPER AIRCRAFT CORPORATION MUST BE PROPERLY INSERTED.

MODEL PA-28-161, CHEROKEE WARRIOR II

PILOT'S OPERATING HANDBOOK, REPORT: VB-880 REVISION 17

PIPER AIRCRAFT CORPORATION
APPROVAL SIGNATURE AND STAMP Susan C Groenwoldt PA 57

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APPLICABILITY

Application of this handbook is limited to the specific Piper PA-28-161 model airplane designated by serial number and registration number on the face of the title page of this handbook.

This handbook cannot be used for operational purposes unless kept in a current status.

WARNING

INSPECTION, MAINTENANCE AND PARTS REQUIREMENTS FOR ALL NON-PIPER APPROVED STC INSTALLATIONS ARE NOT INCLUDED IN THIS HANDBOOK. WHEN A NON-PIPER APPROVED STC INSTALLATION IS INCORPORATED ON THE AIRPLANE, THOSE PORTIONS OF THE AIRPLANE AFFECTED BY THE INSTALLATION MUST BE INSPECTED IN ACCORDANCE WITH THE INSPECTION PROGRAM PUBLISHED BY THE OWNER OF THE STC. SINCE NON-PIPER APPROVED STC INSTALLATIONS MAY CHANGE SYSTEMS INTERFACE, OPERATING CHARACTERISTICS AND COMPONENT LOADS OR STRESSES ON ADJACENT STRUCTURES, PIPER PROVIDED INSPECTION CRITERIA MAY NOT BE VALID FOR AIRPLANES WITH NON-PIPER APPROVED STC INSTALLATIONS.

REVISIONS

The information compiled in the Pilot's Operating Handbook will be kept current by revisions distributed to the airplane owners.

Revision material will consist of information necessary to update the text of the present handbook and/or to add information to cover added airplane equipment.

I. Revisions

Revisions will be distributed whenever necessary as complete page replacements or additions and shall be inserted into the handbook in accordance with the instructions given below:

1. Revision pages will replace only pages with the same page number.
2. Insert all additional pages in proper numerical order within each section.
3. Page numbers followed by a small letter shall be inserted in direct sequence with the same common numbered page.

II. Identification of Revised Material

Revised text and illustrations shall be indicated by a black vertical line along the outside margin of the page, opposite revised, added or deleted material. A line along the outside margin of the page opposite the page number will indicate that an entire page was added.

Black lines will indicate only current revisions with changes and additions to or deletions of existing text and illustrations. Changes in capitalization, spelling, punctuation or the physical location of material on a page will not be identified by symbols.

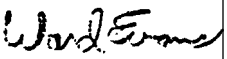
ORIGINAL PAGES ISSUED

The original pages issued for this handbook prior to revision are given below:

Title, ii through v, 1-1 through 1-14, 2-1 through 2-8, 3-1 through 3-12, 4-1 through 4-16, 5-1 through 5-26, 6-1 through 6-56, 7-1 through 7-24, 8-1 through 8-16, 9-1 through 9-10, 10-1 through 10-2.

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS

Current Revisions to the PA-28-161 Cherokee Warrior II Pilot's Operating Handbook, REPORT:-VB-880 issued December 16, 1976.

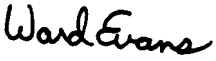
Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 1 - 761 649 (PR770224)	1-6	Corrected to "Meteorological."	 Ward Evans Feb. 24, 1977
	2-5	Revised 2.23.	
	3-11	Revised NOTE.	
	4-4	Revised Starting Engine When Hot procedure.	
	4-9	Revised 4.13 (b).	
	6-4	Revised Leveling Diagram illustration.	
	6-53	Revised Dwg. Nos. of items 277 and 279.	
	7-24	Added ELT test info.	
Rev. 2 - 761 649 (PR770711)	1-11, 1-12, 1-13, 1-14	Revised para. 1.21, Conversion Factors.	
	2-7	In 2.25, revised Takeoff and Landing Checklists to include air cond. info.; added air conditioning warning placard:	
	4-4	Added air cond. check to Ground Check.	
	4-5	Added air cond. directions to Before Takeoff and to Approach and Landing.	
	4-6	Added air cond. directions to Stopping Engine.	
	4-11	Added air cond. check to 4.19, Ground Check.	
	4-12	Added air cond. directions to 4.21, Before Takeoff.	
	4-14	Added air cond. directions to 4.29, Approach and Landing, and 4.31, Stopping Engine.	
	5-3	Revised Fig. nos. in item (b).	
	5-4	Revised Fig. nos. in footnote	
	5-5	Revised Fig. nos. in item (e).	
	5-9	Revised Fig. Nos.; revised Page Nos.; revised titles; added pages; added figures.	
	5-13	Relocated Normal Short Field Takeoff Dist. to 5-14; added new chart.	
	5-14	Relocated Obs. Clearance Short Field Takeoff Dist. to 5-16; added revised chart relocated from 5-13.	
	5-15	Relocated Eng. Perf. to 5-17; added new chart.	
	5-16	Relocated Climb Perf. to 5-19; added revised chart relocated from 5-14.	
	5-17	Relocated Fuel, Time and Dist. to Climb to 5-20; added chart relocated from 5-15; added ser. nos.	
	5-18	Relocated Best Power Cruise Perf. to 5-21; added new chart.	

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 2 - 761 649 (PR770711) (cont)	5-19	Relocated Best Econ. Cruise Perf. to 5-23; added chart relocated from 5-16.	
	5-20	Relocated Best Power Mix. Range to 5-25; added chart relocated from 5-17.	
	5-21	Relocated Best Econ. Mix. Range to 5-27; added chart relocated from 5-18; added ser. nos.	
	5-22	Relocated Endurance to 5-29; added new chart.	
	5-23	Relocated Fuel, Time and Dist. to Descend to 5-30; added chart relocated from 5-19; added ser. nos.	
	5-24	Relocated Glide Perf. to 5-31; added new chart.	
	5-25	Relocated Landing Perf. to 5-32; added chart relocated from 5-20; added ser. nos.	
	5-26	Added new chart.	
	5-27	Added page; added chart relocated from 5-21; added ser. nos.	
	5-28	Added page (new chart).	
	5-29	Added page (chart relocated from 5-22).	
	5-30	Added page (chart relocated from 5-23).	
	5-31	Added page (chart relocated from 5-24).	
	5-32	Added page (chart relocated from 5-25).	
	6-33	Added items 67 and 69.	
	6-45	Revised item nos.; added item 213.	
	6-46	Revised item nos.; added items 229 through 235; relocated items to following page; added footnote.	
	6-47	Revised item nos.; added items 237 and 253; added relocated items; relocated items to following page; removed footnote.	
	6-48	Added items 259 and 263; added relocated items; added footnote.	
	6-49	Revised item nos.; revised items 273 and 275; added item 281; deleted footnote.	
	6-53	Revised item nos.; added item 325; revised item 329.	
	6-54	Revised item nos.	
	7-i	Added 7.37 to contents.	
	7-17	Added climate control panel to Fig. 7-15.	
	7-21	Revised air blower info. in 7.23.	
	7-24	Added 7.37, Air Conditioning; added footnote.	
	7-25, 7-26	Added pages.	
	9-i	Added Supplement 4 to contents.	

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 2 - 761 649 (PR770711) (cont)	9-3	Added STC No. to Section 1 - General; in Section 2 revised (a); in Section 3 revised (a), deleted (c) and renumbered accordingly, revised (c) and (d); added Preflight to Section 4 and relocated material.	
	9-4	Added relocated material; added Inflight heading; revised (a) through (d) and added (e).	
	9-5	Added STC No. to Section 1 - General; in Section 2 revised (a); in Section 3 revised (c) and (d); added (e) and NSD 360 procedures; relocated Section 4.	
	9-6	Added NSD 360 and NSD 360 A procedures; added relocated material; in Section 4 revised (a) (1), added (b) Radio Coupling heading and renumbered (3) and (4) to (b) (1) and (2); relocated material.	
	9-7	Added relocated material; completely revised sections (d) and (e).	Ward Evans
	9-8	Removed item (g).	
	9-11, 9-12, 9 13, 9-14	Added pages (Supplement 4 - Air Conditioning Installation).	Ward Evans July 11, 1977
Rev. 3 - 761 649 (PR780630)	1-4	Revised items 1.13 and 1.17.	
	2-7	Added new takeoff checklist; revised lead sentence.	
	3-3, 3-7	Added info. to Engine Power Loss During Takeoff.	
	4-5	Added info. to Before Takeoff	
	4-9	Revised items 4.13 (a), (b) and (c).	
	4-12	Added info. to Before Takeoff.	
	4-13.	Corrected info. under 4.23, Takeoff.	
	5-5	Revised item 5.5 (e).	
	5-19	Added Note to Figure 5-11.	
	5-25	Added Note to Figure 5-19.	
	5-26	Added Note to Figure 5-20.	
	5-27	Added Note to Figure 5-21.	
	5-28	Added Note to Figure 5-22.	
	6-1	Revised info. under item 6.1.	
	6-9, 6-10	Revised Figure No.	
	6-35	Revised item 93; added item 94.	
	6-41	Revised items 161 and 163.	
	6-43	Revised and added info. to items 179 and 181; relocated info. to pg. 6-44.	
	6-44	Added info. from pg. 6-43; added new items 184 and 185; existing item 185 changed to 186; re-	

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 3-761649 (PR780630) (cont.)	6-45	Added info. from pg. 6-44; revised item 201; relocated info. to pg. 6-46.	
	6-46	Added info. from pg. 6-45; relocated info. to pg. 6-47.	
	6-47	Added info. from pg. 6-46; revised item 235; relocated info. to pg. 6-48.	
	6-48	Added info. from pg. 6-47; revised footnotes.	
	6-49	Revised items 273 and 275.	
	6-54	Revised item 351.	
	7-13	Revised Para. 7.15 info.	
	7-18	Revised Para. 7.21 info.	
	7-21	Added Caution to para. 7.23.	
	7-23	Revised para. 7.35.	
	8-12	Changed 8.21 (d) to (c).	
	8-13	Changed 8.21 (e) to (d).	
	8-i	Revised 8.29.	Ward Evans
	8-16	Revised para. 8.29.	June 30, 1978
Rev. 4 - 761 649 (PR790226)	1-6	Revised para. 1.19.	
	1-12	Revised para. 1.21.	
	1-13	Revised para. 1.21.	
	6-1	Revised para. 6.1.	
	6-19	Revised item 9.	
	6-33	Added items 71 and 73.	
	6-37	Revised item 113.	
	6-41	Added item 170.	
	7-12	Added Warning to para. 7.15.	
	7-13	Revised para. 7. 15.	Ward Evans
	7-23	Revised para. 7.35.	Feb. 26, 1979
Rev. 5 - 761 649 (PR790413)	6-49	Revised item 277; added item 278.	
	7-23	Revised para. 7.35.	
	7-24	Revised para. 7.35; relocated para. 7.37 to pg. 7-25.	Ward Evans
	7-25	Added info. from pg. 7-24.	April 13, 1979
Rev. 6 - 761 649 (PR790703)	1-3	Added new engine designation.	
	1-7	Revised item 1.19 (e).	
	2-2	Added new engine designation.	
	2-3	Revised para. 2.11.	
	2-4	Revised para. 2.13.	
	2-7, 2-8	Revised para. 2.25 (revised existing placards, added new placards).	
	3-i	Added para 3.28, Carb. Icing.	
	3-4	Added Carb. Icing.	
	3-11	Added para. 3.28, Carb. Icing.	
	4-i	Added-para. 4.28, Descent; revised pg. no.	

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 6 - 761 649 (PR790703) (cont.)	4-4	Revised Before Starting Engine.	
	4-5	Added Descent; relocated Approach and Landing to pg. 4-6.	
	4-6	Added Approach and Landing from pg. 4-5.	
	4-7	Revised para. 4.9.	
	4-8	Revised para. 4.11.	
	4-14	Added para. 4.28, Descent; relocated para. 4.31 to pg. 4-15.	
	4-15	Added para. 4.31 from pg. 4-14.	
	5-3	Revised item 5.5 (a).	
	5-4	Revised item 5.5 (b) and (c).	
	5-5	Revised items 5.5 (d) and (e).	
	5-6	Revised items 5.5 (f) and (g).	
	5-20	Revised Figure 5-13.	
	5-25	Revised Figure 5-19.	
	5-26	Revised Figure 5-20.	
	5-27	Revised Figure 5-21.	
	5-28	Revised Figure 5-22.	
	5-29	Revised Figure 5-23.	
	5-30	Revised Figure 5-25.	
	5-31	Revised Figure 5-27.	
	6-1	Revised para. 6.1.	
	6-3	Added Caution to item 6.3 (a)(3).	
	6-11	Revised Figure 6-9.	
	6-12	Revised Figure 6-11.	
	6-14	Revised Figure 6-15.	
	6-17	Revised para. 6.9; added item 3; revised item 1.	
	6-19	Revised items 5, 7 and 9; added new items.	
	6-21	Revised item nos.; added new items.	
	6-23	Added new items.	
	6-25	Revised item nos.; added new items.	
	6-27	Revised items; added new items.	
	6-29, 6-33, 6-35, 6-36	Revised items.	
	6-37	Revised items; added new items.	
	6-38, 6-39	Revised items.	
	6-41, 6-42	Revised items; added item.	
	6-43, 6-44, 6-45, 6-46, 6-47, 6-48, 6-49, 6-53, 6-54	Revised items; added items.	
	7-3	Revised para. 7.5.	
	7-12	Revised para. 7.15 (added Warning).	
	7-16	Revised para. 7.19.	
	7-17	Revised Figure 7-15.	Ward Evans
	7-20	Revised Figure 7-19.	
	8-11	Revised para. 8.19.	Ward Evans
	10-1	Revised para. 10.3.	July 3, 1979

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 7 - 761 649 (PR800530)	1-3	Revised para 1.7.	
	1-4	Revised para. 1.11.	
	2-2	Revised para. 2.7	
	2-4	Revised para. 2.13.	
	2-9	Added page; added to para. 2.25.	
	4-3	Revised para. 4.5.	
	5-5	Revised para. 5.5(d).	
	5-26	Revised fig. 5.20.	
	6-7	Revised fig. 6-5.	
	6-14	Revised fig. 6- 15.	
	6-15	Added para. 6.8.	
	6-16	Added sample problem.	
	6-16a	Added page; cont. sample problem.	
	6-16b	Added page.	
	6-17	Corrected title.	
	6-27	Added to item 39.	
	6-29	Added item 47.	
	6-33	Revised item 67.	
	6-35	Added items 89 thru 91; renumbered items; moved items 94 thru 99 to pg. 6-36.	
	6-36	Relocated items 94 thru 99 from pg. 6-35.	
	6-39	Added item 145, renumbered items.	
	6-40	Added items 151 thru 159.	
	6-42	Added item 176; relocated item 177 from pg. 6-43.	
	6-43	Added item 178; moved item 177 to pg. 6-42.	
	6-44	Added items 196 and 198; renumbered items.	
	6-45	Added item 199; new item 201; renumbered items; moved item 213 to pg. 6-46.	
	6-46	Relocated item 213 from pg. 6-45.	
	6-53	Added item 342.	
	6-54	Added new items 344 and 345; renumbered items.	
	7-i	Added para. 7.39.	
	7-21	Revised para. 7.25.	
	7-26	Added para. 7.39.	
	8-i	Changed page no.	
	8-12, 8-13	Revised para. 8.21; moved para. 8.23 to pg. 8-14.	
	8-14	Relocated para. 8.23 from pg. 8-13; relocated info. to pg. 8-15.	Ward Evans May 30, 1980
	8-15, 8-16	Relocated para. 8.27.	
	10-2	Added para. (j) to para. 10.3.	

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

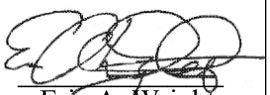
Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 8 - 761 649 (PR800814)	9-i 9-15 thru 9-18	Added Supplement 5 and pages. Added Century 21 Autopilot Supplement	Ward Evans Ward Evans August 14, 1980
Rev. 9 - 761 649 (PR801218)	2-2	Revised 2.7 (e).	
	2-3	Revised 2.9 (c).	
	3-i	Added para. 3.24; change para. title and pg. no.	
	3-4	Changed para. title, added info; moved info. to pg. 3-5.	
	3-5	Relocated info. from pg. 3-4.	
	3-10	Changed para. 3.23, title and contents with info. added; moved para. 3.25 to pg. 3-11.	
	3-11	Continued para. 3.23 addition; relocated para. 3.25 from pg. 3-10, moved para. 3.27, 3.28 and 3.29 to pg. 3-12.	
	3-12	Relocated para. 3.27, 3.28 and 3.29 from pg. 3-11.	
	6-17	Added item 4.	
	6-23	Added item 25.	
	6-25	Renumbered items.	
	6-35	Revised items 90, 91 and 93.	
	6-38	Added item 132.	
	6-42	Relocated revised item 178 from pg. 6-43.	
	6-43	Moved item 178 to pg. 6-42; renumbered item; added new item 181; revised item 182.	
	6-49	Added item 276; moved items 281, 283 and 285 to pg. 6-50.	
	6-50	Relocated items 281, 283 and 285 from pg. 6-49.	
	7-10	Revised para. 7.13.	
	7-12, 7- 13	Revised para. 7.15.	
	7-14	Revised fig. 7-11.	
	7-21	Revised para. 7.25.	
	9-i	Added supplement 6 and pages.	
	9-15 thru 9-18	Retyped supplement 5 Century 21 auto pilot; changed pg. nos.	
	9-19 thru 9-20	Added supplement 6 Piper Control Wheel Clock Installation.	Ward Evans Ward Evans Dec. 18, 1980

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 10 - 761 649 (PR810629)	ii	Revised Warning.	
	1-4	Revised para. 1.13.	
	5-1	Added warning.	
	6-7	Revised Figure 6-5.	
	6-19	Revised item 11 .	
	6-27	Revised items 33, 35 and 39.	
	6-35	Revised item 81.	
	6-38	Revised item 132.	
	6-41	Renumbered items.	
	6-42	Renumbered items; added new items 175 and 176.	
	6-43	Renumbered items; added new item 183.	
	6-45	Revised item 199.	
	6-53	Revised item 342.	
	6-54	Revised items 343, 344 and 345.	Ward Evans
	6-55	Added item 361; removed info.	Ward Evans
	7-8	Revised para 7.11	June 29, 1981
	7-12	Revised para. 7.15	
Rev. 11 - 761 649 (PR8 11120)	2-1	Revised para. 2.1.	
	3-1	Revised para. 3.1.	
	4-i	Amended pg. number.	
	4-1	Revised para 4.1 and 4.3.	
	4-3, 4-5, 4-6	Revised para. 4.5.	
	4-8	Added Note to para. 4.9.	
	4-12	Added Note to para. 4.21; moved info. to pg. 413.	
	4-13	Relocated info. from pg. 4-12; moved info. to pg. 4-14.	
	4-14	Relocated info. from pg. 4-13; added Note to para. 4.29; moved info. to pg. 4-15.	
	4-15	Relocated info. from pg. 4-14; moved para. 4.35 to pg. 4-16.	
	4-16	Relocated para. 4.35 from pg. 4-15.	
	6-17	Revised para. 6.9.	
	6-19	Revised item 7.	
	6-40	Revised item 151.	
	6-44	Added new item 184; renumbered existing items 185 thru 1882 moved item 198 to pg. 6-45.	
	6-45	Relocated item 198 from pg. 6-44; moved items 209 and 211 to pg. 6-46.	
	6-46	Relocated items 209 and 211 from pg. 6-45; moved items 229 and 231 to pg. 6-47.	

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 11 - 761 649 (PR811120) (cont)	6-47	Relocated items 229 and 231 from pg. 6-46; moved item 247 to pg. 6-48.	
	6-48	Relocated item 247 from pg. 6-47.	
	6-48	Added new item 277; renumbered items 278 thru 280.	
	6-55	Added item 361.	Ward Evans
	7-10	Revised para. 7.13.	Ward Evans
	7-21	Revised para. 7.25.	Nov. 20, 1981
	9-20	Corrected spelling errors.	
Rev. 12 - 761 649 (PR890417)	3-i	Revised Table of Contents	
	3-1	Revised para. 3.1	
	4-i thru 4-ii	Revised Table of Contents	
	6-7	Revised Figure 6-5.	
	6-15	Revised para. 6.8.	
	7-22	Revised para. 7.31	
	8-1	Revised para. 8.1	D.H. Trompler
	8-3	Revised para 8.3	D.H. Trompler
	8-4	Revised para. 8.5	
	8-11	Revised para. 8.19 and 8.21(b).	Aug. 23, 1989
	9-5	Revised Section 3 (a)	Date
	9-9	Revised Section 3 (a)	
Rev. 13 - 761 649 (PR900913)	1-3	Revised para. 1.5 (c).	
	2-2	Revised para's. 2.7 (j) and (l). Added para. 2.7 (m) and Notes.	D.H. Trompler
	7-3	Revised para. 7.5.	D.H. Trompler
	8-4	Revised para. 8.5.	October 9, 1990
Rev. 14 - 761-649 (PR050425)	iii	Added Warning.	
	4-13	Revised para. 4.27.	
	4-15	Revised para. 4.31.	
	8-1	Revised para. 8.1.	Linda J. Dicken
	8-2	Moved info. from page 8-1.	Linda J. Dicken
	8-3	Revised para. 8.3.	April 25, 2005
Rev. 15 - 761-649 (PR150731)	iv-h	Added Rev. 15 to L of R.	Eric A. Wright
	2-9	Revised Para. 2.25	Eric A. Wright
			July 31, 2015

PILOT'S OPERATING HANDBOOK LOG OF REVISIONS (cont)

Revision Number and Code	Revised Pages	Description of Revision	FAA Approval Signature and Date
Rev. 16 - 761 649 (PR170609)	ii iv-i iv-j 4-5 4-12 4-13 4-14	Added copyright info. Added Rev. 16 to L of R. Added page. Revised Para. 4.5, Cruising. Moved info. from page 4-13. Revised Para. 4.27. Moved info. from page 4-13.	 Eric A. Wright June 9, 2017
Rev. 17 - 761 649 (PR220720)	ii iv-i 4-3 4-8	Revised copyright info. Added Rev. 17 to L of R. Revised Para. 4.5, Preflight Check. Revised Para. 4.9, Preflight Check.	 Mitchell R. Cannon July 20, 2022

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SECTION 1

GENERAL

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SECTION 1

GENERAL

1.1 INTRODUCTION

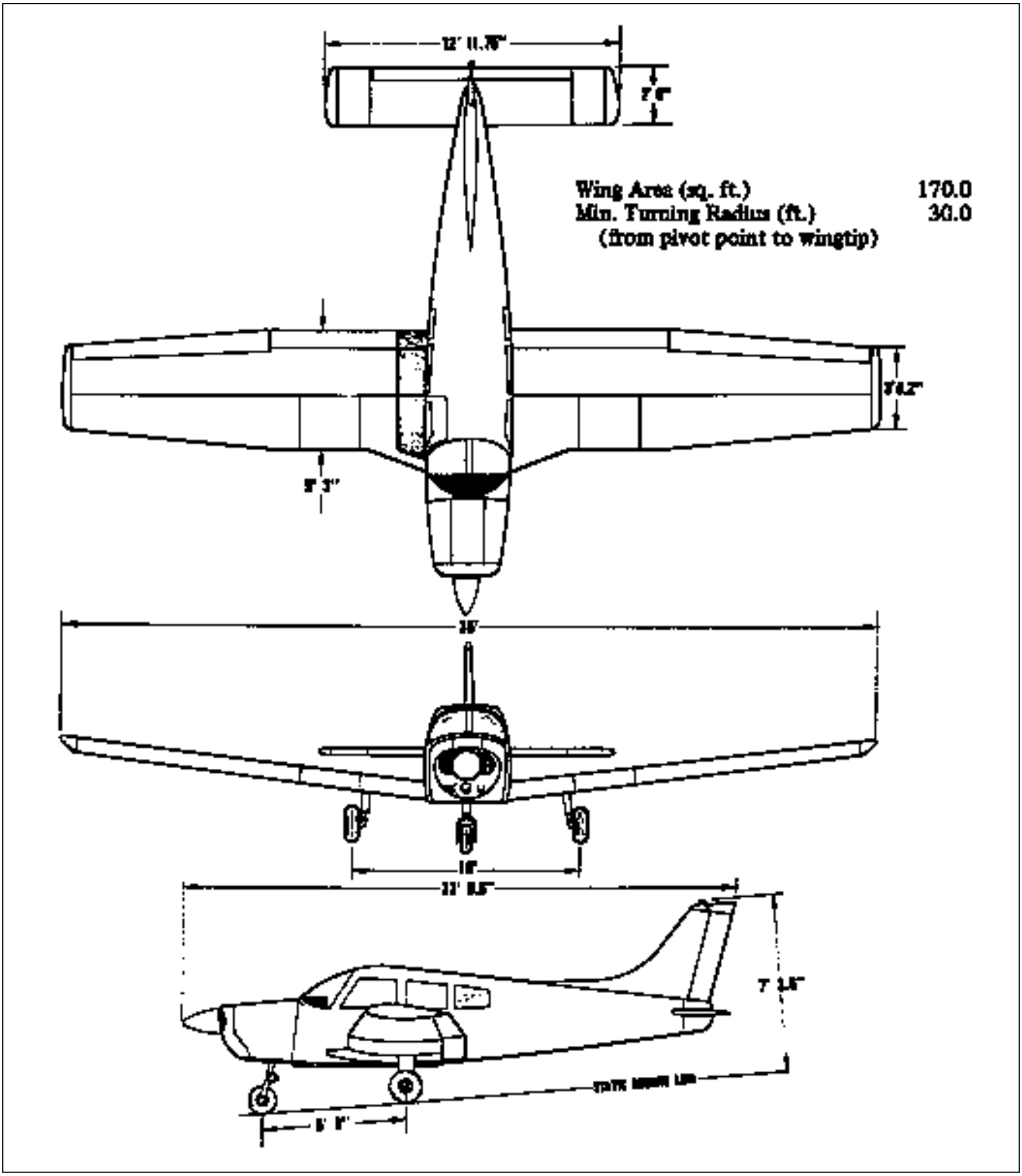
This Pilot's Operating Handbook is designed for maximum utilization as an operating guide for the pilot. It includes the material required to be furnished to the pilot by C.A.R. 3 and FAR Part 21, Subpart J. It also contains supplemental data supplied by the airplane manufacturer.

This handbook is not designed as a substitute for adequate and competent flight instruction, knowledge of current airworthiness directives, applicable federal air regulations or advisory circulars. It is not intended to be a guide for basic flight instruction or a training manual and should not be used for operational purposes unless kept in a current status.

Assurance that the airplane is in an airworthy condition is the responsibility of the owner. The pilot in command is responsible for determining that the airplane is safe for flight. The pilot is also responsible for remaining within the operating limitations as outlined by instrument markings, placards, and this handbook.

Although the arrangement of this handbook is intended to increase its in-flight capabilities, it should not be used solely as an occasional operating reference. The pilot should study the entire handbook to familiarize himself with the limitations, performance, procedures and operational handling characteristics of the airplane before flight.

The handbook has been divided into numbered (arabic) sections, each provided with a ''finger-tip'' tab divider for quick reference. The limitations and emergency procedures have been placed ahead of the normal procedures, performance and other sections to provide easier access to information that may be required in flight. The "Emergency Procedures" Section has been furnished with a red tab divider to present an instant reference to the section. Provisions for expansion of the handbook have been made by the deliberate omission of certain paragraph numbers, figure numbers, item numbers and pages noted as being left blank intentionally.



THREE VIEW

Figure 1-1

1.3 ENGINES

(a)	Number of Engines	1
(b)	Engine Manufacturer	Lycoming
(c)	Engine Model Number	0-320-D2A or 0-320-D3G
(d)	Rated Horsepower	160
(e)	Rated Speed (rpm)	2700
(f)	Bore (inches)	5.125
(g)	Stroke (inches)	3.875
(h)	Displacement (cubic inches)	319.8
(i)	Compression Ratio	8.5:1
(h)	Engine Type	Four Cylinder, Direct Drive, Horizontally Opposed, Air Cooled

1.5 PROPELLERS

(a)	Number of Propellers	1
(b)	Propeller Manufacturer	Sensenich
(c)	Model	74DM6-0-60 or 74DM6-0-58
(d)	Number of Blades	2
(e)	Propeller Diameter (inches)	
	(1) Maximum	74
	(2) Minimum	72
(f)	Propeller Type	Fixed Pitch

1.7 FUEL

(a)	Fuel Capacity (U.S. gal) (total)-	50
(b)	Usable Fuel (U.S. gal) (total)	48
(c)	Fuel	
	(1) Minimum Octane	100 - Green or 100 LL - Blue Aviation Grade
	(2) Alternate Fuel	Refer to Fuel Requirements, Section 8 - Handling, Servicing and Maintenance - paragraph 8.1, item (b).

1.9 OIL

(a)	Oil Capacity (U.S. quarts)		8
(b)	Oil Specification		Refer to latest issue of Lycoming Service Instruction 1014.
(c)	Oil Viscosity per Average Ambient Temp. for Starting	SINGLE	MULTI
	(1) Above 60°F	S.A.E. 50	S.A.E. 40 or 50
	(2) 30°F to 90°F	S.A.E. 40	S.A.E. 40
	(3) 0°F to 70°F	S.A.E. 30	S.A.E. 40 or 20W-30
	(4) Below 10°F	S.A.E. 20	S.A.E. 20W-30

1.11 MAXIMUM WEIGHTS

	NORMAL	UTILITY
(a) Maximum Takeoff Weight (lbs)	2325	2020
(b) Maximum Landing Weight (lbs)	2325	2020
(c) Maximum Weights in Baggage Compartment	200	0

1.13 STANDARD AIRPLANE WEIGHTS

Refer to Figure 6-5 for the Standard Empty Weight and the Useful Load.

1.15 BAGGAGE SPACE

(a) Compartment Volume (cubic feet)	24
(b) Entry Width (inches)	22
(c) Entry Height (inches)	20

1.17 SPECIFIC LOADINGS

(a) Wing Loading (lbs per sq ft)	13.7
(b) Power Loading (lbs per hp)	14.5

1.19 SYMBOLS, ABBREVIATIONS AND TERMINOLOGY

The following definitions are of symbols, abbreviations and terminology used throughout the handbook and those which may be of added operational significance to the pilot.

(a) General Airspeed Terminology and Symbols

CAS	Calibrated Airspeed means the indicated speed of an aircraft, corrected for position and instrument error. Calibrated airspeed is equal to true airspeed in standard atmosphere at sea level.
KCAS	Calibrated Airspeed expressed in "Knots."
GS	Ground Speed is the speed of an airplane relative to the ground.
IAS	Indicated Airspeed is the speed of an aircraft as shown on the airspeed indicator when corrected for instrument error. IAS values published in this handbook assume zero instrument error.
KIAS	Indicated Airspeed expressed in "Knots."
M	Mach Number is the ratio of true airspeed to the speed of sound.
TAS	True Airspeed is the airspeed of an airplane relative to undisturbed air which is the CAS corrected for altitude, temperature and compressability.
V_A	Maneuvering Speed is the maximum speed at which application of full available aerodynamic control will not overstress the airplane.
V_{FE}	Maximum Flap Extended Speed is the highest speed permissible with wing flaps in a prescribed extended position.
V_{NE}/M_{NE}	Never Exceed Speed or Mach Number is the speed limit that may not be exceeded at any time.
V_{NO}	Maximum Structural Cruising Speed is the speed that should not be exceeded except in smooth air and then only with caution.
V_S	Stalling Speed or the minimum steady flight speed at which the airplane is controllable.
V_{SO}	Stalling Speed or the minimum steady flight speed at which the airplane is controllable in the landing configuration.
V_X	Best Angle-of-Climb Speed is the airspeed which delivers the greatest gain of altitude in the shortest possible horizontal distance.
V_Y	Best Rate-of-Climb Speed is the airspeed which delivers the greatest gain in altitude in the shortest possible time.

(b) Meteorological Terminology

ISA	International Standard Atmosphere in which: The air is a dry perfect gas; The temperature at sea level is 15° Celsius (59° Fahrenheit); The pressure at sea level is 29.92 inches hg. (1013 mb); The temperature gradient from sea level to the altitude at which the temperature is -56.5°C (-69.7°F) is -0.00198°C (-0.003566°F) per foot and zero above that altitude.
OAT	Outside Air Temperature is the free air static temperature, obtained either from inflight temperature indications or ground meteorological sources, adjusted for instrument error and compressibility effects.
Indicated Pressure Altitude	The number actually read from an altimeter when the barometric subscale has been set to 29.92 inches of mercury (1013 millibars).
Pressure Altitude	Altitude measured from standard sea-level pressure (29.92 in. Hg) by a pressure or barometric altimeter. It is the indicated pressure altitude corrected for position and instrument error. In this handbook, altimeter instrument errors are assumed to be zero.
Station Pressure	Actual atmospheric pressure at field elevation.
Wind	The wind velocities recorded as variables on the charts of this handbook are to be understood as the headwind or tailwind components of the reported winds..

(c) Power Terminology

Takeoff Power	Maximum power permissible for takeoff.
Maximum Continuous Power	Maximum power permissible continuously during flight.
Maximum Climb Power	Maximum power permissible during climb.
Maximum Cruise Power	Maximum power permissible during cruise.

(d) Engine Instruments

EGT Gauge	Exhaust Gas Temperature Gauge

(e) Airplane Performance and Flight Planning Terminology

Climb Gradient	The demonstrated ratio of the change in height during a portion of a climb, to the horizontal distance traversed in the same time interval.
Demonstrated Crosswind Velocity WIND)	The demonstrated crosswind velocity is the velocity of the crosswind component for which adequate control of the airplane during takeoff and landing was actually demonstrated during certification tests.
Accelerate-Stop Distance	The distance required to accelerate an airplane to a specified speed and, assuming failure of an engine at the instant that speed is attained, to bring the airplane to a stop.
MEA	Minimum en route IFR altitude.
Route Segment	A part of a route. Each end of that part is identified by: (1) a geographical location: or (2) a point at which a definitive radio fix can be established.

(f) Weight and Balance Terminology

Reference Datum	An imaginary vertical plane from which all horizontal distances are measured for balance purposes.
Station	A location along the airplane fuselage usually given in terms of distance from the reference datum.
Arm	The horizontal distance from the reference datum to the center of gravity (C.G.) of an item.
Moment	The product of the weight of an item multiplied by its arm. (Moment divided by a constant is used to simplify balance calculations by reducing the number of digits.)
Center of Gravity (C.G.)	The point at which an airplane would balance if suspended. Its distance from the reference datum is found by dividing the total moment by the total weight of the airplane.
C.G. Arm	The arm obtained by adding the airplane's individual moments and dividing the sum by the total weight.
C.G. Limits	The extreme center of gravity locations within which the airplane must be operated at a given weight.
Usable Fuel	Fuel available for flight planning.
Unusable Fuel	Fuel remaining after a runout test has been completed in accordance with governmental regulations.
Standard Empty Weight	Weight of a standard airplane including unusable fuel, full operating fluids and full oil.
Basic Empty Weight	Standard empty weight plus optional equipment.
Payload	Weight of occupants, cargo and baggage.
Useful Load	Difference between takeoff weight, or ramp weight if applicable, and basic empty weight.
Maximum Ramp Weight	Maximum weight approved for ground maneuver. (It includes weight of start, taxi and run up fuel.)
Maximum Takeoff Weight	Maximum weight approved for the start of the takeoff run.
Maximum Landing Weight	Maximum weight approved for the landing touchdown.
Maximum Zero Fuel Weight	Maximum weight exclusive of usable fuel.

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1.21 CONVERSION FACTORS

MULTIPLY	BY	TO OBTAIN
acres	0.4047	ha
	43560	sq. ft.
	0.0015625	sq. mi.
atmospheres (atm)	76	cm Hg
	29.92	in. Hg
	1.0133	bar
	1.033	kg/cm^2
	14.70	lb./sq. in.
	2116	lb./sq. ft.
bars (bar)	0.98692	atm.
	14.503768	lb./sq. in.
British Thermal Unit (BTU)	0.2519958	kg-cal
centimeters (cm)	0.3937	in.
	0.032808	ft.
centimeters of mercury at 0°C (cm Hg)	0.01316	atm
	0.3937	in. Hg
	0.1934	lb./sq. in.
	27.85	lb./sq. ft.
	135.95	kg/m^2
centimeters per second (cm/sec.)	0.032808	ft./sec.
	1.9685	ft./min.
	0.02237	mph
cubic centimeters (cm^3)	0.03381	fl. oz.
	0.06102	cu. in.
	3.531×10^{-5}	cu. ft.
	0.001	1
	2.642×10^{-4}	U.S. gal.
cubic feet (cu.ft.)	28317	cm^3
	0.028317	m^3
	1728	cu. in.
	0.037037	cu. yd.
	7.481	U.S. gal.
	28.32	1
cubic feet per minute (cu. ft./min.)	0.472	1/sec.
	0.028317	m^3/min.

MULTIPLY	BY	TO OBTAIN
cubic inches (cu. in.)	16.39	cm^3
	1.639×10^{-5}	m^3
	5.787×10^{-4}	cu. ft.
	0.5541	fl. oz.
	0.01639	1
	4.329×10^{-3}	U.S. gal.
	0.01732	U.S. qt.
cubic meters (m^3)	61024	cu. in.
	1.308	cu. yd.
	35.3147	cu. ft.
	264.2	U.S. gal.
cubic meters per minute (m^3/min.)	35.3147	cu. ft./min.
cubic yards (cu. yd.)	27	cu. ft.
	0.7646	m^3
	202	U.S. gal.
degrees (arc)	0.01745	radians
degrees per second (deg./sec.)	0.01745	radians/sec.
drams, fluid (dr. fl.)	0.125	fl. oz.
drams, avdp. (dr. avdp.)	0.0625	oz. avdp.
feet (ft.)	30.48	cm
	0.3048	m
	12	in.
	0.33333	yd.
	0.0606061	rod
	1.894×10^{-4}	mi.
	1.645×10^{-4}	NM
feet per minute (ft./min.)	0.01136	mph
	0.01829	km/hr.
	0.508	cm/sec.
	0.00508	m/sec.

MULTIPLY	BY	TO OBTAIN
feet per second (ft./sec.)	0.6818	mph
	1.097	km/hr.
	30.48	cm/sec.
	0.5921	kts.
foot-pounds (ft.-lb.)	0.138255	m-kg
	3.24×10^{-4}	kg-cal
foot-pounds per minute (ft.-lb./min.)	3.030×10^{-5}	hp
foot-pounds per second (ft.-lb./sec.)	1.818×10^{-5}	hp
gallons, Imperial (Imperial gal.)	277.4	cu. in.
	1.201	U.S. gal.
	4.546	l
gallons, U.S. dry (U.S. gal. dry)	268.8	cu. in.
	1.556×10^{-1}	cu. ft.
	1.164	U.S. gal.
	4.405	l
gallons, U.S. Iiquid (U.S. gal.)	231	cu. in.
	0.1337	cu. ft.
	4.951×10^{-3}	cu. yd.
	3785.4	cm^3
	3.785×10^{-3}	m^3
	3.785	l
	0.83268	Imperial gal.
	128	fl. oz.
gallons per acre (gal./acre)	9.353	l/ha
grams (g)	0.001	kg
	0.3527	oz. avdp.
	2.205×10^{-3}	lb.
grams per centimeter (g/cm)	0.1	kg/m
	6.721×10^{-2}	lb./ft.
	5.601×10^{-3}	lb./in.
grams per cubic centimeter (g/cm^3)	1000	kg/m^3
	0.03613	lb./cu. in.
	62.43	lb./cu. ft.

MULTIPLY	BY	TO OBTAIN
hectares (ha)	2.471	acres
	107639	sq. ft.
	10000	m^2
horsepower (hp)	33000	ft.-lb./min.
	550	ft.-lb./sec.
	76.04	m-kg/sec.
	1.014	metric hp
horsepower, metric	75	m-kg/sec.
	0.9863	hp
inches (in.)	25.40	mm
	2.540	cm
	0.0254	m
	0.08333	ft.
	0.027777	yd.
inches of mercury at 0°C (in. Hg)	0.033421	atm
	0.4912	lb./sq. in.
	70.73	lb./sq. ft.
	345.3	kg/m^2
	2.540	cm Hg
	25.40	mm Hg
inch-pounds (in.-lb.)	0.011521	m-kg
kilograms (kg)	2.204622	lb.
	35.27	oz. avdp.
	1000	g
kilogram-calories (kg-cal)	3.9683	BTU
	3087	ft.-lb.
	426.9	m-kg
kilograms per cubic meter (kg/m^3)	0.06243	lb./cu. ft.
	0.001	g/cm^3
kilograms per hectare (kg/ha)	0.892	lb./acre
kilograms per square centimeter (kg/cm^2)	0.9678	atm
	28.96	in. Hg
	14.22	lb./sq. in.
	2048	lb./sq. ft.

MULTIPLY	BY	TO OBTAIN
kilograms per square meter (kg/m^2)	2.896×10^{-3}	in. Hg
	1.422×10^{-3}	lb./sq. in.
	0.2048	lb./sq. ft.
kilometers (km)	1×10^{-5}	cm
	3280.8	ft.
	0.6214	mi.
	0.53996	NM
kilometers per hour (km/hr.)	0.9113	ft./sec.
	58.68	ft./min.
	0.53996	kt
	0.6214	mph
	0.27778	m/sec.
	16.67	m/min.
knots (kt)	1	nautical mph
	1.689	ft./sec.
	1.1516	statute mph
	1.852	km/hr.
	51.48	m/sec.
liters (1)	1000	cm^3
	61.02	cu. in.
	0.03531	cu. ft.
	33.814	fl. oz.
	0.264172	U.S. gal.
	0.2200	Imperial gal.
	1.05669	qt.
liters per hectare (l/ha)	13.69	fl. oz./acre
	0.107	gal./acre
liters per second (1/sec.)	2.12	cu. ft./min.
meters (m)	39.37	in.
	3.280840	ft.
	1.0936	yd.
	0.198838	rod
	6.214×10^{-4}	mi.
	5.3996×10^{-4}	NM
meter-kilogram (m-kg)	7.23301	ft.-lb.
	86.798	in.-lb.

MULTIPLY	BY	TO OBTAIN
meters per minute (m/min.)	0.06	km/hr.
meters per second (m/sec.)	3.280840	ft./sec.
	196.8504	ft./min.
	2.237	mph
	3.6	km/hr.
microns	3.937×10^{-5}	in.
miles, statute (mi.)	5280	ft.
	1.6093	km
	1609.3	m
	0.8684	NM
miles per hour (mph)	44.7041	cm/sec.
	4.470×10^{-1}	m/sec.
	1.467	ft./sec.
	88	ft./min.
	1.6093	km/hr.
	0.8684	kt
miles per hour square (m/hr. sq.)	2.151	ft./sec. sq.
millibars	2.953×10^{-2}	in. Hg
millimeters (mm)	0.03937	in.
millimeters of mercury at 0°C (mm Hg)	0.03937	in. Hg
nautical miles (NM)	6080	ft.
	1.1516	statute mi.
	1852	m
	1.852	km
ounces, avdp. (oz. avdp.)	28.35	g
	16	dr. avdp.
ounces, fluid (fl. oz.)	8	dr. fl.
	29.57	cm^3
	1.805	cu. in.
	0.0296	1
	0.0078	U.S. gal.

MULTIPLY	BY	TO OBTAIN
ounces, fluid per acre (fl. oz./ acre)	0.073	l/ha
pounds (lb.)	0.453592	kg
	453.6	g
	3.108×10^{-2}	slug
pounds per acre (lb./acre)	1.121	kg/ha
pounds per cubic foot (lb./cu. ft.)	16.02	kg/m^3
pounds per cubic inch (lb./cu. in.)	1728	lb./cu. ft.
	27.68	g/cm^3
pounds per square foot (lb./sq. ft.)	0.1414	in. Hg
	4.88243	kg/m^2
	4.725×10^{-4}	atm
pounds per square inch (psi or lb./sq. in.)	5.1715	cm Hg
	2.036	in. Hg
	0.06804	atm
	0.0689476	bar
	703.1	kg/m^2
quart, U.S. (qt.)	0.94635	1
	57.749	cu. in.
radians	57.30	deg. (arc)
	0.1592	rev.
radians per second (radians/sec.)	57.30	deg./sec.
	0.1592	rev./sec.
	9.549	rpm
revolutions (rev.)	6.283	radians
revolutions per minute (rpm or rev./min.)	0.1047	radians/sec.
revolutions per second (rev./sec.)	6.283	radians/sec.

MULTIPLY	BY	TO OBTAIN
rod	16.5	ft.
	5.5	yd.
	5.029	m
slug	32.174	lb.
square centimeters (cm^2)	0.1550	sq. in.
	0.001076	sq. ft.
square feet (sq. ft.)	929	cm^2
	0.092903	m^2
	144	sq. in.
	0.1111	sq. yd.
	2.296×10^{-5}	acres
square inches (sq. in.)	6.4516	cm^2
	6.944×10^{-3}	sq. ft.
square kilometers (km^2)	0.3861	sq. mi.
square meters (m^2)	10.76391	sq. ft.
	1.196	sq. yd.
	0.0001	ha
square miles (sq. mi.)	2.590	km^2
	640	acres
square rods (sq. rods)	30.25	sq. yd.
square yards (sq. yd.)	0.8361	m^2
	9	sq. ft.
	0.0330579	sq. rods
yards (yd.)	0.9144	m
	3	ft.
	36	in.
	0.181818	rod

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SECTION 2

LIMITATIONS

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SECTION 2

LIMITATIONS

2.1 GENERAL

This section provides the "FAA Approved" operating limitations, instrument markings, color coding and basic placards necessary for operation of the airplane and its systems.

This airplane must be operated as a normal or utility category airplane in compliance with the operating limitations stated in the form of placards and markings and those given in this section and handbook.

Limitations associated with those optional systems and equipment which require handbook supplements can be found in Section 9 (Supplements).

2.3 AIRSPEED LIMITATIONS

SPEED	KIAS	KCAS
Never Exceed Speed (V_{NE}) - Do not exceed this speed in any operation.	160	153
Maximum Structural Cruising Speed (V_{NO}) - Do not exceed this speed except in smooth air and then only with caution.	126	122
Design Maneuvering Speed (V_A) - Do not make full or abrupt control movements above this speed.		
At 2325 LBS. G.W.	111	108
At 1531 LBS. G.W.	88	89
Maximum Flaps Extended Speed (V_{FE}) - Do not exceed this speed with the flaps extended.	103	100

CAUTION

Maneuvering speed decreases at lighter weight as the effects of aerodynamic forces become more pronounced. Linear interpolation may be used for intermediate gross weights. Maneuvering speed should not be exceeded while operating in rough air.

2.5 AIRSPEED INDICATOR MARKINGS

MARKING	IAS
Red Radial Line (Never Exceed)	160 KTS
Yellow Arc (Caution Range - Smooth Air Only)	126 KTS to 160 KTS
Green Arc (Normal Operating Range)	50 KTS to 126 KTS
White Arc (Flap Down)	44 KTS to 103 KTS

2.7 POWER PLANT LIMITATIONS

(a)	Number of Engines	1
(b)	Engine Manufacturer	Lycoming
(c)	Engine Model No.	0-320-D2A or 0-320-D3G
(d)	Engine Operating Limits	
	(1) Maximum Horsepower	160
	(2) Maximum Rotation Speed (RPM)	2700
	(3) Maximum Oil Temperature	245°F
(e)	Oil Pressure	
	Minimum (red line)	25 PSI
	Maximum (red line)	100 PSI
(f)	Fuel Pressure	
	Minimum (red line)	5 PSI
	Maximum (red line)	8 PSI
(g)	Fuel (minimum grade)	100 or 100LL Aviation Grade
(h)	Number of Propellers	1
(i)	Propeller Manufacturer	Sensenich
(j)	Propeller Model	74DM6-0-60 or 74DM6-0-58
(k)	Propeller Diameter	
	Minimum	72 IN.
	Maximum	74 IN.
(l)	74DM6-0-60 Propeller Tolerance (static rpm at maximum permissible throttle-setting, Sea Level, ISA)	Not above 2430 RPM Not below 2330 RPM

NOTE

Refer to the airplane maintenance manual for test procedure to determine approved static rpm under non standard conditions.

(m)	74DM6-0-58 Propeller Tolerance (static RPM at maximum permissible throttle setting, Sea Level, ISA)	Not above 2465 RPM Not below 2365 RPM

NOTE

Refer to the airplane maintenance manual for test procedure to determine approved static rpm under non standard conditions.

2.9 POWER PLANT INSTRUMENT MARKINGS

(a)	Tachometer	
	Green Arc (Normal Operating Range)	500 to 2700 RPM
	Red Line (Maximum Continuous Power)	2700 RPM
(b)	Oil Temperature	
	Green Arc (Normal Operating Range)	75° to 245°F
	Red Line (Maximum)	245°F
(c)	Oil Pressure	
	Green Arc (Normal Operating Range)	60 PSI to 90 PSI
	Yellow Arc (Caution Range) (Idle)	25 PSI to 60 PSI
	Yellow Arc (Ground Warm-Up)	90 PSI to 100 PSI
	Red Line (Minimum)	25 PSI
	Red Line (Maximum)	100 PSI
(d)	Fuel Pressure	
	Green Arc (Normal Operating Range)	.5 PSI to 8 PSI
	Red Line (Minimum)	.5 PSI
	Red Line (Maximum)	8 PSI

2.1 WEIGHT LIMITS

		NORMAL	UTILITY
(a)	Maximum Weight	2325 LBS	2020 LBS
(b)	Maximum Baggage	200 LBS	0 LBS

NOTE

Refer to Section 5 (Performance) for maximum weight as limited by performance.

2.13 CENTER OF GRAVITY LIMITS

(a) Normal Category

Weight Pounds	Forward Limit Inches Aft of Datum	Rearward Limit Inches Aft of Datum
2325	87.0	93.0
1950 (and less)	83.0	93.0

(b) Utility Category

Weight Pounds	Forward Limit Inches Aft of Datum	Rearward Limit Inches Aft of Datum
1950 (and less)	83.0	93.0
2020	83.8	93.0

NOTES

Straight line variation between points given.

The datum used is 78.4 inches ahead of the wing leading edge at the inboard intersection of the straight and tapered section.

It is the responsibility of the airplane owner and the pilot to insure that the airplane is properly loaded. See Section 6 (Weight and Balance) for proper loading instructions.

2.15 MANEUVER LIMITS

(a) Normal Category - All acrobatic maneuvers including spins prohibited.
(b) Utility Category - Approved maneuvers for bank angles exceeding 60°:

	Entry Speed
Steep Turns	111 KIAS
Lazy Eights	111 KIAS
Chandelles	111 KIAS

2.17 FLIGHT LOAD FACTORS

		NORMAL	UTILITY
(a)	Positive Load Factor (Maximum)	3.8 G	4.4 G
(b)	Negative Load Factor (Maximum)	No inverted maneuvers approved	

2.19 TYPES OF OPERATION

The airplane is approved for the following operations when equipped in accordance with FAR 91 or FAR 135.

(a) Day V.F.R.
(b) Night V.F.R.
(c) Day I.F.R.
(d) Night I.F.R.
(e) Non Icing

2.21 FUEL LIMITATIONS

(a)	Total Capacity	50 U.S. GAL
(b)	Unusable Fuel The unusable fuel for this airplane has been determined as 1.0 gallon in each wing in critical flight attitudes.	2 U.S. GAL
(c)	Usable Fuel The usable fuel in this airplane has been determined as 24.0 gallons in each wing.	48 U.S. GAL

2.23 NOISE LEVEL

The noise level of this aircraft is 72dBA.

No determination has been made by the Federal Aviation Administration that the noise levels of this airplane are or should be acceptable or unacceptable for operation at, into, or out of, any airport.

The above statement not withstanding, the noise level stated above has been verified by and approved by the Federal Aviation Administration in noise level test flights conducted in accordance with FAR 36, Noise Standards - Aircraft Type and Airworthiness Certification. This aircraft model is in compliance with all FAR 36 noise standards applicable to this type.

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2.25 PLACARDS

In full view of the pilot:

"THIS AIRPLANE MUST BE OPERATED AS A NORMAL OR UTILITY CATEGORY AIRPLANE IN COMPLIANCE WITH THE OPERATING LIMITATIONS STATED IN THE FORM OF PLACARDS, MARKINGS AND MANUALS.

ALL MARKINGS AND PLACARDS ON THIS AIRPLANE APPLY TO ITS OPERATION AS A UTILITY CATEGORY AIRPLANE. FOR NORMAL AND UTILITY CATEGORY OPERATION, REFER TO THE PILOT'S OPERATING HANDBOOK.

NO ACROBATIC MANEUVERS ARE APPROVED FOR NORMAL CATEGORY OPERATIONS. SPINS ARE PROHIBITED FOR NORMAL AND UTILITY CATEGORY."

In full view of the pilot, one of the following takeoff checklists and one of the following landing check lists will be installed:

TAKEOFF CHECK LIST

Fuel on proper tank	Mixture set	Fasten belts/harness
Electric fuel pump on	Seat backs erect	Trim tab - set
Engine gauges checked		Controls - free
Flaps - set		Door - latched
Carb heat off		Air conditioner - off

TAKEOFF CHECK LIST

Fuel on proper tank	Mixture set	Fasten belts/harness
Electric fuel pump-on	Primer locked	Trim tab - set
Engine gages checked	Seat backs erect	Controls - free
Flaps - set		Door - latched
Carb. heat off		Air conditioner off

LANDING CHECK LIST

Fuel on proper tank		Flaps - set (103 KIAS max.)
Mixture rich	Seat backs erect	Fasten belts/harness
Electric fuel pump on		Air conditioner off

LANDING CHECK LIST

Fuel on proper tank		Flaps - set (White Arc).
Mixture rich	Seat backs erect	Fasten belts/harness
Electric fuel pump on		Air conditioner off

The "AIR COND OFF" item in the above takeoff and landing check lists is mandatory for air conditioned aircraft only.

In full view of the pilot, in the area of the air conditioner control panel when the air conditioner is installed:

"WARNING—AIR CONDITIONER MUST BE OFF TO INSURE NORMAL TAKEOFF CLIMB PERFORMANCE."

Adjacent to upper door latch:

"ENGAGE LATCH BEFORE FLIGHT."

On inside of the baggage compartment door:

"BAGGAGE MAXIMUM 200 LBS"
"UTILITY CATEGORY OPERATION - NO BAGGAGE OR AFT PASSENGERS ALLOWED. NORMAL CATEGORY OPERATION - SEE PILOT'S OPERATING HANDBOOK WEIGHT AND BALANCE SECTION FOR BAGGAGE AND AFT PASSENGER LIMITATIONS."

In full view of the pilot:

"MANEUVERING SPEED 111 KIAS AT 2325 LBS. (SEE P.O.H.)" OR "VA = 111 KIAS AT 2325 #(SEE P.O.H.)"

"UTILITY CATEGORY OPERATION - NO AFT PASSENGERS ALLOWED."

"DEMONSTRATED CROSS WIND COMPONENT - 17 KTS." or "DEMO. X-WIND 17 KTS."

In full view of the pilot when the oil cooler winterization kit is installed:

"OIL COOLER WINTERIZATION PLATE TO BE REMOVED WHEN AMBIENT TEMPERATURE EXCEEDS 50°F."

In full view of the pilot:

"UTILITY CATEGORY OPERATION ONLY."

(1) NO AFT PASSENGERS ALLOWED.
(2) ACROBATIC MANEUVERS ARE LIMITED TO THE FOLLOWING:

	ENTRY SPEED
SPINS PROHIBITED	
STEEP TURNS	111 KIAS
LAZY EIGHTS	111 KIAS
CHANDELLES	111 KIAS

In full view of the pilot:

"WARNING — TURN OFF STROBE LIGHTS WHEN IN CLOSE PROXIMITY TO GROUND OR DURING FLIGHT THROUGH CLOUD, FOG OR HAZE."

Adjacent to fuel filler caps:

OR

FUEL - 100 OR 100LL AVIATION GRADE

OR

FUEL - 100 AVIATION GRADE MIN.
USABLE CAPACITY 24 GAL.
USABLE CAPACITY TO BOTTOM OF
FILLER NECK INDICATOR 17 GAL.

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EMERGENCY PROCEDURES

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SECTION 3

EMERGENCY PROCEDURES

3.1 GENERAL

This section provides the recommended procedures for coping with various emergency or critical situations. All of the emergency procedures required by the FAA as well as those necessary for operation of the airplane, as determined by the operating and design features of the airplane, are presented.

Emergency procedures associated with optional systems and equipment which require handbook supplements are presented in Section 9, Supplements.

This section is divided into two basic parts. The first part contains the emergency procedures checklists. These checklists supply an immediate action sequence to be followed during critical situations with little emphasis on the operation of the systems.

The second part of the section provides amplified emergency procedures corresponding to the emergency procedures checklist items. These amplified emergency procedures contain additional information to provide the pilot with a more complete description of the procedures so they may be more easily understood.

Pilots must familiarize themselves with the procedures given in this section and must be prepared to take the appropriate action should and emergency situation arise. The procedures are offered as a course of action for coping with the particular situation or condition described. They are not a substitute for sound judgement and common sense.

Most basic emergency procedures are a normal part of pilot training. The information presented in this section is not intended to replace this training. This information is intended to provide a source of reference for the procedures which are applicable to this airplane. The pilot should review standard emergency procedures periodically to remain proficient in them.

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3.3 EMERGENCY PROCEDURES CHECK LIST

ENGINE FIRE DURING START

Starter .....crank engine
Mixture .....idle cut-off
Throttle .....open
Electric fuel pump.....OFF
Fuel selector.....OFF
Abandon if fire continues

ENGINE POWER LOSS DURING TAKEOFF

If sufficient runway remains for a normal landing, land straight ahead.

If insufficient runway remains:
Maintain safe airspeed
Make only shallow turn to avoid obstructions
Flaps as situation requires

If sufficient altitude has been gained to attempt a restart:
Maintain safe airspeed
Fuel selector.....switch to tank containing fuel
Electric fuel pump.....check ON
Mixture .....check RICH
Carburetor heat .....ON
Primer .....locked
If power is not regained, proceed with power off landing.

ENGINE POWER LOSS IN FLIGHT

Fuel selector.....switch to tank containing fuel
Electric fuel pump .....ON
Mixture .....RICH
Carburetor heat .....ON
Engine gauges .....check for indication of cause of power loss
Primer.....check locked
If no fuel pressure is indicated, check tank selector position to be sure it is on a tank containing fuel.

When power is restored:
Carburetor heat .....OFF
Electric fuel pump.....OFF
If power is not restored prepare for power off landing.
Trim for 73 KIAS

POWER OFF LANDING

Locate suitable field.
Establish spiral pattern.
1000 ft. above field at downwind position for normal landing approach.
When field can easily be reached slow to 63 KIAS for shortest landing.

Touchdowns should normally be made at lowest possible airspeed with full flaps.

When committed to landing:
Ignition.....OFF
Master switch.....OFF
Fuel selector.....OFF
Mixture .....idle cut-off
Seat belt and harness.....tight

FIRE IN FLIGHT

Source of fire .....check

Electrical fire (smoke in cabin):
Master switch.....OFF
Vents .....open
Cabin heat .....OFF
Land as soon as practicable.

Engine fire:
Fuel selector.....OFF
Throttle .....CLOSED
Mixture .....idle cut-off
Electric fuel pump .....check OFF
Heater.....OFF
Defroster .....OFF
Proceed with POWER OFF LANDING procedure.

LOSS OF OIL PRESSURE

Land as soon as possible and investigate cause.
Prepare for power off landing.

LOSS OF FUEL PRESSURE

Electric fuel pump .....ON
Fuel selector .....check on full tank

HIGH OIL TEMPERATURE

Land at nearest airport and investigate the problem. Prepare for power off landing.

ELECTRICAL FAILURES

ALT annunciator light illuminated:
Ammeter.....Check to verify inop. alt.

If ammeter shows zero:
ALT switch .....OFF

Reduce electrical loads to minimum:
ALT circuit breaker.....Check and reset as required
ALT switch .....ON

If power not restored:
ALT switch .....OFF

If alternator output cannot be restored, reduce electrical loads and land as soon as practical. The battery is the only remaining source of electrical power.

ELECTRICAL OVERLOAD (Alternator over 20 amps above known electrical load)

FOR AIRPLANES WITH INTERLOCKED BAT AND ALT SWITCH OPERATION.

Electrical load .....Reduce

If alternator loads are reduced:
ALT switch .....OFF

Land as soon as practical. Battery is the only remaining source of power. Anticipate complete electrical failure.

ELECTRICAL OVERLOAD (Alternator over 20 amps above known electrical load)

FOR AIRPLANES WITH SEPARATE BAT AND ALT SWITCH OPERATION

ALT switch .....ON
BATT switch .....OFF

If alternator loads are reduced:
Electrical load .....Reduce to Minimum

Land as soon as practical.

NOTE

Due to increased system voltage and radio frequency noise, operation with ALT switch ON and BATT switch OFF should be made only when required by an electrical system failure.

If alternator loads are not reduced:
ALT switch .....OFF
BATT switch.....As required

Land as soon as possible. Anticipate complete electrical failure.

SPIN RECOVERY

Throttle.....idle
Ailerons.....neutral
Rudder.....full opposite to direction of rotation
Control wheel.....full forward
Rudder.....neutral (when rotation stops)
Control wheel .....as required to smoothly regain level flight altitude

OPEN DOOR

If both upper and lower latches are open, the door will trail slightly open and airspeeds will be reduced slightly.

To close the door in flight:
Slow airplane to 89 KIAS
Cabin vents.....close
Storm window.....open
If upper latch is open.....latch
If side latch is open.....pull on arm rest while moving latch handle to latched position.

If both latches are open .....latch side latch then top latch

ENGINE ROUGHNESS

Carburetor heat.....ON

If roughness continues after one min:
Carburetor heat.....OFF
Mixture .....adjust for max. smoothness
Electric fuel pump.....ON
Fuel selector.....switch tanks
Engine gauges.....check
Magneto switch .....“L” then “R” then “BOTH”

If operation is satisfactory on either one, continue on that magneto at reduced power and full “RICH” mixture to first airport.

Prepare for power off landing.

CARBURETOR ICING

Carburetor heat.....ON
Mixture .....adjust for max. smoothness

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3.5 AMPLIFIED EMERGENCY PROCEDURES (GENERAL)

The following paragraphs are presented to supply additional information for the purpose of providing the pilot with a more complete understanding of the recommended course of action and probable cause of an emergency situation.

3.7 ENGINE FIRE DURING START

Engine fires during start are usually the result of overpriming. The first attempt to extinguish the fire is to try to start the engine and draw the excess fuel back into the induction system.

If a fire is present before the engine has started, move the mixture control to idle cut-off, open the throttle and crank the engine. This is an attempt to draw the fire back into the engine.

If the engine has started, continue operating to try to pull the fire into the engine.

In either case (above), if fire continues more than a few seconds, the fire should be extinguished by the best available external means.

The fuel selector valves should be “OFF” and the mixture at idle cut-off if an external fire extinguishing method is to be used.

3.9 ENGINE POWER LOSS DURING TAKEOFF

The proper action to be taken if loss of power occurs during takeoff will depend on the circumstances of the particular situation.

If sufficient runway remains to complete a normal landing, land straight ahead.

If insufficient runway remains, maintain a safe airspeed and make only a shallow turn if necessary to avoid obstructions. Use of flaps depends on the circumstances. Normally, flaps should be fully extended for touchdown.

If sufficient altitude has been gained to attempt a restart, maintain a safe airspeed and switch the fuel selector to another tank containing fuel. Check the electric fuel pump to insure that it is “ON” and that the mixture is “RICH.” The carburetor heat should be “ON” and the primer locked.

If engine failure was caused by fuel exhaustion, power will not be regained after switching fuel tanks until the empty fuel lines are filled. This may require up to ten seconds.

If power is not regained, proceed with the Power Off Landing procedure (refer to the emergency check list and paragraph 3.13).

3.11 ENGINE POWER LOSS IN FLIGHT

Complete engine power loss is usually caused by fuel flow interruption and power will be restored shortly after fuel flow is restored. If power loss occurs at a low altitude, the first step is to prepare for an emergency landing (refer to paragraph 3.13). An airspeed of at least 73 KIAS should be maintained.

If altitude permits, switch the fuel selector to another tank containing fuel and turn the electric fuel pump "ON." Move the mixture control to "RICH" and the carburetor heat to "ON." Check the engine gauges for an indication of the cause of the power loss. Check to insure the primer is locked. If no fuel pressure is indicated, check the tank selector position to be sure it is on a tank containing fuel.

When power is restored move the carburetor heat to the "OFF" position and turn "OFF" the electric fuel pump.

If the preceding steps do not restore power, prepare for an emergency landing.

If time permits, turn the ignition switch to "L" then to "R" then back to "BOTH." Move the throttle and mixture control levers to different settings. This may restore power if the problem is too rich or too lean a mixture or if there is a partial fuel system restriction. Try other fuel tanks. Water in the fuel could take some time to be used up, and allowing the engine to windmill may restore power. If power loss is due to water, fuel pressure indications will be normal.

If engine failure was caused by fuel exhaustion power will not be restored after switching fuel tanks until the empty fuel lines are filled. This may require up to ten seconds.

If power is not regained, proceed with the Power Off Landing procedure (refer to the emergency check list and paragraph 3.13).

3.13 POWER OFF LANDING

If loss of power occurs at altitude, trim the aircraft for best gliding angle (73 KIAS) and look for a suitable field. If measures taken to restore power are not effective, and if time permits, check your charts for airports in the immediate vicinity; it may be possible to land at one if you have sufficient altitude. If possible, notify the FAA by radio of your difficulty and intentions. If another pilot or passenger is aboard, let him help.

When you have located a suitable field, establish a spiral pattern around this field. Try to be at 1000 feet above the field at the downwind position, to make a normal landing approach. When the field can easily be reached, slow to 63 KIAS for the shortest landing. Excess altitude may be lost by widening your pattern, using flaps or slipping, or a combination of these.

Touchdown should normally be made at the lowest possible airspeed.

When committed to a landing shut "OFF" the master and ignition switches. Flaps may be used as desired. Turn the fuel selector valve to "OFF" and move the mixture to idle cut-off. The seat belts and shoulder harness should be tightened. Touchdown should be normally made at the lowest possible airspeed.

3.15 FIRE IN FLIGHT

The presence of fire is noted through smoke, smell and heat in the cabin. It is essential that the source of the fire be promptly identified through instrument readings, character of the smoke, or other indications since the action to be taken differs somewhat in each case.

Check for the source of the fire first.

If an electrical fire is indicated (smoke in the cabin), the master switch should be turned “OFF.” The cabin vents should be opened and the cabin heat turned “OFF.” A landing should be made as soon as possible.

If an engine fire is present, switch the fuel selector to “OFF” and close the throttle. The mixture should be at idle cut-off. Turn the electric- fuel pump “OFF.” In all cases, the heater and defroster should be “OFF.” If radio communication is not required, select master switch “OFF.” Proceed with power off landing procedure.

NOTE

The possibility of an engine fire in flight is extremely remote. The procedure given is general and pilot judgment should be the determining factor for action in such an emergency.

3.17 LOSS OF OIL PRESSURE

Loss of oil pressure may be either partial or complete. A partial loss of oil pressure usually indicates a malfunction in the oil pressure regulating system, and a landing should be made as soon as possible to investigate the cause and prevent engine damage.

A complete loss of oil pressure indication may signify oil exhaustion or may be the result of a faulty gauge. In either case, proceed toward the nearest airport, and be prepared for a forced landing. If the problem is not a pressure gauge malfunction, the engine may stop suddenly. Maintain altitude until such time as a dead stick landing can be accomplished. Don’t change power settings unnecessarily, as this may hasten complete power loss.

Depending on the circumstances, it may be advisable to make an off airport landing while power is still available, particularly if other indications of actual oil pressure loss, such as sudden increases in temperatures, or oil smoke, are apparent, and an airport is not close.

If engine stoppage occurs, proceed with Power Off Landing.

3.19 LOSS OF FUEL PRESSURE

If loss of fuel pressure occurs, turn "ON" the electric fuel pump and check that the fuel selector is on a full tank.

If the problem is not an empty tank, land as soon as practical and have the engine-driven fuel pump and fuel system checked.

3.21 HIGH OIL TEMPERATURE

An abnormally high oil temperature indication may be caused by a low oil level, an obstruction in the oil cooler, damaged or improper baffle seals, a defective gauge, or other causes. Land as soon as practical at an appropriate airport and have the cause investigated.

A steady, rapid rise in oil temperature is a sign of trouble. Land at the nearest airport and let a mechanic investigate the problem. Watch the oil pressure gauge for an accompanying loss of pressure.

3.23 ELECTRICAL FAILURES

Loss of alternator output is detected through zero reading on the ammeter. Before executing the following procedure, insure that the reading is zero and not merely low by actuating an electrically powered device, such as the landing light. If no increase in the ammeter reading is noted, alternator failure can be assumed.

The electrical load should be reduced as much as possible. Check the alternator circuit breakers for a popped circuit.

The next step is to attempt to reset the overvoltage relay. This is accomplished by moving the ALT switch to OFF for one second and then to ON. If the trouble was caused by a momentary overvoltage condition (16.5 volts and up) this procedure should return the ammeter to a normal reading.

If the ammeter continues to indicate "O" output, or if the alternator will not remain reset turn off the ALT switch, maintain minimum electrical load and land as soon as practical. All electrical load is being supplied by the battery.

3.24 ELECTRICAL OVERLOAD (Alternator over 20 amps above known electrical load)

If abnormally high alternator output is observed (more than 20 amps above known electrical load for the operating conditions) it may be caused by a low battery, a battery fault or other abnormal electrical load. If the cause is a low battery, the indication should begin to decrease toward normal within 5 minutes. If the overload condition persists attempt to reduce the load by turning off non-essential equipment. For airplanes with interlocked BATT and ALT switch operation, when the electrical load cannot be reduced turn the ALT switch OFF and land as soon as practical. The battery is the only remaining source of electrical power. Also anticipate complete electrical failure.

For airplanes with separate BATT and ALT switch operations, turn the BATT switch OFF and the ammeter should decrease. Turn the BATT switch ON and continue to monitor the ammeter. If the alternator output does not decrease within 5 minutes, turn the BATT switch OFF and land as soon as practical. All electrical loads are being supplied by the alternator.

NOTE

Due to higher voltage and radio frequency noise, operation with the ALT switch ON and the BATT switch OFF should be made only when required by an electrical failure.

3.25 SPIN RECOVERY

Intentional spins are prohibited in this airplane. If a spin is inadvertently entered, immediately move the throttle to idle and the ailerons to neutral.

Full rudder should then be applied opposite to the direction of rotation followed by control wheel full forward. When the rotation stops, neutralize the rudder and ease back on the control wheel as required to smoothly regain a level flight attitude.

3.27 OPEN DOOR

The cabin door on the Cherokee is double latched, so the chances of its springing open in flight at both the top and side are remote. However, should you forget the upper latch, or not fully engage the side latch, the door may spring partially open. This will usually happen at takeoff or soon afterward. A partially open door will not affect normal flight characteristics, and a normal landing can be made with the door open.

If both upper and side latches are open, the door will trail slightly open, and airspeed will be reduced slightly.

To close the door in flight, slow the airplane to 89 KIAS, close the cabin vents and open the storm window. If the top latch is open, latch it. If the side latch is open, pull on the arm rest while moving the latch handle to the latched position. If both latches are open, close the side latch then the top latch.

3.28 CARBURETOR ICING

Under certain moist atmospheric conditions at temperatures of -5°C to 20°C, it is possible for ice to form in the induction system, even in summer weather. This is due to the high air velocity through the carburetor venturi and the absorption of heat from this air by vaporization of the fuel.

To avoid this, carburetor preheat is provided to replace the heat lost by vaporization. Carburetor heat should be full on when carburetor ice is encountered. Adjust mixture for maximum smoothness.

3.29 ENGINE ROUGHNESS

Engine roughness is usually due to carburetor icing which is indicated by a drop in RPM, and may be accompanied by a slight loss of airspeed or altitude. If too much ice is allowed to accumulate, restoration of full power may not be possible; therefore, prompt action is required.

Turn carburetor heat on (See Note). RPM will decrease slightly and roughness will increase. Wait for a decrease in engine roughness or an increase in RPM, indicating ice removal. If no change in approximately one minute, return the carburetor heat to “OFF.”

If the engine is still rough, adjust the mixture for maximum smoothness. The engine will run rough if too rich or too lean. The electric fuel pump should be switched to “ON” and the fuel selector switched to the other tank to see if fuel contamination is the problem. Check the engine gauges for abnormal readings. If any gauge readings are abnormal, proceed accordingly. Move the magneto switch to “L” then to “R,” then back to “BOTH.” If operation is satisfactory on either magneto, proceed on that magneto at reduced power, with mixture full “RICH,” to a landing at the first available airport.

If roughness persists, prepare for a precautionary landing at pilot’s discretion.

NOTE

Partial carburetor heat may be worse than no heat at all, since it may melt part of the ice, which will refreeze in the intake system. When using carburetor heat, therefore, always use full heat, and when ice is removed return the control to the full cold position.

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NORMAL PROCEDURES

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SECTION 4

NORMAL PROCEDURES

4.1 GENERAL

This section describes the recommended procedures for the conduct of normal operations for the Cherokee Warrior II. All of the required (FAA regulations) procedures and those necessary for operation of the airplane as determined by the operating and design features of the airplane are presented.

Normal procedures associated with those optional systems and equipment which require handbook supplements are provided by Section 9 (Supplements).

These procedures are provided to present a source of reference and review and to supply information on procedures which are not the same for all aircraft. Pilots should familiarize themselves with the procedures given in this section in order to become proficient in the normal operations of the airplane.

The first portion of this section consists of a short form check list which supplies an action sequence for normal operations with little emphasis on the operation of the systems.

The remainder of the section is devoted to amplified normal procedures which provide detailed information and explanations of the procedures and how to perform them. This portion of the section is not intended for use as an in-flight reference due to the lengthly explanations. The short form check list should be used for this purpose.

4.3 AIRSPEEDS FOR SAFE OPERATIONS

The following airspeeds are those which are significant to operation of the airplane. These figures are for standard airplanes flown at gross weight under standard conditions at sea level.

Performance for a specific airplane may vary from published figures depending upon the equipment installed, the condition of the engine, airplane and equipment, atmospheric conditions and piloting technique.

(a)	Best Rate of Climb Speed	79 KIAS
(b)	Best Angle of Climb Speed	63 KIAS
(c)	Turbulent Air Operating Speed (See Subsection 2.3)	111 KIAS
(d)	Maximum Flap Speed	103 KIAS
(e)	Landing Final Approach Speed (Flaps 40°)	63 KIAS
(f)	Maximum Demonstrated Crosswind Velocity	17 KTS

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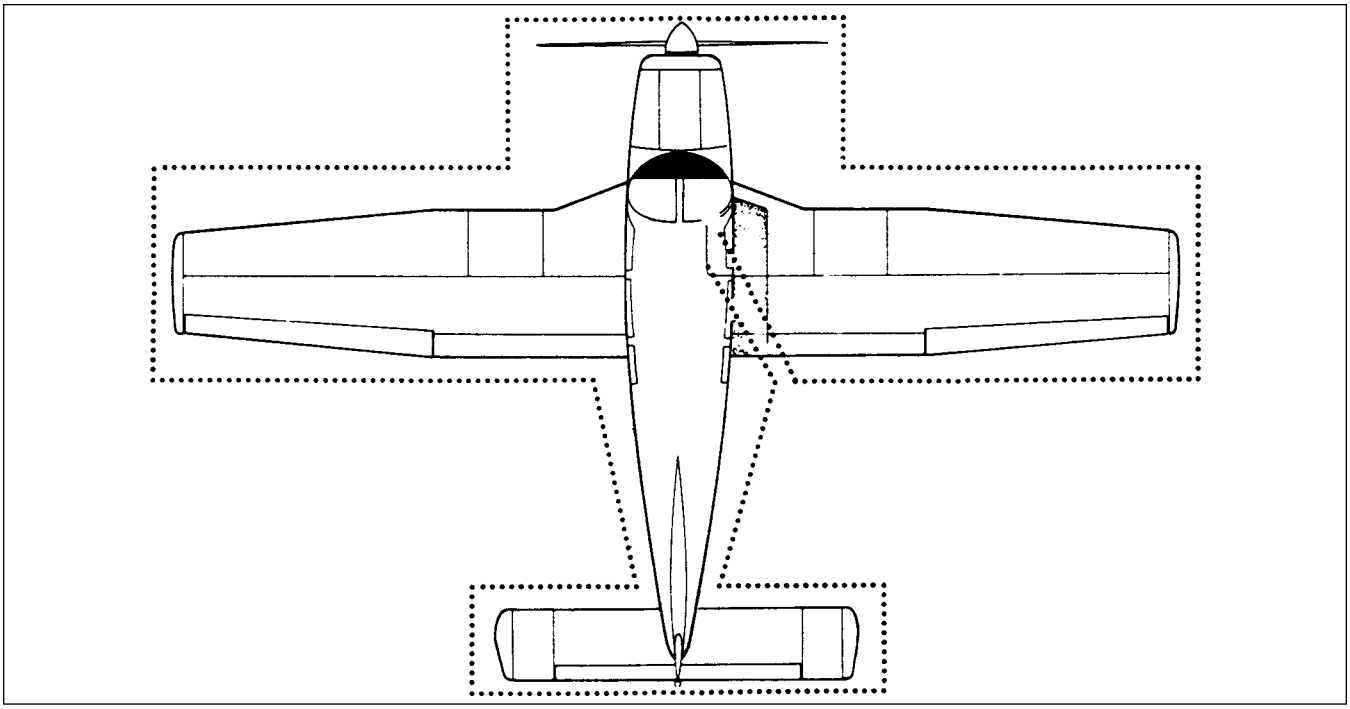

WALK-AROUND

Figure 4-1

4.5 NORMAL PROCEDURES CHECK LIST

PREFLIGHT CHECK

Control wheel .....release belts
Avionics .....OFF
Master switch .....ON
Fuel quantity gauges .....check
Master switch.....OFF
Ignition.....OFF
Exterior .....check for damage
Control surfaces .....check for interference - free of ice, snow, frost
Hinges .....check for interference
Wings.....free of ice, snow, frost
Stall warning .....check
Tie down and chocks .....remove
Navigation lights.....check
Fuel tanks .....check supply visually - secure caps
Fuel tank sumps .....drain
Fuel vents .....open
Main gear struts .....proper inflation (4.50 in.)
Tires .....check
Brake blocks .....check
Pitot head.....remove cover - holes clear
Pitot heat.....check (if installed)

Windshield .....clean
Propeller and spinner .....check
Fuel and oil .....check for leaks
Oil.....check level
Dipstick.....properly seated
Cowling.....secure
Inspection covers .....secure
Nose wheel tire .....check
Nose gear strut .....proper inflation (3.25 in.)
Air inlets .....clear
Alternator belt .....check tension
Tow bar and control locks.....stow
Baggage.....stowed properly - secure
Baggage door .....close and secure
Fuel strainer .....drain
Primary flight controls.....proper operation
Cabin door.....close and secure
Required papers .....on board
Seat belts and harness.....fasten/adjust-check inertia reel

BEFORE STARTING ENGINE

Brakes .....set
Carburetor heat .....full OFF
Fuel selector .....desired tank
Radios .....OFF

STARTING ENGINE WHEN COLD

Throttle .....1/4" open
Master switch .....ON
Electric fuel pump .....ON
Mixture .....full RICH
Starter .....engage
Throttle .....adjust
Oil pressure .....check

If engine does not start within 10 sec. prime and repeat starting procedure.

STARTING ENGINE WHEN HOT

Throttle .....1/2" open
Master switch .....ON
Electric fuel pump .....ON
Mixture .....full RICH
Starter .....engage
Throttle .....adjust
Oil pressure .....check

STARTING ENGINE WHEN FLOODED

Throttle .....open full
Master switch .....ON
Electric fuel pump .....OFF
Mixture .....idle cut-off
Starter .....engage
Mixture .....advance
Throttle .....retard
Oil pressure .....check

STARTING WITH EXTERNAL POWER SOURCE

Master switch .....OFF
All electrical equipment .....OFF
Terminals .....connect
External power plug .....insert in fuselage
Proceed with normal start
Throttle .....lowest possible RPM
External power plug .....disconnect from fuselage
Master switch .....ON - check ammeter
Oil pressure .....check

WARM-UP

Throttle .....800 to 1200 RPM

TAXIING

Chocks .....removed
Taxi area .....clear
Throttle .....apply slowly
Brakes .....check
Steering .....check

GROUND CHECK

Throttle .....2000 RPM
Magnetos .....max. drop 175 RPM
-max. diff. 50 RPM
Vacuum .....5.0" Hg. ± .1
Oil temp .....check
Oil pressure .....check
Air conditioner .....check
Annunciator panel .....press-to-test
Carburetor heat .....check
Engine is warm for takeoff when throttle can be opened without engine faltering.
Electric fuel pump .....OFF
Fuel pressure .....check
Throttle .....retard

BEFORE TAKEOFF

Master switch .....ON
Flight instruments .....check
Fuel selector.....proper tank
Electric fuel pump .....ON
Engine gauges.....check
Carburetor heat .....OFF
Seat backs.....erect
Mixture .....set
Primer .....locked
Belts/harness.....fastened/adjusted
Empty seats .....seat belts snugly fastened
Flaps.....set
Trim tab .....set
Controls.....free
Doors .....latched
Air conditioner.....OFF

TAKEOFF

NORMAL

Flaps.....set
Tab .....set
Accelerate to 45 to 55 KIAS
Control wheel.....back pressure to rotate to climb attitude

SHORT FIELD, OBSTACLE CLEARANCE

Flaps.....25 ° (second notch)
Accelerate to 52 KIAS
Control wheel.....back pressure to rotate to climb attitude
Maintain 52 KIAS until obstacle clearance
Accelerate to 79 KIAS after obstacle is cleared
Flaps .....retract slowly

SHORT FIELD, NO OBSTACLE

Flaps.....UP
Accelerate to 50 KIAS
Control wheel.....back pressure to rotate to climb attitude
After breaking ground accelerate to best rate of climb speed 79 KIAS

SOFT FIELD, OBSTACLE CLEARANCE

Flaps.....25° (second notch)
Accelerate and lift off nose gear as soon as possible. Lift off at lowest possible airspeed
Accelerate just above ground to 52 KIAS to climb past obstacle height.
Continue climbing while accelerating to best rate of climb speed, 79 KIAS
Flaps .....slowly retract

SOFT FIELD, NO OBSTACLE

Flaps.....25° (second notch)
Accelerate and lift off nose gear as soon as possible. Lift off at lowest possible airspeed
Accelerate just above ground to best rate of climb speed, 79 KIAS
Flaps .....slowly retract

CLIMB

Best rate (flaps up).....79 KIAS
Best angle (flaps up) .....63 KIAS
En route.....87 KIAS
Electric fuel pump.....OFF at desired altitude

CRUISING

Reference performance charts and Lycoming Operator's Manual.
Normal max power 75%
Power .....set per power table
Mixture .....adjust

DESCENT

NORMAL

Throttle .....2500 rpm
Airspeed.....126 KIAS
Mixture.....rich
Carburetor heat .....ON if required

DESCENT

POWER OFF

Carburetor heat .....ON if required
Throttle.....closed
Airspeed.....as required
Mixture .....as required
Power .....verify with throttle every 30 seconds

APPROACH AND LANDING

Fuel selector.....proper tank
Seat backs .....erect
Belts/harness .....fasten/adjust
Electric fuel pump .....ON
Mixture .....set
Flaps .....set - 103 KIAS max
Air conditioner.....OFF
Trim to 70 KIAS
Final approach speed (flaps 40°) .....63 KIAS

STOPPING ENGINE

Flaps.....retract
Electric fuel pump.....OFF
Air conditioner.....OFF
Radios .....OFF
Throttle .....full aft
Mixture .....idle cut-off
Magnetos.....OFF
Master switch.....OFF

PARKING

Parking brake .....set
Control wheel .....secured with belts
Flaps .....full up
Wheel chocks.....in place
Tie downs .....secure

4.7 AMPLIFED NORMAL PROCEDURES (GENERAL)

The following paragraphs are provided to supply detailed information and explanations of the normal procedures necessary for the safe operation of the airplane.

4.9 PREFLIGHT CHECK

The airplane should be given a thorough preflight and walk-around check. The preflight should include a check of the airplane's operational status, computation of weight and C.G. Iimits, takeoff distance and in-flight performance. A weather briefing should be obtained for the intended flight path, and any other factors relating to a safe flight should be checked before takeoff.

CAUTION

The flap position should be noted before boarding the aircraft. The flaps must be placed in the "UP" position before they will lock and support weight on the step.

Upon entering the cockpit, release the seat belts securing the control wheel. Turn off all avionics equipment. Turn the master switch "ON" and check the fuel quantity gauges for sufficient fuel. After the fuel quantity check is made turn the master switch "OFF' and check that the ignition switch is "OFF."

To begin the exterior walk-around, check for external damage and operational interference of the control surfaces or hinges. Insure that the wings and control surfaces are free of snow, ice, frost or any other foreign materials.

An operational check of the stall warning system and navigation lights should now be made. Turn the master switch "ON." Lift the detector while checking to determine if the horn is actuated and check that the navigation lights are illuminated. The master switch should be returned to the "OFF" position after the checks are complete.

A visual check of the fuel tank quantity should be performed. Remove the filler cap from each tank and visually check the supply and color. Be sure to secure the caps properly after the check is complete.

The fuel system sumps and strainer should be drained daily prior to the first flight and after refueling to avoid the accumulation of contaminants such as water or sediment. Each fuel tank is equipped with an individual quick drain located at the lower inboard rear comer of the tank. The fuel strainer is equipped with a quick drain located on the front lower corner of the firewall. Each of the fuel tank sumps should be drained first. Then the fuel strainer should be drained twice, once with the fuel selector valve on each tank. Each time fuel is drained, sufficient fuel should be allowed to flow to ensure removal of contaminants. This fuel should be collected in a suitable container, examined for contaminants, and then discarded.

CAUTION

When draining any amount of fuel, care should be taken to ensure that no fire hazard exists before starting the engine.

After draining, each quick drain should be checked to make sure it has closed completely and is not leaking.

Check all of the fuel tank vents to make sure they are open.

Next, a complete check of the landing gear. Check the main gear shock struts for proper inflation. There should be 4.50 inches of strut exposure under a normal static load. The nose gear should be checked for 3.25 inches of strut exposure. Check all tires for cuts and wear and insure proper inflation. Make a visual check of the brake blocks for wear or damage.

Remove the cover from the pitot head on the underside of the wing.Check the pitot head to make sure the holes are open and clear of obstructions. If pitot heat is installed, then turn the master switch "ON" and the pitot heat switch "ON". With the pitot heat switch "ON", check that the pitot head is hot to the touch. The pitot heat switch and the master switch should be returned to the "OFF" position after the check is complete.

Don’t forget to clean and check the windshield.

The propeller and spinner should be checked for defects or nicks.

Lift the cowling and check for any obvious fuel or oil leaks. Check the oil level. Make sure that the dipstick has properly seated after checking. Secure the cowling and check the inspection covers.

Check the air inlets for foreign matter and the alternator belt for proper tension.

Stow the tow bar and check the baggage for proper storage and security. The baggage compartment doors should be closed and secure.

Upon entering the aircraft, ascertain that all prirnary flight controls operate properly. Close and secure the cabin door and check that all the required papers are in order and in the airplane.

Fasten the seat belts and shoulder harness and check the function of the inertia reel by pulling sharply on the strap. Fasten seat belts on empty seats.

NOTE

If the fixed shoulder harness (non-inertia reel type) is installed, it must be connected to the seat belt and adjusted to allow proper accessibility to all controls including fuel selector, flaps, trim. etc., while maintaining adequate restraint for the occupant.

If the inertia reel type shoulder harness is installed, a pull test of its locking restraint feature should be performed.

4.11 BEFORE STARTING ENGINE

Before starting the engine the brakes should be set “ON” and the carburetor heat lever moved to the full OFF position. The fuel selector should then be moved to the desired tank. Check to make sure that all the radios are OFF.

4.13 STARTING ENGINE

(a) Starting Engine When Cold

Open the throttle lever approximately 1/4 inch. Turn "ON" the master switch and the electric fuel pump.

Move the mixture control to full "RICH" and engage the starter by rotating the magneto switch clockwise. When the engine fires, release the magneto switch, and move the throttle to the desired setting.

If the engine does not fire within five to ten seconds, disengage the starter, prime the engine and repeat the starting procedure.

(b) Starting Engine When Hot

Open the throttle approximately 1/2 inch. Turn "ON" the master switch and the electric fuel pump. Move the mixture control lever to full RICH and engage the starter by rotating the magneto switch clockwise. When the engine fires, release the magneto switch and move the throttle to the desired setting.

(c) Starting Engine When Flooded

The throttle lever should be full "OPEN." Turn "ON" the master switch and turn "OFF" the electric fuel pump. Move the mixture control lever to idle cut-off and engage the starter by rotating the magneto switch clockwise. When the engine fires, release the magneto switch, advance the mixture and retard the throttle.

(d) Starting Engine With External Power Source

An optional feature called the Piper External Power (PEP) allows the operator to use an external battery to crank the engine without having to gain access to the airplane's battery.

Turn the master switch OFF and turn all electrical equipment OFF. Connect the RED lead of the PEP kit jumper cable to the POSITIVE (+) terminal of an external 12-volt battery and the BLACK lead to the NEGATIVE (-) terminal. Insert the plug of the jumper cable into the socket located on the fuselage. Note that when the plug is inserted, the electrical system is ON. Proceed with the normal starting technique.

After the engine has started, reduce power to the lowest possible RPM, to reduce sparking, and disconnect the jumper cable from the aircraft. Turn the master switch ON and check the alternator ammeter for an indication of output. DO NOT ATTEMPT FLIGHT IF THERE IS NO INDICATION OF ALTERNATOR OUTPUT.

NOTE

For all normal operations using the PEP jumper cables, the master switch should be OFF, but it is possible to use the ship's battery in parallel by turning the master switch ON. This will give longer cranking capabilities, but will not increase the amperage.

CAUTION

Care should be exercised because if the ship's battery has been depleted, the external power supply can be reduced to the level of the ship's battery. This can be tested by turning the master switch ON momentarily while the starter is engaged. If cranking speed increases, the ship's battery is at a higher level than the external power supply.

When the engine is firing evenly, advance the throttle to 800 RPM. If oil pressure is not indicated within thirty seconds, stop the engine and determine the trouble. In cold weather it will take a few seconds longer to get an oil pressure indication. If the engine has failed to start, refer to the Lycoming Operating Handbook, Engine Troubles and Their Remedies.

Starter manufacturers recommend that cranking periods be limited to thirty seconds with a two minute rest between cranking periods. Longer cranking periods will shorten the life of the starter.

4.15 WARM-UP

Warm-up the engine at 800 to 1200 RPM for not more than two minutes in warm weather and four minutes in cold. Avoid prolonged idling at low RPM, as this practice may result in fouled spark plugs.

Takeoff may be made as soon as the ground check is completed, provided that the throttle may be opened fully without backfiring or skipping, and without a reduction in engine oil pressure.

Do not operate the engine at high RPM when running up or taxiing over ground containing loose stones, gravel or any loose material that may cause damage to the propeller blades.

4.17 TAXIING

Before attempting to taxi the airplane, ground personnel should be instructed and approved by a qualified person authorized by the owner. Ascertain that the propeller back blast and taxi areas are clear.

Power should be applied slowly to start the taxi roll. Taxi a few feet forward and apply the brakes to determine their effectiveness. While taxiing, make slight turns to ascertain the effectiveness of the steering.

Observe wing clearances when taxiing near buildings or other stationary objects. If possible, station an observer outside the airplane.

Avoid holes and ruts when taxiing over uneven ground.

Do not operate the engine at high RPM when running up or taxiing over ground containing loose stones, gravel or any loose material that may cause damage to the propeller blades.

4.19 GROUND CHECK

The magnetos should be checked at 2000 RPM. Drop off on either magneto should not exceed 175 RPM and the difference between the magnetos should not exceed 50 RPM. Operation on one magneto should not exceed 10 seconds.

Check the vacuum gauge; the indicator should read 5.0” ± .1” Hg at 2000 RPM.

Check the annunciator panel lights with the press-to-test button. Also check the air conditioner.

Carburetor heat should also be checked prior to takeoff to be sure the control is operating properly and to clear any ice which may have formed during taxiing. Avoid prolonged ground operation with carburetor heat “ON” as the air is unfiltered.

The electric fuel pump should be turned “OFF” after starting or during warm-up to make sure that the engine driven pump is operating. Prior to takeoff the electric pump should be turned ON again to prevent loss of power during takeoff should the engine driven pump fail. Check both oil temperature and oil pressure. The temperature may be low for some time if the engine is being run for the first time of the day. The engine is warm enough for takeoff when the throttle can be opened without the engine faltering.

4.21 BEFORE TAKEOFF

All aspects of each particular takeoff should be considered prior to executing the takeoff procedure.

Turn "ON" the master switch and check and set all of the flight instruments as required. Check the fuel selector to make sure it is on the proper tank (fullest). Turn "ON" the electric fuel pump and check the engine gauges. The carburetor heat should be in the "OFF" position.

All seat backs should be erect and the seat belts and shoulder harness fastened. Fasten the seat belts snugly around the empty seats.

NOTE

If the fixed shoulder harness (non-inertia reel type) is installed, it must be connected to the seat belt and adjusted to allow proper accessibility to all controls including fuel selector, flaps, trim, etc., while maintaining adequate restraint for the occupant.

If the inertia reel type shoulder harness is installed, a pull test of its locking restraint feature should be performed.

The mixture should be set and the primer checked to insure that it is locked.

NOTE

The mixture should be set FULL RICH except a minimum amount of leaning is permitted for smooth engine operation when taking off at high elevation.

Exercise and set the flaps and trim tab. Insure proper flight control movement and response. All doors should be properly secured and latched. On air conditioned models, the air conditioner must be "OFF" to insure normal takeoff performance.

4.23 TAKEOFF

The normal takeoff technique is conventional. The tab should be set slightly aft of neutral, with the exact setting determined by the loading of the airplane. Allow the airplane to accelerate to 45 to 55 KIAS depending on the weight of the aircraft and ease back on the control wheel to rotate to climb attitude. Premature raising of the nose or raising it to an excessive angle will result in a delayed takeoff. After takeoff, let the airplane accelerate to the desired climb speed by lowering the nose slightly.

Takeoffs are normally made with flaps up; however, for short field takeoffs and for takeoffs under difficult conditions such as deep grass or a soft surface, total distances can be reduced appreciably by lowering the flaps to 25° and rotating at lower airspeed.

A short field takeoff with an obstacle clearance is accomplished by first lowering the flaps to 25°. Apply full power before brake release and accelerate to 52 KIAS and rotate. Maintain 52 KIAS until obstacle clearance is attained. After the obstacle has been cleared, accelerate to 79 KIAS and then slowly retract the flaps.

A short field takeoff with no obstacle is accomplished with no flaps and applying full power before brake release, lift off at 50 KIAS and accelerate to best rate of climb speed, 79 KIAS.

Takeoff from a soft field with an obstacle clearance requires the use of 25° flaps. Accelerate the airplane and lift the nose gear off as soon as possible and lift off at the lowest possible airspeed. Accelerate just above the ground to 52 KIAS to climb past obstacle clearance height. Continue climbing while-accelerating to the best rate of climb speed, 79 KIAS and slowly retract the flaps.

For a soft field takeoff without an obstacle to clear, extend the flaps 25°, accelerate the airplane and lift the nose gear off as soon as possible. Lift off at the lowest possible airspeed. Accelerate just above the ground to the best rate of climb speed, 79 KIAS and retract the flaps while climbing out.

4.25 CLIMB

The best rate of climb at gross weight will be obtained at 79 KIAS. The best angle of climb may be obtained at 63 KIAS. At lighter than gross weight these speeds are reduced somewhat. For climbing en route, a speed of 87 KIAS is recommended. This will produce better forward speed and increased visibility over the nose during the climb.

When reaching the desired altitude, the electric fuel pump may be turned off.

4.27 CRUISING

The cruising speed is determined by many factors, including power setting, altitude, temperature, loading and equipment installed in the airplane.

The normal maximum cruising power is 75% of the rated horsepower of the engine. Airspeeds which may be obtained at various altitudes and power settings can be determined from the performance graphs provided by Section 5.

Use of the mixture control in cruising flight reduces fuel consumption significantly, especially at higher altitudes, and reduces lead deposits when the alternate fuels are used. During letdown and low power flight operations, it may be necessary to lean because of excessively rich mixture. The mixture should be leaned during cruising operation when 75% power or less is being used. If any doubt exists as to the amount of power being used, the mixture should be in the FULL RICH position for all operations. Always enrich the mixture before increasing power settings.

To lean the mixture for best power cruise performance, disengage the friction adjustment lever and place the mixture control full forward and set the throttle slightly below (approximately 35 RPM) the desired cruise power setting and lean the mixture to peak RPM. Adjust the throttle, if necessary, for final RPM setting.

To lean for best economy cruise performance, disengage the friction adjustment lever and place the mixture control full forward and set the throttle to obtain the desired power setting for the conditions in Section 5. Gradually lean the mixture control until the engine operation becomes rough or until engine power rapidly diminishes as noted by an undesirable decrease in airspeed or engine RPM. When either condition occurs, enrich the mixture sufficiently to obtain a smooth and evenly firing engine or to regain most of the lost airspeed or engine RPM.

CAUTION

Prolonged operation at powers above 75% with a leaned mixture can result in engine damage. While establishing Best Economy Cruise Mixture, below 6,000 feet, care must be taken not to remain in the range above 75% power more than 15 seconds while leaning. Above 6,000 feet the engine is incapable of generating more than 75%.

Always remember that the electric fuel pump should be turned "ON" before switching tanks, and should be left on for a short period thereafter. In order to keep the airplane in best lateral trim during cruising flight, the fuel should be used alternately from each tank. It is recommended that one tank be used for one hour after takeoff, then the other tank be used for two hours; then return to the first tank, which will have approximately one and one half hours of fuel remaining if the tanks were full at takeoff. The second tank will contain approximately one half hour of fuel. Do not run tanks completely dry in flight. The electric fuel pump should be normally "OFF" so that any malfunction of the engine driven fuel pump is immediately apparent. If signs of fuel starvation should occur at any time during flight, fuel exhaustion should be suspected, at which time the fuel selector should be immediately positioned to the other tank and the electric fuel pump switched to the "ON" position.

4.28 DESCENT

NORMAL

To achieve the performance on Figure 5-25 the power on descent must be used. The throttle should be set for 2500 RPM, mixture full rich and maintain an airspeed of 126 KIAS. In case carburetor ice is encountered apply full carburetor heat.

POWER OFF

If a prolonged power off descent is to be made, apply full carburetor heat prior to power reduction if icing conditions are suspected. Throttle should be retarded and mixture control leaned as required. Power response should be verified approximately every 30 seconds by partially opening and then closing the throttle (clearing the engine). When leveling off enrichen mixture, set power as required and select carburetor heat off unless carburetor icing conditions are suspected.

4.29 APPROACH AND LANDING

Check to insure the fuel selector is on the proper (fullest) tank and that the seat backs are erect. The seat belts and shoulder harness should be fastened and the inertia reel checked.

NOTE

If the fixed shoulder harness (non-inertia reel type) is installed, it must be connected to the seat belt and adjusted to allow proper accessibility to all controls including fuel selector, flaps, trim, etc., while maintaining adequate restraint for the occupant.

If the inertia reel type shoulder harness is installed, a pull test of its locking restraint feature should be performed.

Turn the electric fuel pump "ON" and turn the air conditioner "OFF." The mixture should be set in the full "RICH" position.

The airplane should be trimmed to an initial-approach speed of about 70 KIAS with a final-approach speed of 63 KIAS with flaps extended to 40°. The flaps can be lowered at speeds up to 103 KIAS, if desired.

The mixture control should be kept in full "RICH" position to insure maximum acceleration if it should be necessary to open the throttle again. Carburetor heat should not be applied unless there is an indication of carburetor icing, since the use of carburetor heat causes a reduction in power which may be critical in case of a go-around. Full throttle operation with carburetor heat on can cause detonation.

The amount of flap used during landings and the speed of the aircraft at contact with the runway should be varied according to the landing surface and conditions of wind and airplane loading. It is generally good practice to contact the ground at the minimum possible safe speed consistent with existing conditions.

Normally, the best technique for short and slow landings is to use full flap and enough power to maintain the desired airspeed and approach flight path. Mixture should be full "RICH," fuel on the fullest tank, and electric fuel pump "ON." Reduce the speed during the flareout and contact the ground close to the stalling speed. After ground contact hold the nose wheel off as long as possible. As the airplane slows down, gently lower the nose and apply the brakes. Braking is most effective when flaps are raised and back pressure is applied to the control wheel, putting most of the aircraft weight on the main wheels. In high wind conditions, particularly in strong crosswinds, it may be desirable to approach the ground at higher than normal speeds with partial or no flaps.

4.31 STOPPING ENGINE

At the pilot's discretion, the flaps should be raised and the electric fuel pump turned "OFF." The air conditioner and radios should be turned "OFF," and the engine stopped by disengaging the friction adjustment lever and pulling the mixture control back to idle cut-off. The throttle should be left full aft to avoid engine vibration while stopping. Then the magneto and master switches must be turned "OFF."

NOTE

When alternate fuels are used, the engine should be run up to 1200 RPM for one minute prior to shutdown to clean out any unburned fuel.

NOTE

The flaps must be placed in the "UP" position for the flap step to support weight. Passengers should be cautioned accordingly.

4.33 PARKING

If necessary, the airplane should be moved on the ground with the aid of the nose wheel tow bar provided with each airplane and secured behind the rear seats. The aileron and stabilator controls should be secured by looping the safety belt through the control wheel and pulling it snug. The flaps are locked when in the "UP" position and should be left retracted.

Tie downs can be secured to rings provided under each wing and to the tail skid. The rudder is held in position by its connections to the nose wheel steering and normally does not have to be secured.

4.35 STALLS

The stall characteristics are conventional. An approaching stall is indicated by a stall warning horn which is activated between five and ten KTS above stall speed. Mild airframe buffeting and gentle pitching may also precede the stall.

The gross weight stalling speed with power off and full flaps is 44 KIAS. With the flaps up this speed is increased. Loss of altitude during stalls varies from 100 to 275 feet, depending on configuration and power.

NOTE

The stall warning system is inoperative with the master switch "OFF."

During preflight, the stall warning system should be checked by turning the master switch "ON," lifting the detector and checking to determine if the horn is actuated. The master switch should be returned to the "OFF" position after the check is complete.

4.37 TURBULENT AIR OPERATION

In keeping with good operating practice used in all aircraft, it is recommended that when turbulent air is encountered or expected, the airspeed be reduced to maneuvering speed to reduce the structural loads caused by gusts and to allow for inadvertent speed build-ups which may occur as a result of the turbulence or of distractions caused by the conditions. (See Subsection 2.3)

4.39 WEIGHT AND BALANCE

It is the responsibility of the owner and pilot to determine that the airplane remains within the allowable weight vs. center of gravity envelope while in flight.

For weight and balance data, refer to Section 6 (Weight and Balance).

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SECTION 5

PERFORMANCE

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SECTION 5

PERFORMANCE

5.1 GENERAL

All of the required (FAA regulations) and complementary performance information applicable to this aircraft is provided by this section.

Performance information associated with those optional systems and equipment which require handbook supplements is provided by Section 9 (Supplements).

5.3 INTRODUCTION TO PERFORMANCE AND FLIGHT PLANNING

The performance information presented in this section is based on measured Flight Test Data corrected to I.C.A.O. standard day conditions and analytically expanded for the various parameters of weight, altitude, temperature, etc.

The performance charts are unfactored and do not make any allowance for varying degrees of pilot proficiency or mechanical deterioration of the aircraft. This performance, however, can be duplicated by following the stated procedures in a properly maintained airplane.

Effects of conditions not considered on the charts must be evaluated by the pilot, such as the effect of soft or grass runway surface on takeoff and landing performance, or the effect of winds aloft on cruise and range performance. Endurance can be grossly affected by improper leaning procedures, and inflight fuel flow and quantity checks are recommended.

REMEMBER! To get chart performance, follow the chart procedures.

The information provided by paragraph 5.5 (Flight Planning Example) outlines a detailed flight plan using the performance charts in this section. Each chart includes its own example to show how it is used.

WARNING

Performance information derived by extrapolation beyond the limits shown on the charts should not be used for flight planning purposes .

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5.5 FLIGHT PLANNING EXAMPLE

(a) Aircraft Loading

The first step in planning our flight is to calculate the airplane weight and center of gravity by utilizing the information provided by Section 6 (Weight and Balance) of this handbook.

The basic empty weight for the airplane as delivered from the factory has been entered in Figure 6-5. If any alterations to the airplane have been made effecting weight and balance, reference to the aircraft logbook and Weight and Balance Record (Figure 6-7) should be made to determine the current basic empty weight of the airplane.

Make use of the Weight and Balance Loading Form (Figure 6-11) and the C.G. Range and Weight graph (Figure 6-15) to determine the total weight of the airplane and the center of gravity position.

After proper utilization of the information provided we have found the following weights for consideration in our flight planning example.

The landing weight cannot be determined until the weight of the fuel to be used has been established [refer to item (g)(1)].

(1)	Basic Empty Weight	1391 lbs.
(2)	Occupants (4 x 170 lbs.)	680 lbs.
(3)	Baggage and Cargo	50 lbs.
(4)	Fuel (6 lb/gal x 30)	180 lbs.
(5)	Takeoff Weight	2316 lbs.
(6)	Landing Weight	
	(a)(5) minus (g)(1), (2316 lbs. minus 134.4 lbs.)	2181.6 lbs

Our takeoff weight is below the maximum of 2325 lbs. and our weight and balance calculations have determined our C.G. position within the approved limits.

(b) Takeoff and Landing

Now that we have determined our aircraft loading, we must consider all aspects of our takeoff and landing.

All of the existing conditions at the departure and destination airport must be acquired, evaluated and maintained throughout the flight.

Apply the departure airport conditions and takeoff weight to the appropriate Takeoff Performance graph (Figures 5-5 and 5-6 or 5-7 and 5-8) to determine the length of runway necessary for the takeoff and/or the barrier distance.

The landing distance calculations are performed in the same manner using the existing conditions at the destination airport and, when established, the landing weight.

The conditions and calculations for our example flight are listed below. The takeoff and landing distances required for our example flight have fallen well below the available runway lengths.

		Departure Airport	Destination Airport
(1)	Pressure Altitude	1500 ft.	2500 ft.
(2)	Temperature	80°F (27°C)	75°F (24°C)
(3)	Wind Component	15 KTS (Headwind)	0 KTS
(4)	Runway Length Available	4800 ft.	7600 ft.
(5)	Runway Required	2100 ft.*	1190**

NOTE

The remainder of the performance charts used in this flight plan example assume a no wind condition. The effect of winds aloft must be considered by the pilot when computing climb, cruise and descent performance.

(c) Climb

The next step in our flight plan is to determine the necessary climb segment components.

The desired cruise pressure altitude and corresponding cruise outside air temperature values are the first variables to be considered in determining the climb components from the Time, Distance, and Fuel to Climb graph (Figure 5-13). After the time, distance and fuel for the cruise pressure altitude and outside air temperature values have been established, apply the existing conditions at the departure field to graph (Figure 5-13). Now, subtract the values obtained from the graph for the field of departure conditions from those for the cruise pressure altitude.

The remaining values are the true fuel, distance and time components for the climb segment of the flight plan corrected for field pressure altitude and temperature.

The following values were determined from the above instructions in our flight planning example.

(1)	Cruise Pressure Altitude	5000 ft.
(2)	Cruise OAT	60°F (16°C)
(3)	Time to Climb (10.0 min. minus 2.5 min.)	7.5 min.***
(4)	Distance to Climb (13.5 miles minus 3.5 miles)	10.0 miles***
(5)	Fuel to Climb (2 gal. minus .5 gal.)	1.5 gal.***

* reference Figure 5-6
** reference Figure 5-29
*** reference Figure 5-13

(d) Descent

The descent data will be determined prior to the cruise data to provide the descent distance for establishing the total cruise distance.

Utilizing the cruise pressure altitude and OAT we determine the basic time, distance and fuel for descent (Figure 5-25). These figures must be adjusted for the field pressure altitude and temperature at the destination airport. To find the necessary adjustment values, use the existing pressure altitude and temperature conditions at the destination airport as variables to find the time, distance and fuel values from the graph (Figure 5-25). Now, subtract the values obtained from the field conditions from the values obtained from the cruise conditions to find the true time, distance and fuel values needed for the flight plan.

The values obtained by proper utilization of the graphs for the descent segment of our example are shown below.

(1)	Time to Descend (6.5 min. minus 3.5 min.)	3.0 min.*
(2)	Distance to Descend (14 miles minus 7.5 miles)	6.5 miles*
(3)	Fuel to Descend (1.0 gal. minus.5 gal.)	.5 gal.*

(e) Cruise

Using the total distance to be traveled during the flight, subtract the previously calculated distance to climb and distance to descend to establish the total cruise distance. Refer to the appropriate Avco Lycoming Operator's Manual when selecting the cruise power setting. The established pressure altitude and temperature values and the selected cruise power should now be utilized to determine the true airspeed from the Cruise Performance graph (Figures 5-15 through 5-18).

Calculate the cruise fuel consumption for the cruise power setting from the information provided by the Avco Lycoming Operator's Manual.

The cruise time is found by dividing the cruise distance by the cruise speed and the cruise fuel is found by multiplying the cruise fuel consumption by the cruise time.

The cruise calculations established for the cruise segment of our flight planning example are as follows:

(1)	Total Distance	300 miles
(2)	Cruise Distance (e)(1)minus (c)(4) minus (d)(2), (300 minus 10 miles minus 6.5 miles)	283.5 miles
(3)	Cruise Power, Best Economy Mixture	75% rated power (2645 RPM)
(4)	Cruise Speed	118 KTS TAS**
(5)	Cruise Fuel Consumption	8.5 GPH
(6)	Cruise Time (e)(2) divided by (c)(4), (283.5 miles divided by 118 KTS)	2.40 hrs.
(7)	Cruise Fuel (e)(5) multiplied by (e)(6), (8.5 GPH multiplied by 2.40 hrs.)	20.4 gal.

* reference Figure 5-25
** reference Figure 5-17

(f) Total Flight Time

The total flight time is determined by adding the time to climb, the time to descend and the cruise time. Remember! The time values taken from the climb and descent graphs are in minutes and must be converted to hours before adding them to the cruise time.

The following flight time is required for our flight planning example.

(1) Total Flight Time

(c)(3) plus (d)(l) plus (e)(6), (.13 hrs. plus .05 hrs. plus 2.40 hrs.)	2.58 hrs.

(g) Total Fuel Required

Determine the total fuel required by adding the fuel to climb, the fuel to descend and the cruise fuel. When the total fuel (in gallons) is determined, multiply this value by 6 lb/gal to determine the total fuel weight used for the flight.

The total fuel calculations for our example flight plan are shown below.

(1) Total Fuel Required

(c)(5) plus (d)(3) plus (e)(7), (1.5 gal. plus .5 gal. plus 20.4 gal.)	22.4 gal.
(22.4 gal. multiplied by 6 lb/gal.)	134.4 lbs.

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5.7 PERFORMANCE GRAPHS

LIST OF FIGURES

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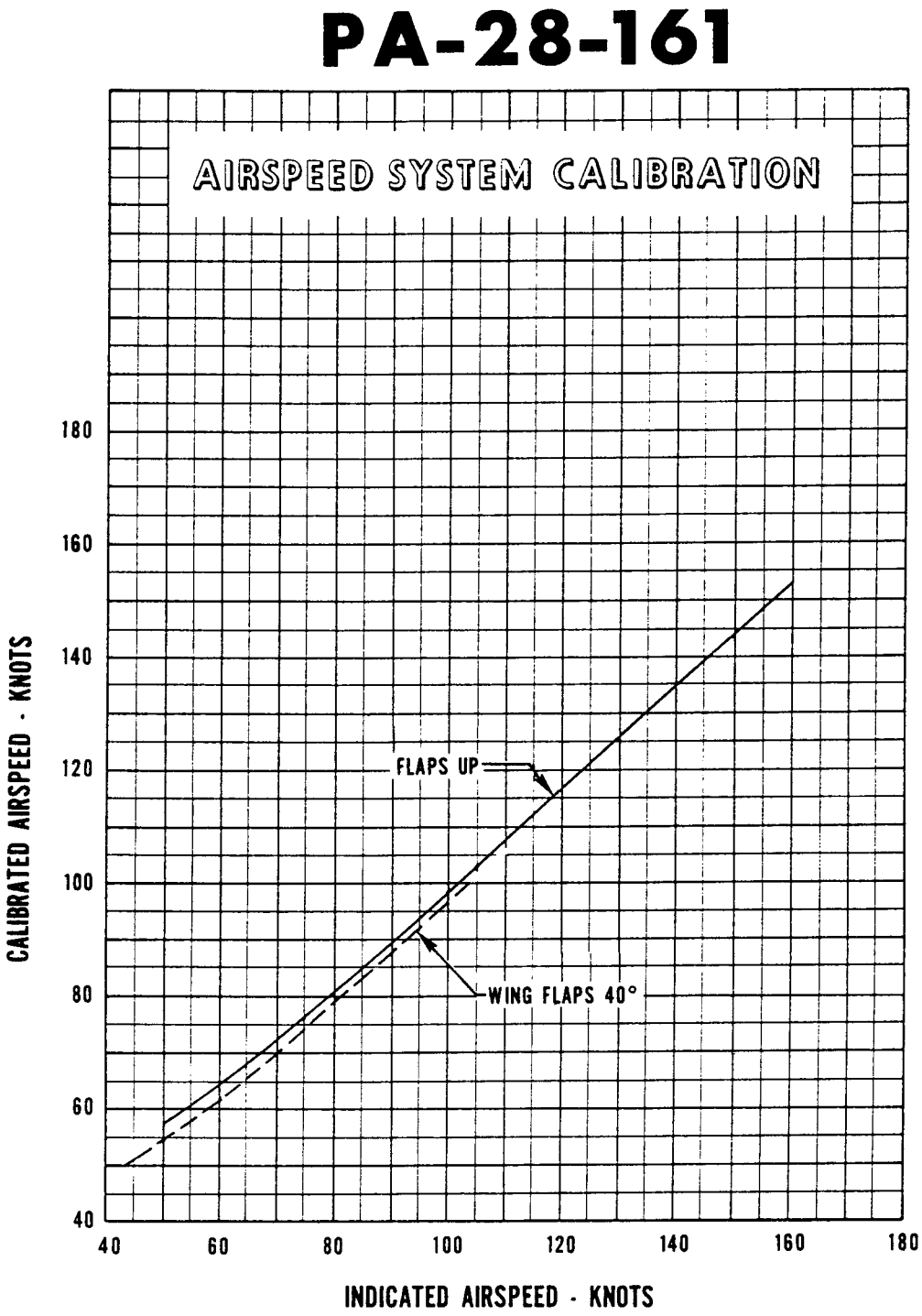
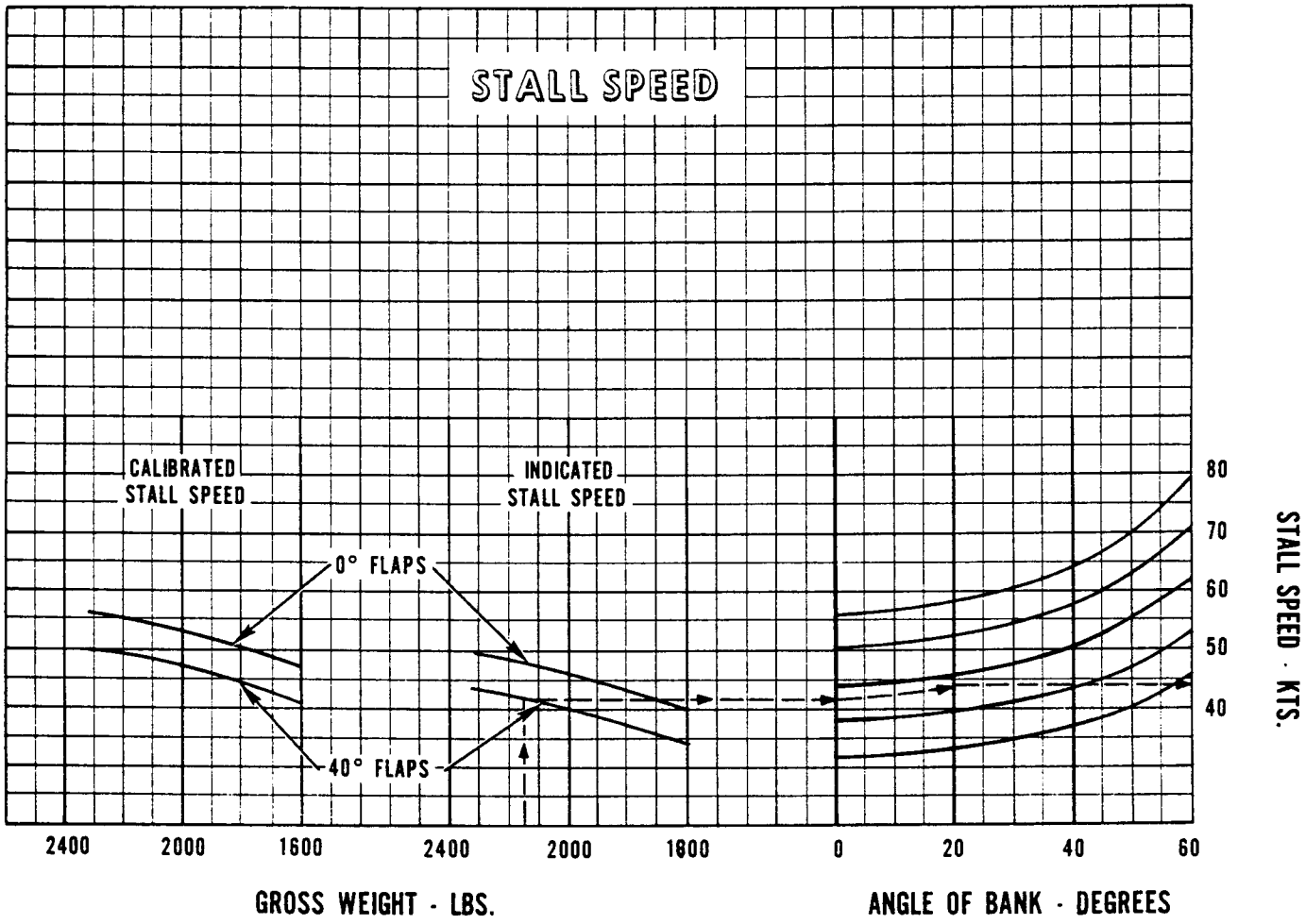


AIRSPEED SYSTEM CALIBRATION

Figure 5-1

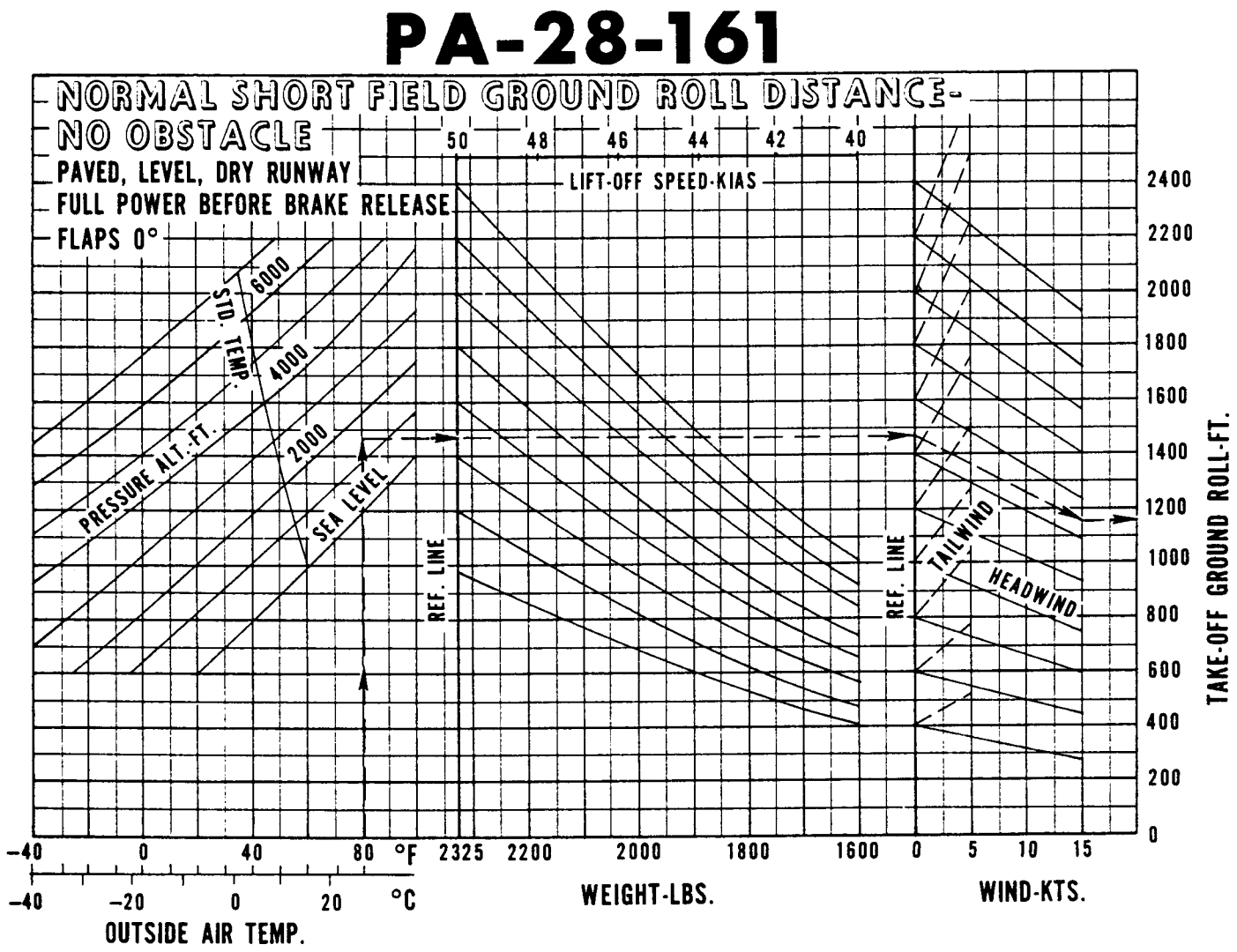
PA-28-161



Example:
Gross weight: 2170 lbs.
Angle of bank: 20°
Flap position: 40°
Stall speed, indicated: 44 KTS

STALL SPEED

Figure 5-3

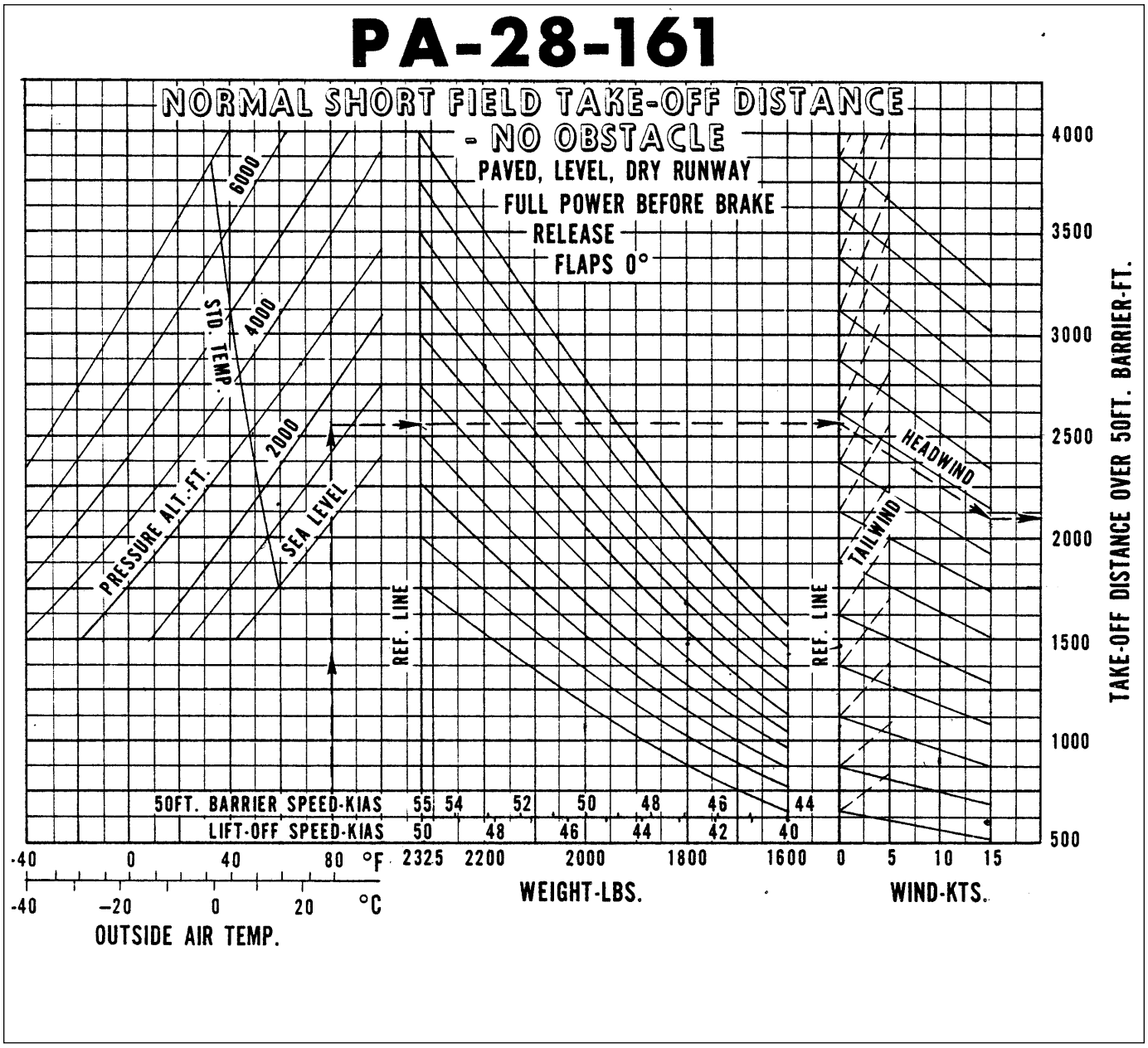


Example:

Departure airport pressure altitude: 1500 ft.
Departure airport temperature: 80°F
Weight: 2325 lbs.
Wind: 15 KTS headwind
Ground roll: 1150 ft.
Lift-off speed: 50 KIAS

NORMAL SHORT FIELD GROUND ROLL DISTANCE - NO OBSTACLE

Figure 5-5



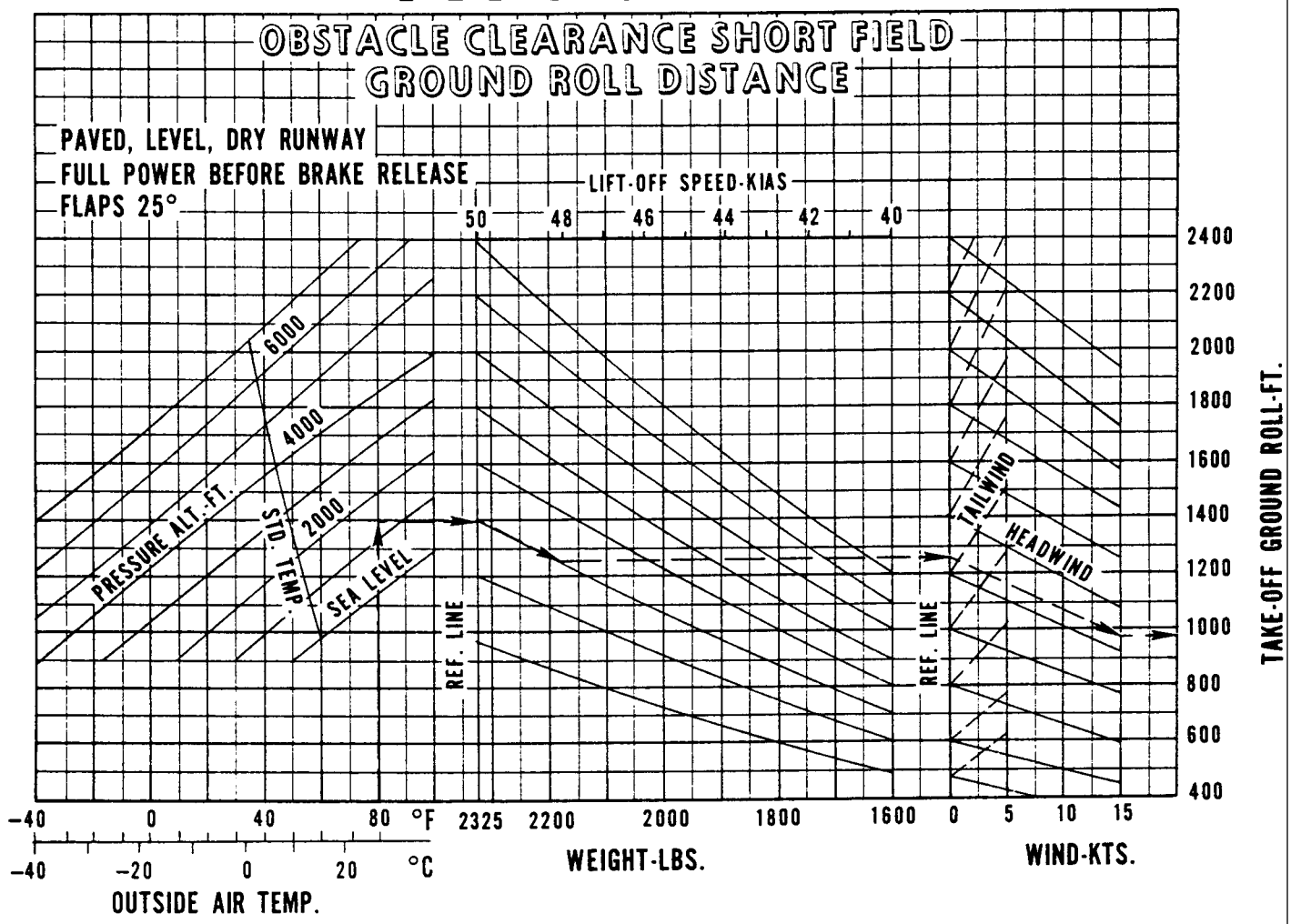
Example:

Departure airport pressure altitude: 1500 ft.
Departure airport temperature: 80°F
Weight: 2325 lbs.
Wind: 15 KTS headwind
Distance over 50 ft. barrier: 2100 ft.
Lift-off speed: 50 KIAS
Barrier speed: 55 KIAS

NORMAL SHORT FIELD TAKEOFF DISTANCE - NO OBSTACLE

Figure 5-6

PA-28-161

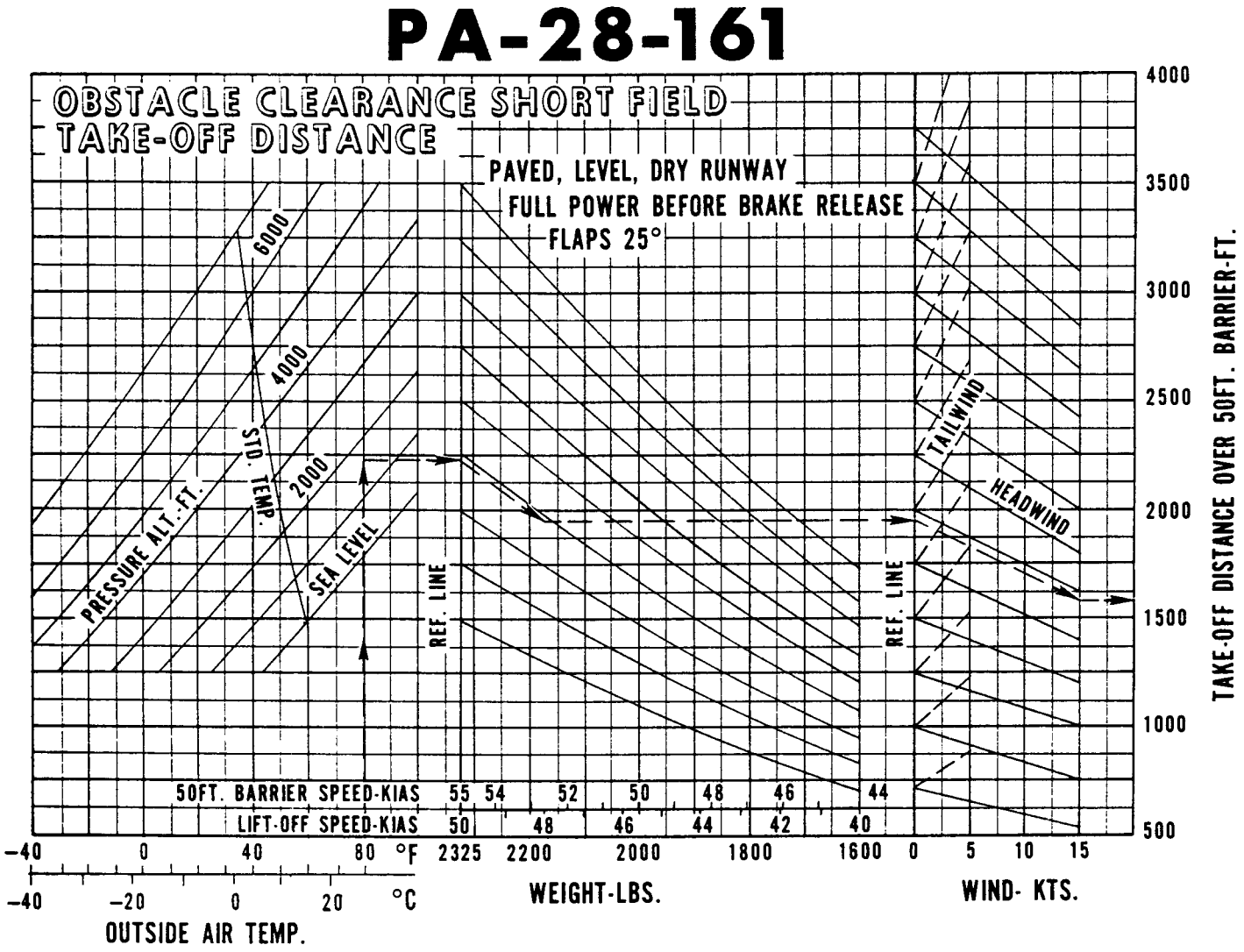


Example:

Departure airport pressure altitude: 1500 ft.
Departure airport temperature: 80°F
Weight: 2175 lbs.
Wind: 15 KTS headwind
Ground roll: 975 ft.
Lift-off speed: 48 KIAS

OBSTACLE CLEARANCE SHORT FIELD GROUND ROLL DISTANCE

Figure 5-7

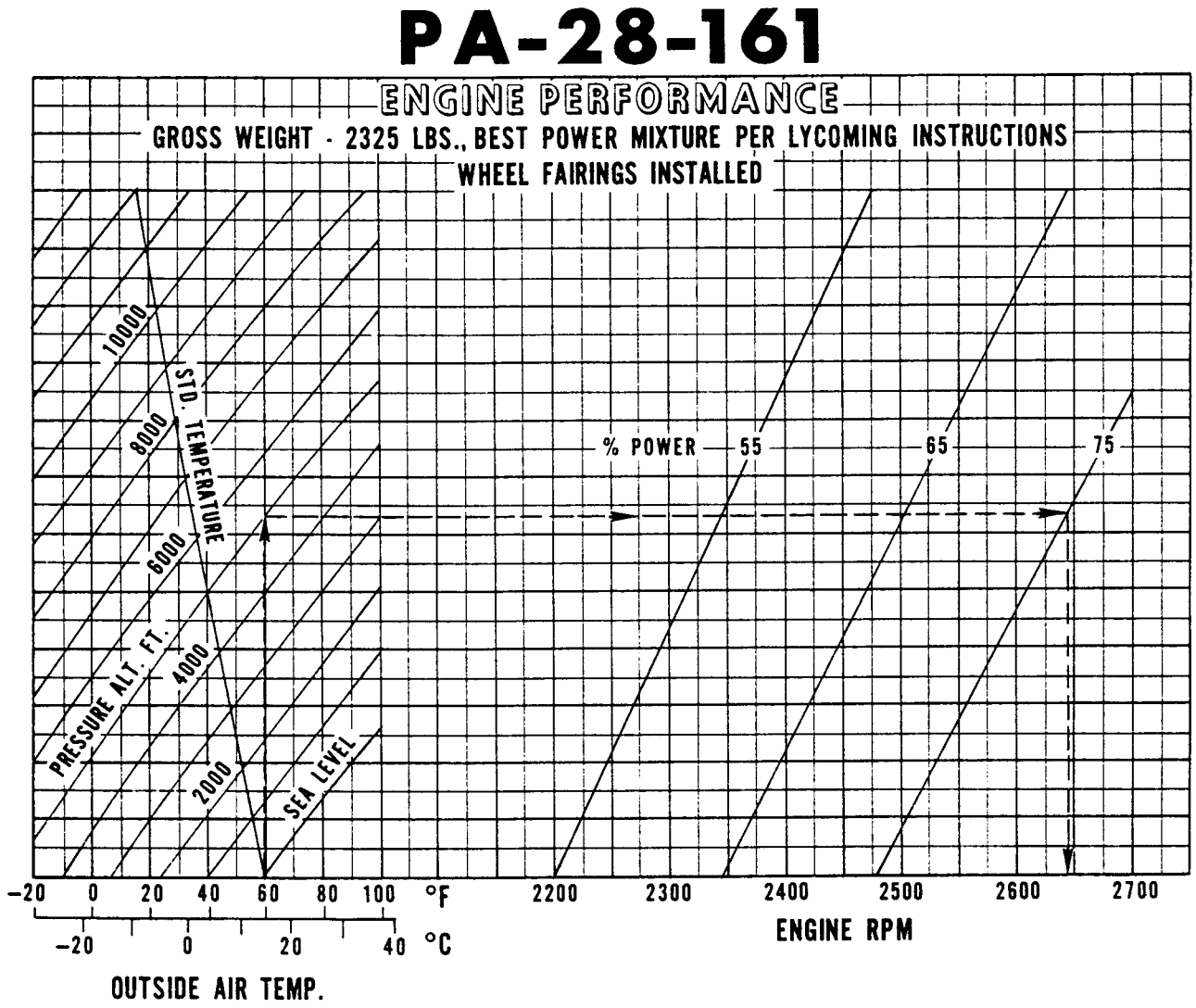


Example:

Departure airport pressure altitude: 1500 ft.
Departure airport temperature: 80°F
Weight: 2175 lbs.
Wind: 15 KTS headwind
Distance over 50 ft. barrier: 1600 ft.
Lift-off speed: 48 KIAS
Barrier speed: 53 KIAS

OBSTACLE CLEARANCE SHORT FIELD TAKEOFF DISTANCE

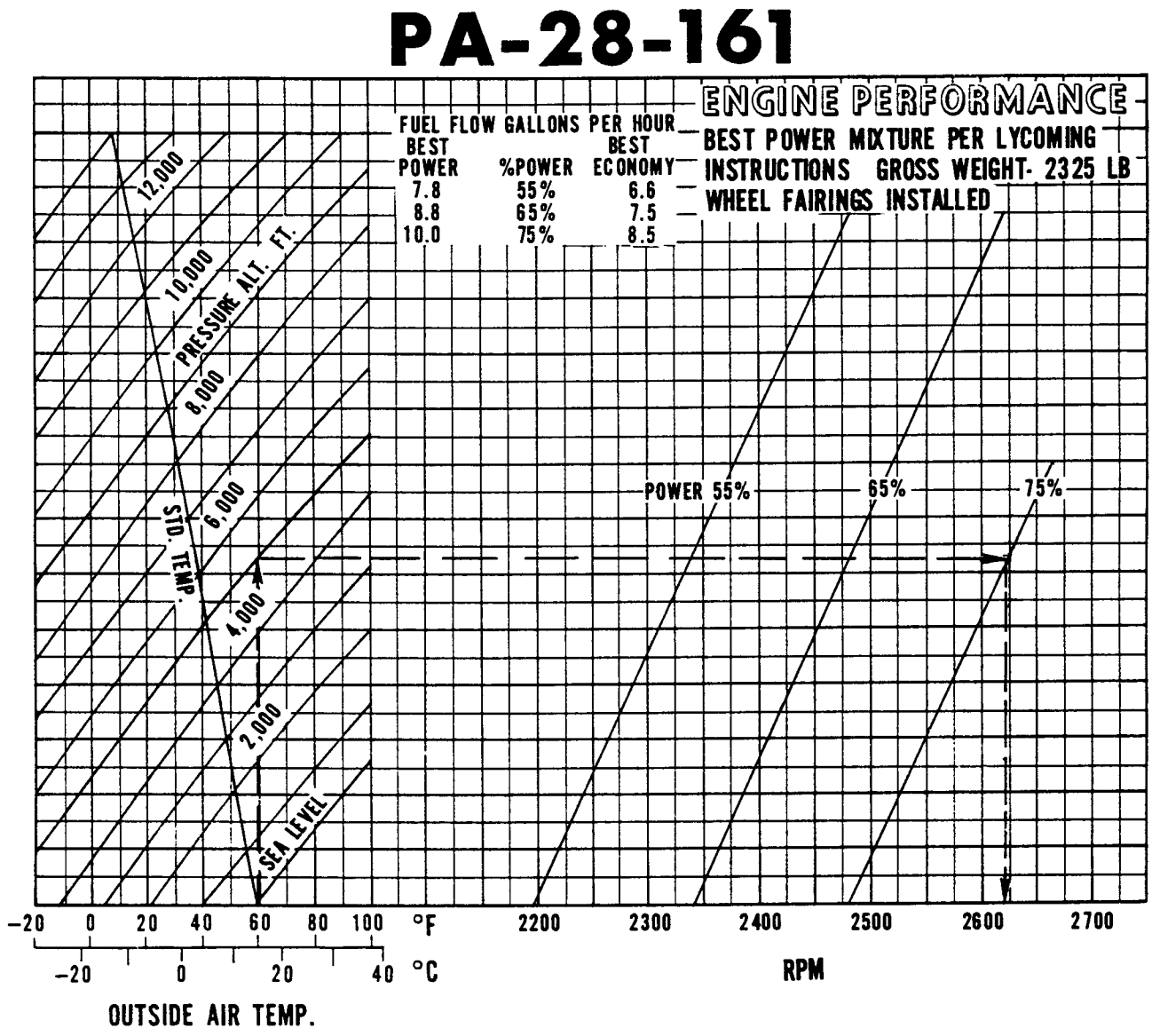
Figure 5-8



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F
Cruise power: 75%
Engine RPM: 2645

ENGINE PERFORMANCE (SERIAL NOS. 28-7716001 THROUGH 7716323)

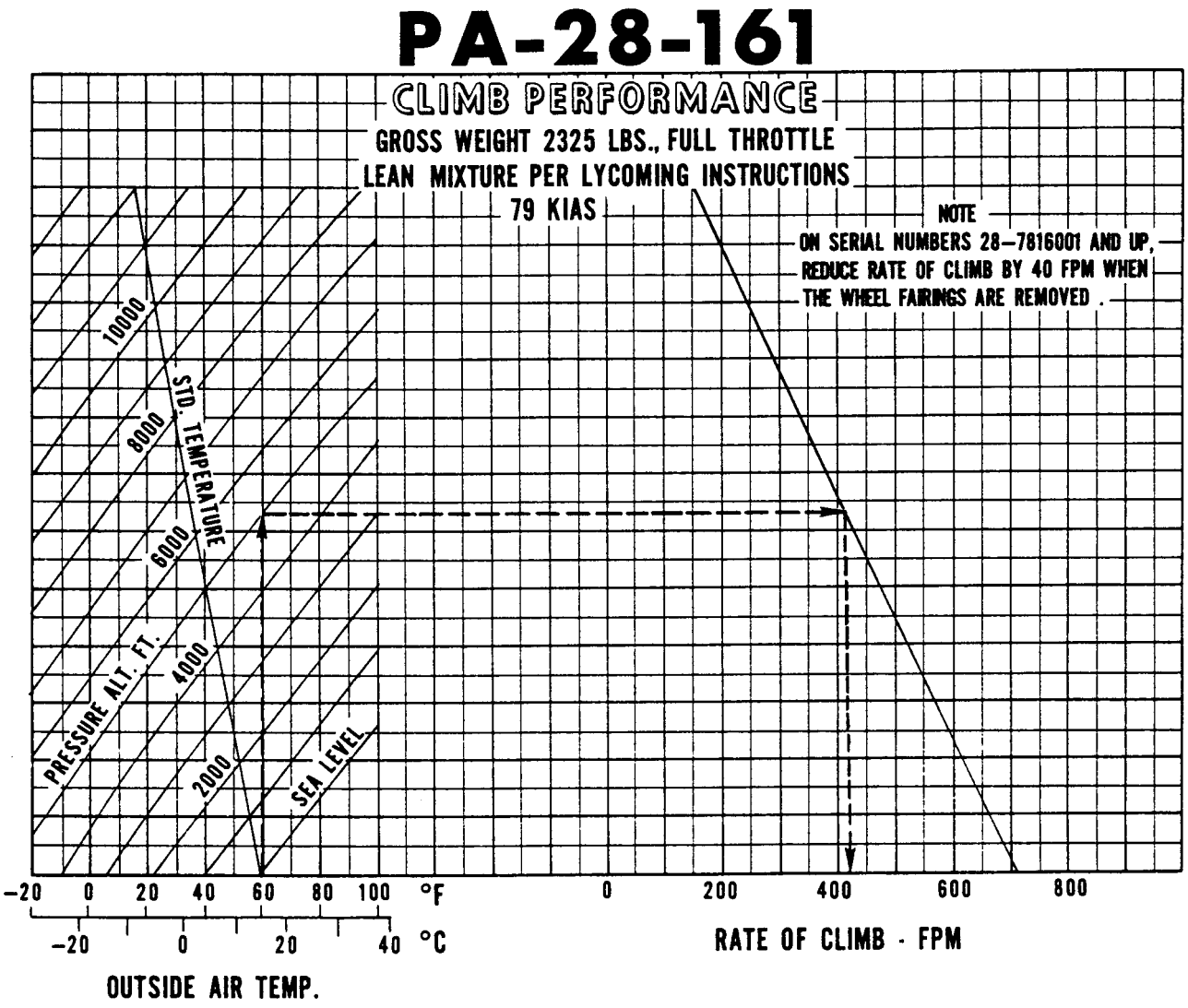
Figure 5-9



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F
Cruise power: 75%
Engine RPM: 2620

ENGINE PERFORMANCE (SERIAL NOS. 28-7816001 AND UP)

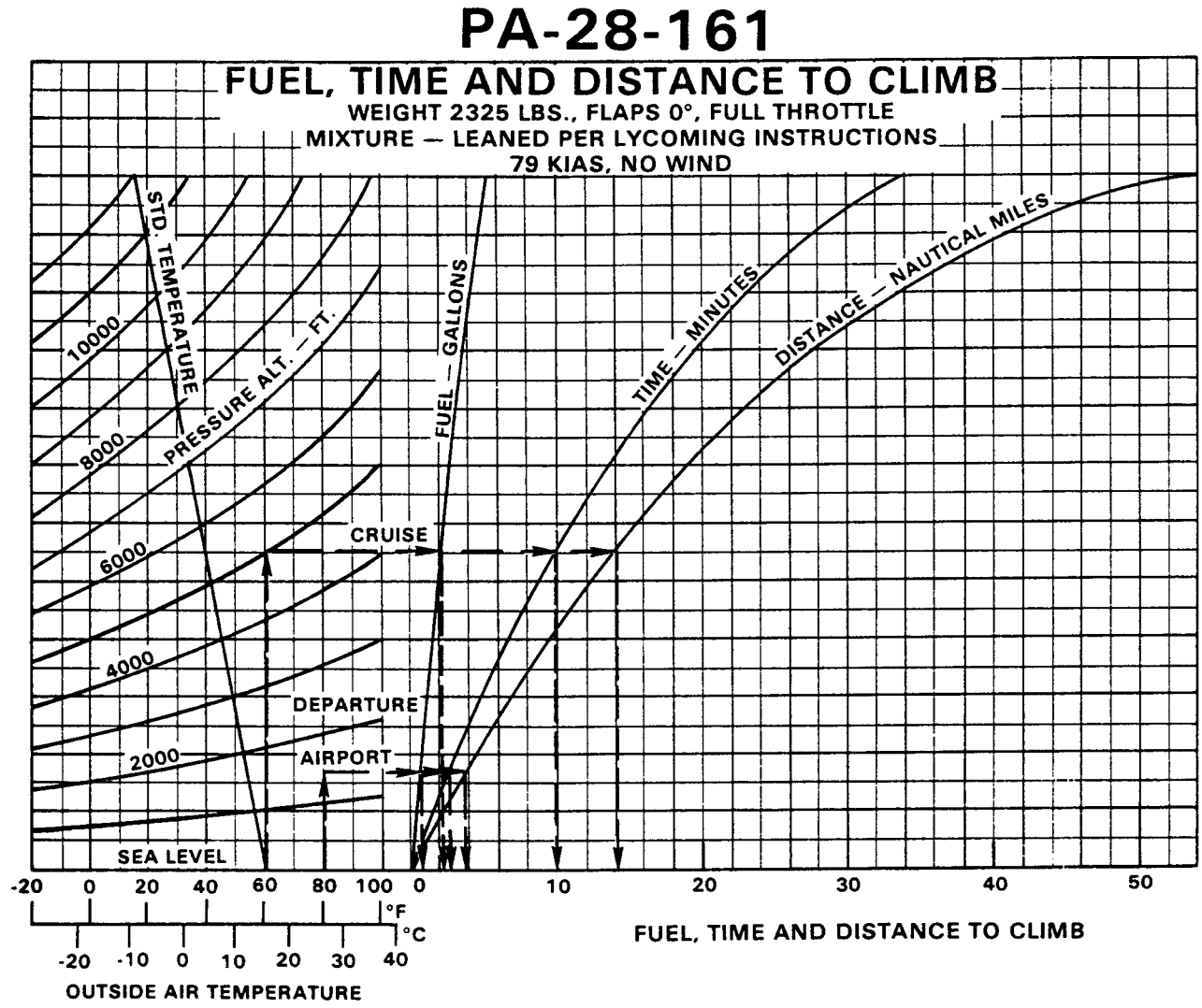
Figure 5-10



Example:
Climb pressure altitude: 5000 ft.
Climb OAT: 60°F
Rate of climb: 420 ft/min.

CLIMB PERFORMANCE

Figure 5-11

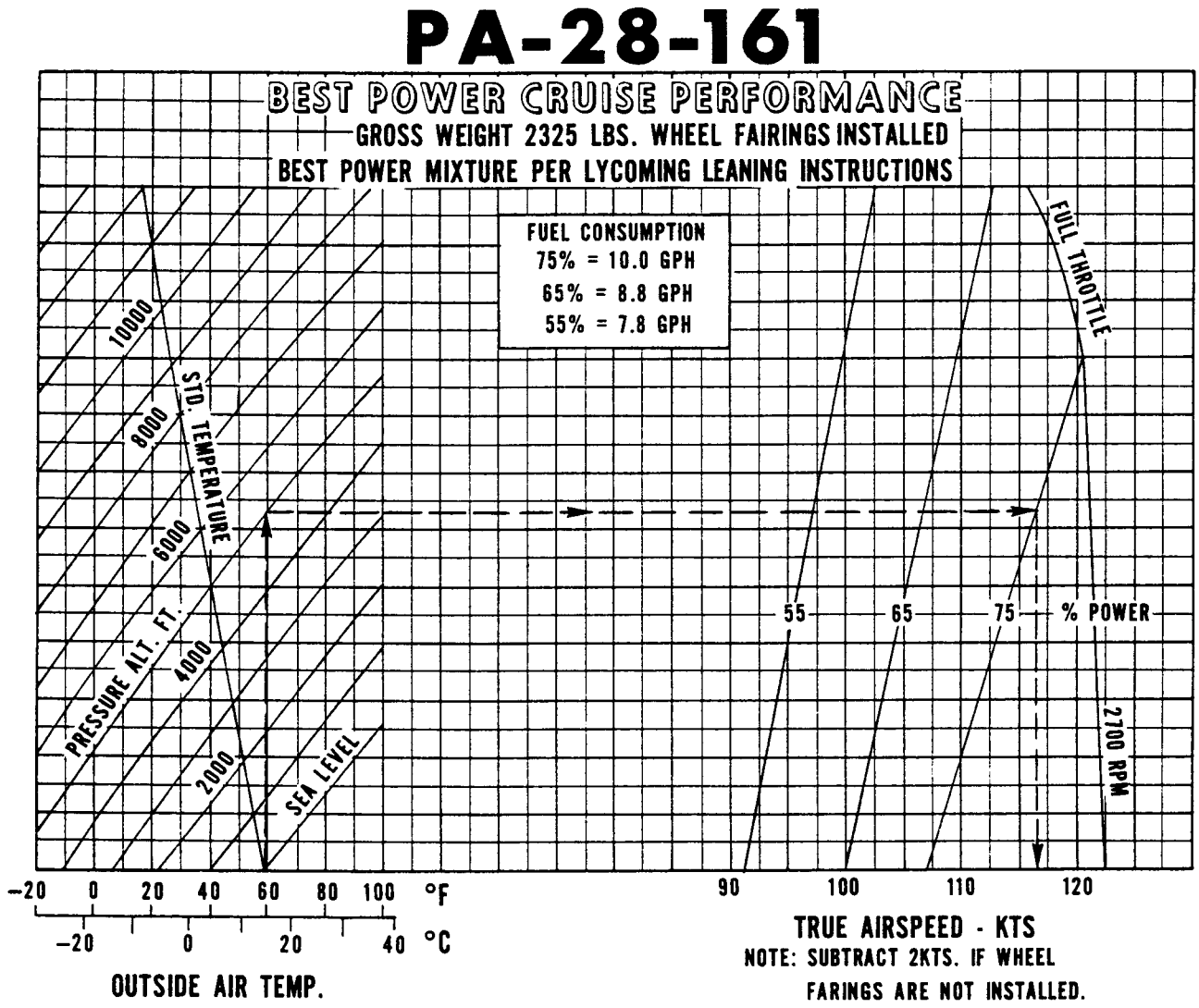


Example:

Departure airport pressure altitude: 1500 ft.
Departure airport temperature: 80°F (27°C)
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F (16°C)
Time to climb (10 min. minus 2.5 min.): 7.5 min.
Distance to climb (13.5 miles minus 3.5 miles): 10 nautical miles
Fuel to climb (2 gal. minus .5 gal.): 1.5 gal.

FUEL, TIME AND DISTANCE TO CLIMB

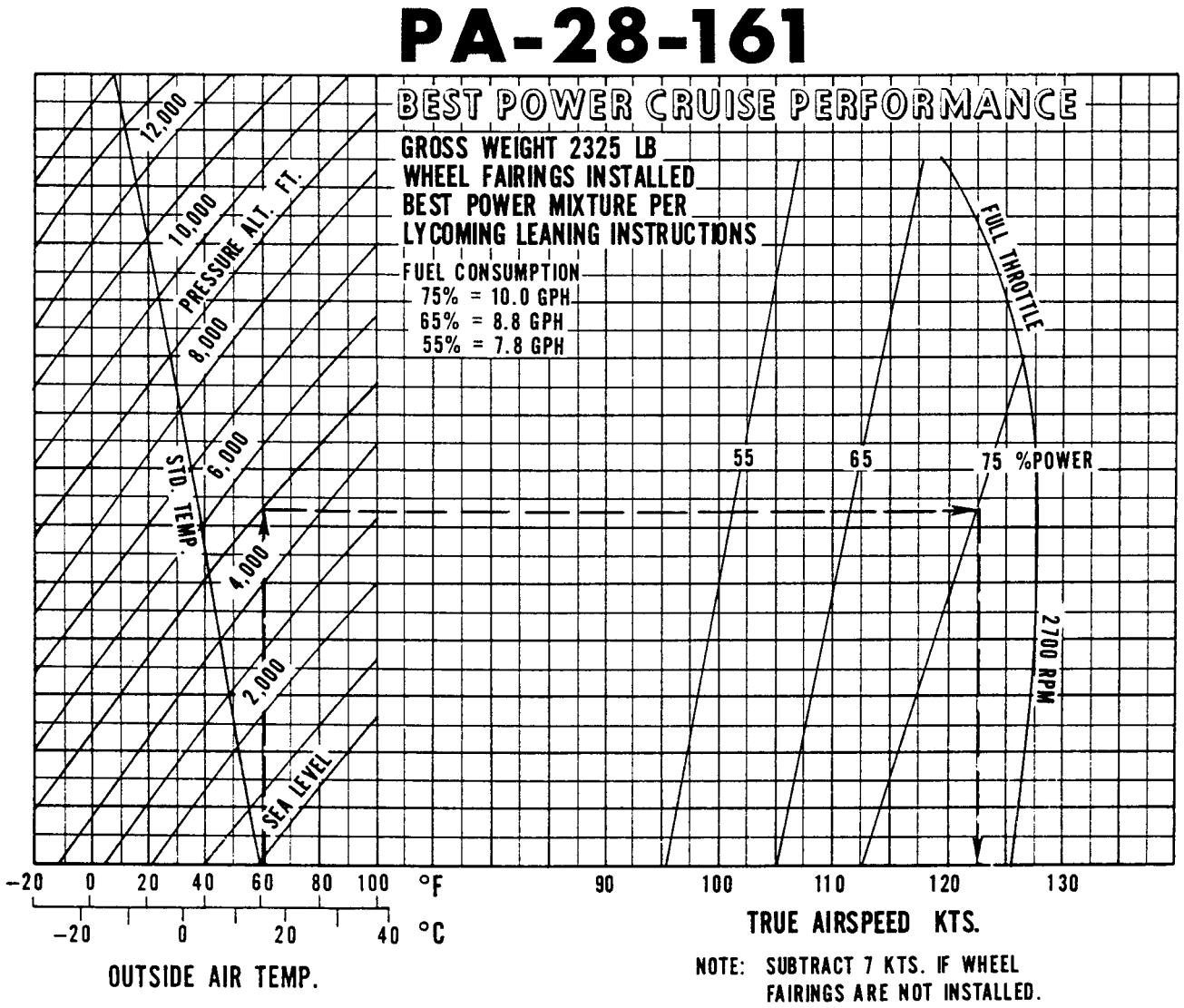
Figure 5-13



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F
Cruise power: 75% best power mixture
Cruise speed: 116.5 KTS TAS

BEST POWER CRUISE PERFORMANCE (SERIAL NOS. 28-7716001 THROUGH 7716323)

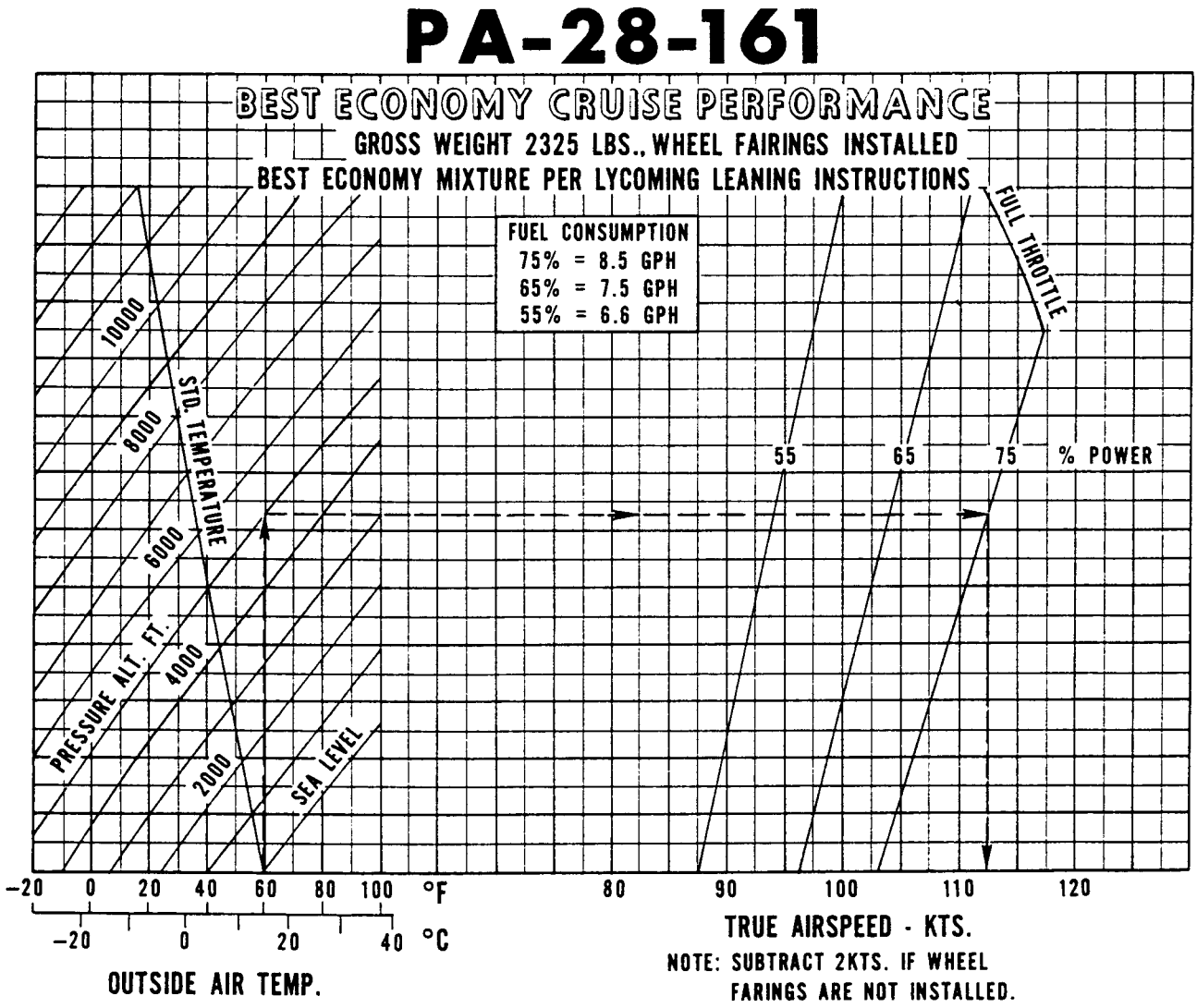
Figure 5-15



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F
Cruise power: 75% best power mixture
Cruise speed: 122.5 KTS TAS

BEST POWER CRUISE PERFORMANCE (SERIAL NOS. 28-7816001 AND UP)

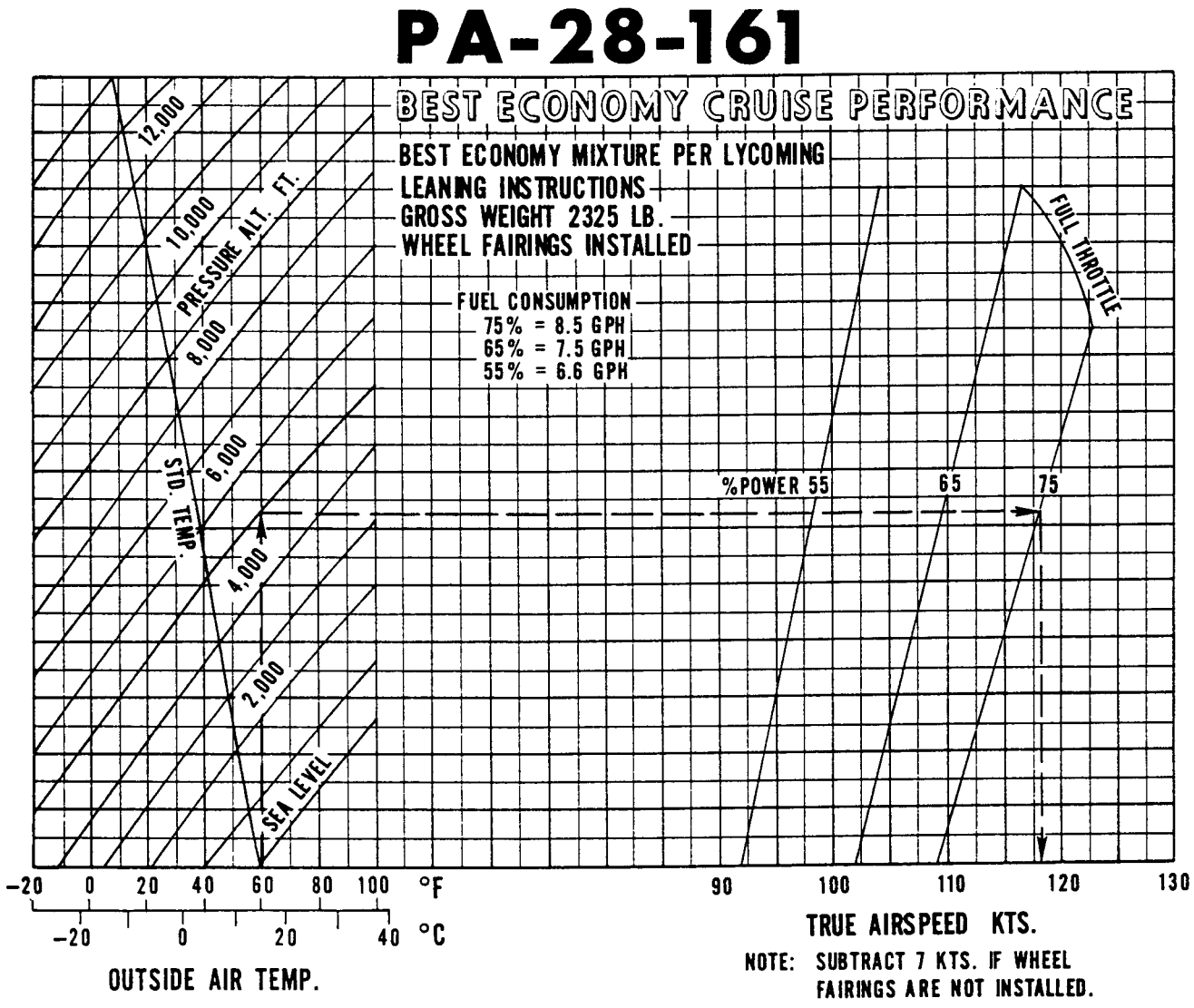
Figure 5-16



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F
Cruise power: 75% best economy mixture
Cruise speed: 112.5 KTS TAS

BEST ECONOMY CRUISE PERFORMANCE (SERIAL NOS. 28-7716001 THROUGH 7716323)

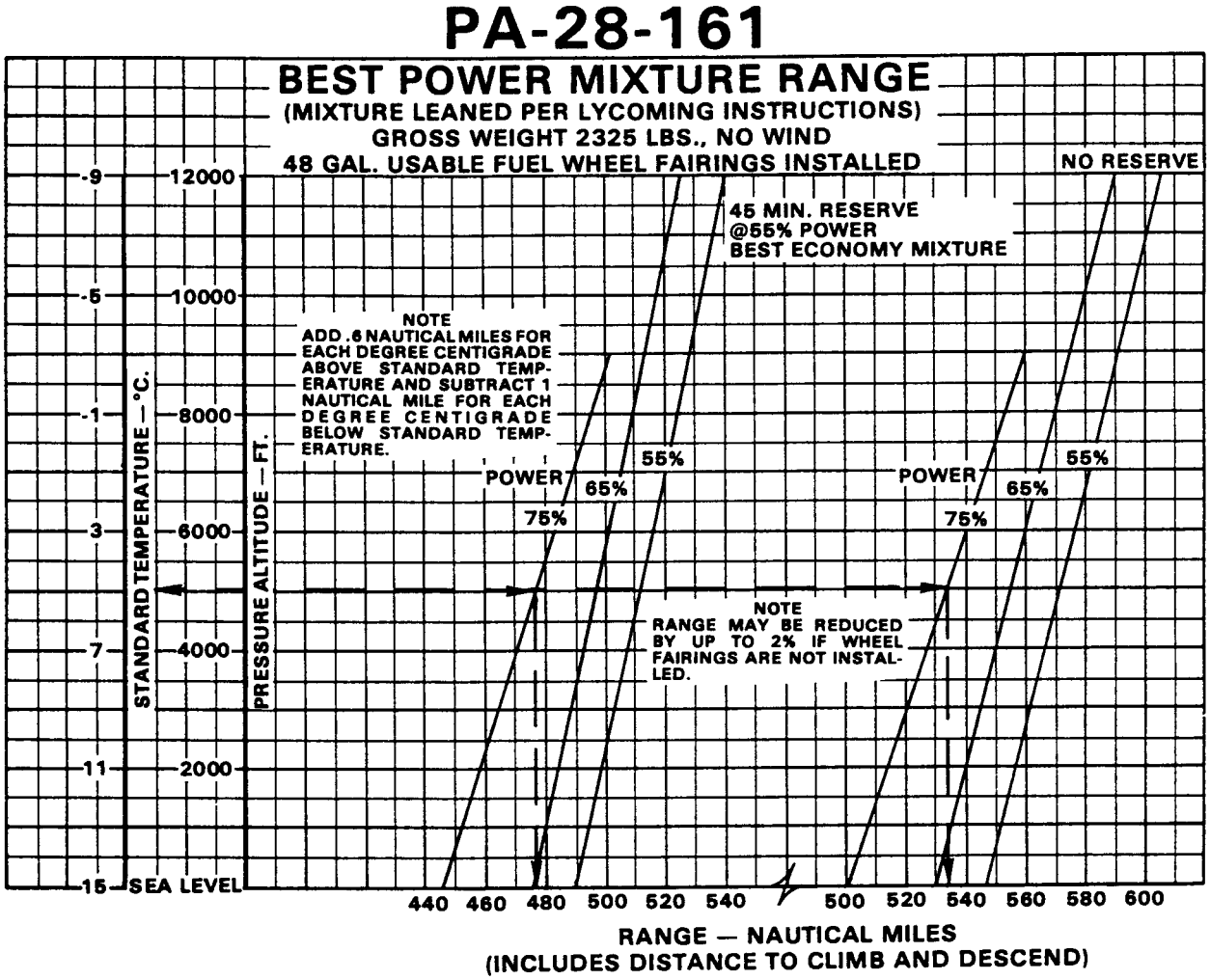
Figure 5-17



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F
Cruise power: 75% best power mixture
Cruise speed: 118 KTS TAS

BEST ECONOMY CRUISE PERFORMANCE (SERIAL NOS. 28-7816001 AND UP)

Figure 5-18

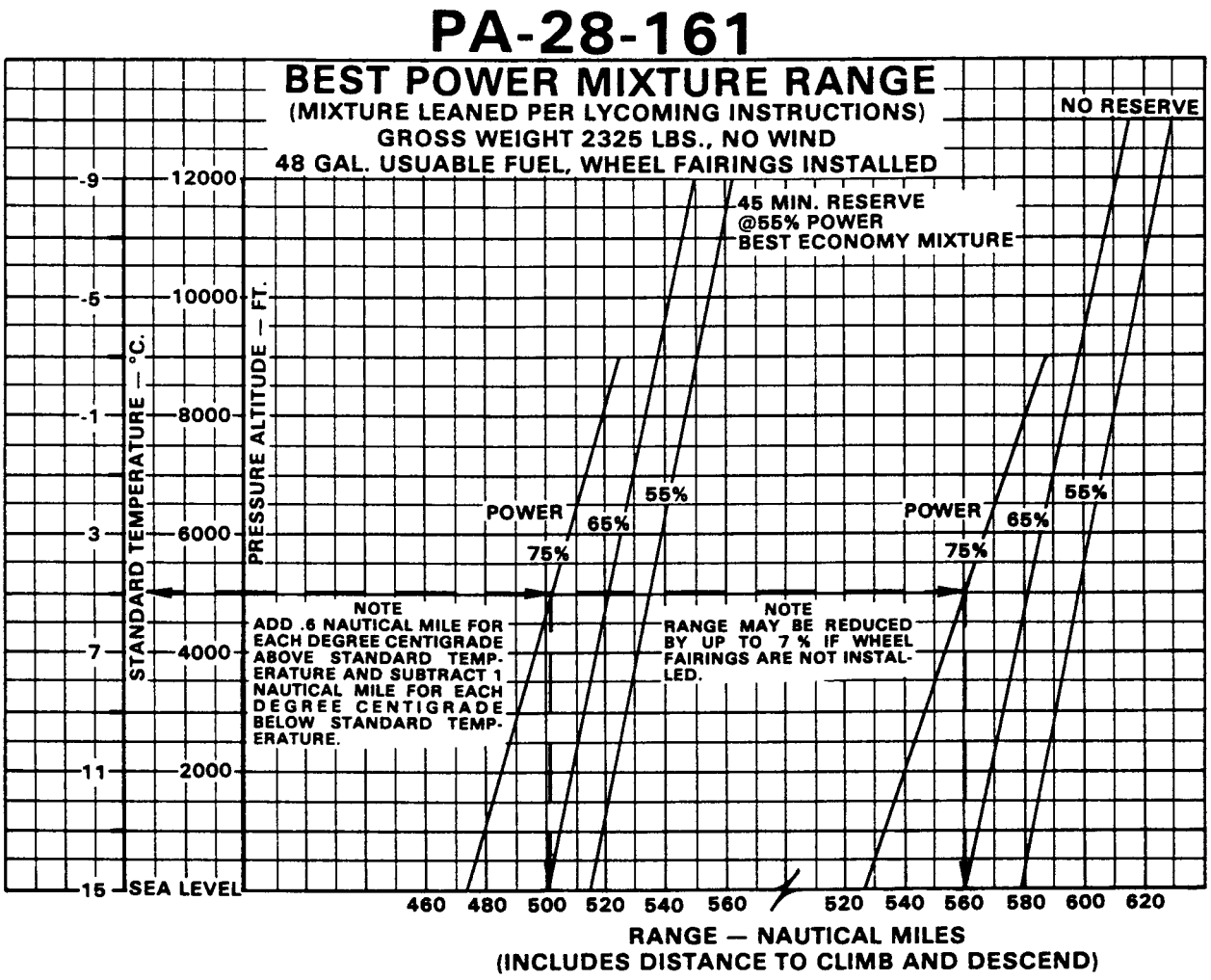


Example:

Cruise pressure altitude: 5000 ft.
Cruise OAT: 16°C (11°C above standard)
Cruise power: 75% best power mixture
Range w/45 min. reserve @ 55% power: 476 + (.6 x 11) = 482.6 nautical miles
Range w/no reserve: 533 + (.6 x 11) = 539.6 nautical miles

BEST POWER MIXTURE RANGE (SERIAL NOS. 28-7716001 THROUGH 7716323)

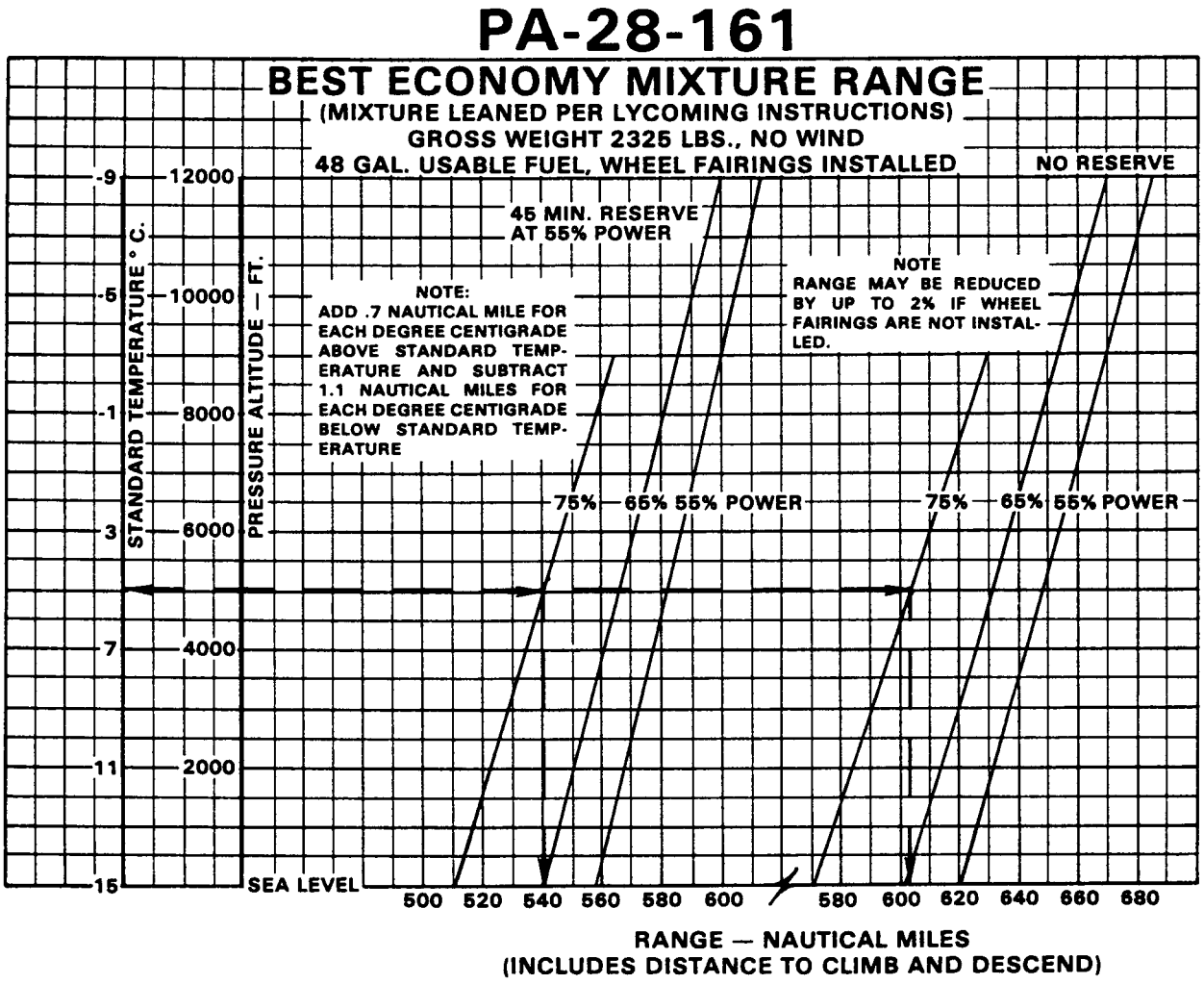
Figure 5-19



Example:
Cruise pressure altitude: 5000 ft.
Cruise OAT: 16°C (11°C above standard)
Cruise power: 75% best power mixture
Range w/45 min. reserve @ 55% power: 501 + (.6 x 11) = 507.6 nautical miles
Range w/no reserve: 561 + (.6 x 11) = 567.6 nautical miles.

BEST POWER MIXTURE RANGE (SERIAL NOS. 28-7816001 AND UP)

Figure 5-20

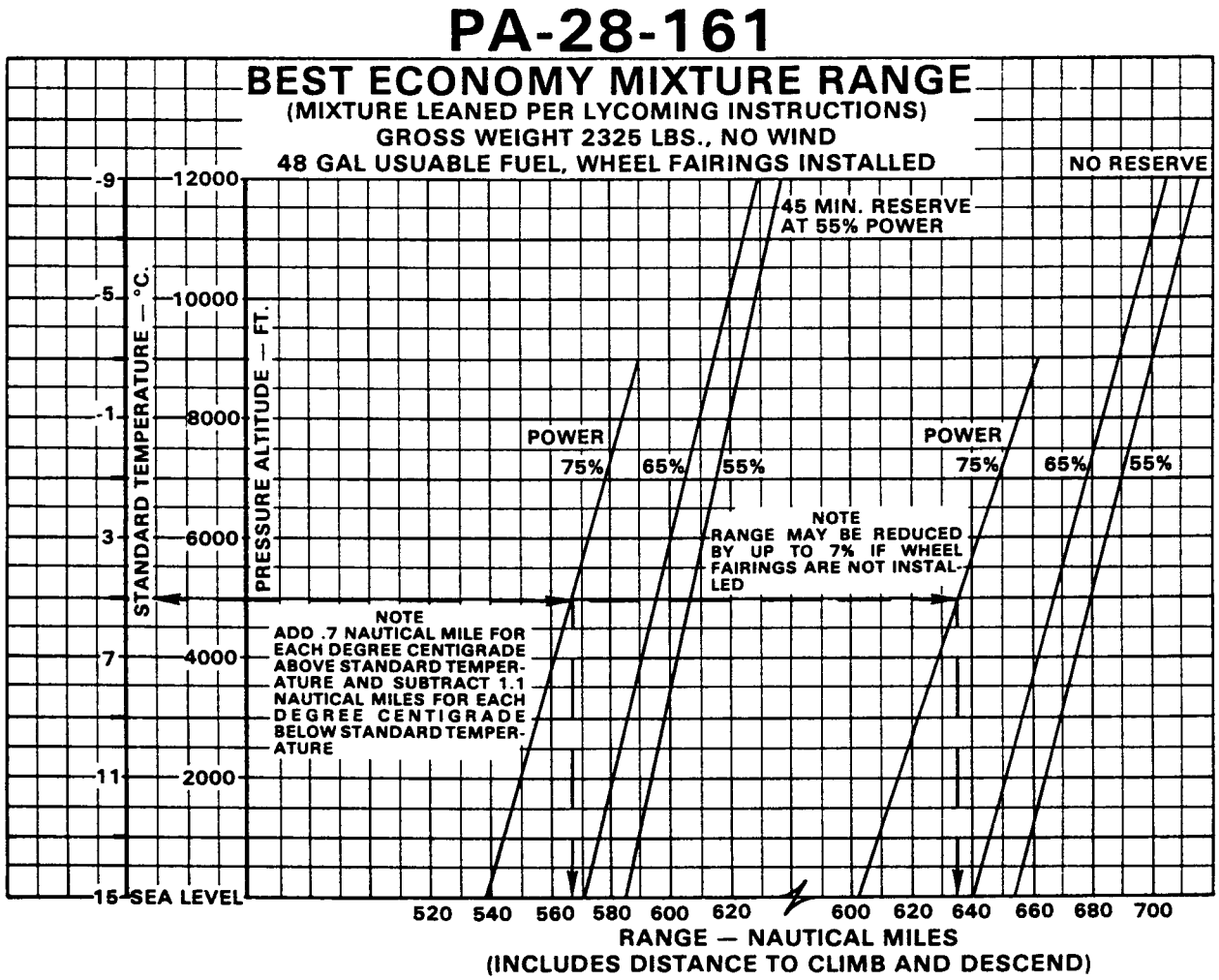


Example:

Cruise pressure altitude: 5000 ft.
Cruise OAT: 16°C (11°C above standard)
Cruise power: 75% best economy mixture
Range w/45 min. reserve @ 55% power: 540 + (.7 x 11) = 547.7 nautical miles
Range w/no reserve: 602 + (.7 x 11) = 609.7 nautical miles

BEST ECONOMY MIXTURE RANGE (SERIAL NOS. 28-7716001 THROUGH 7716323)

Figure 5-21

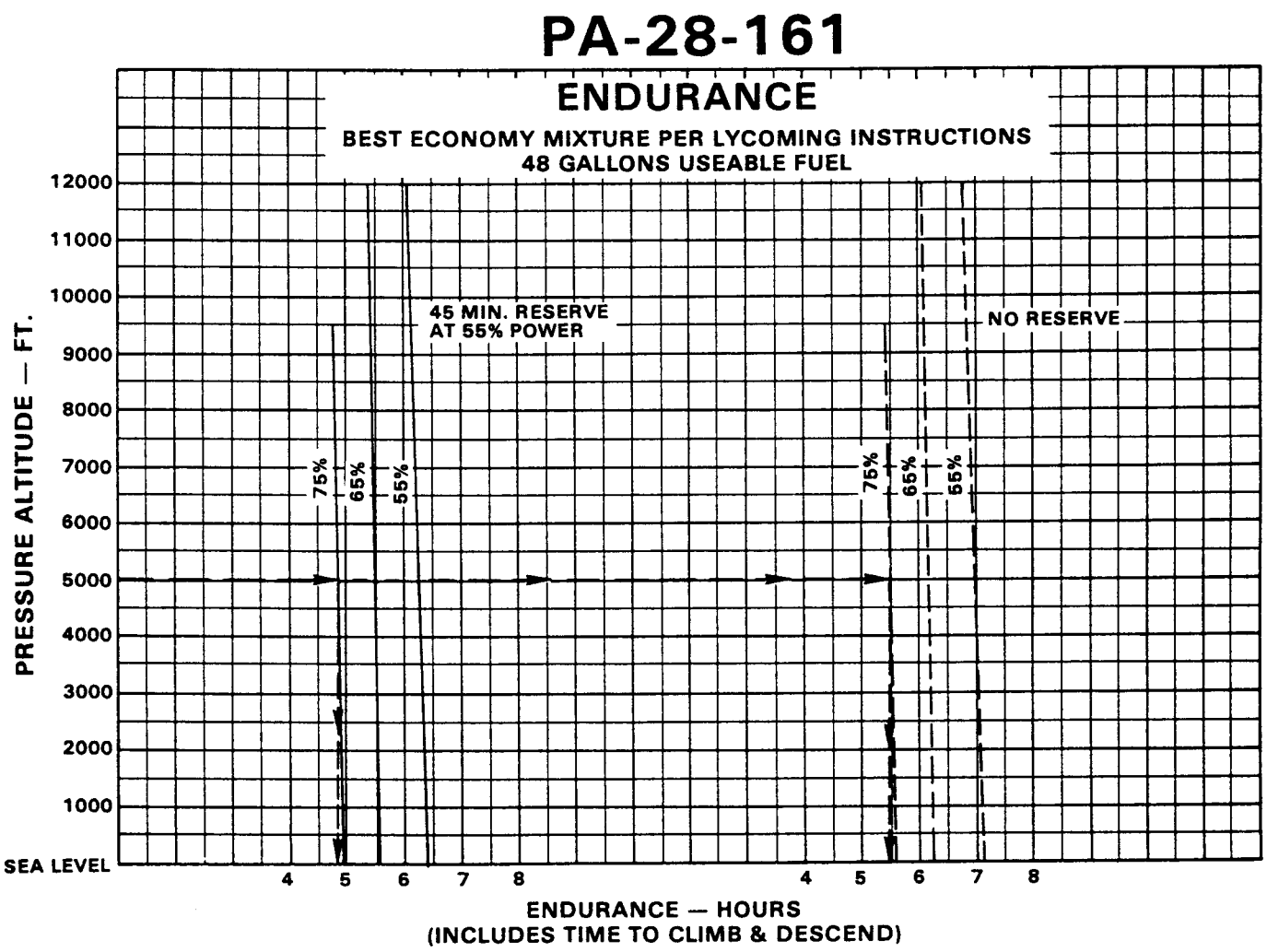


Example:

Cruise pressure altitude: 5000 ft.
Cruise OAT: 16°C (11°C above standard)
Cruise power: 75% best economy mixture
Range w/ 45 min. reserve @ 55% power: 567 + (.7 x 11) = 574.7 nautical miles
Range w/no reserve: 635 + (.7 x 11) = 642.7 nautical miles

BEST ECONOMY MIXTURE RANGE (SERIAL NOS. 28-7816001 AND UP)

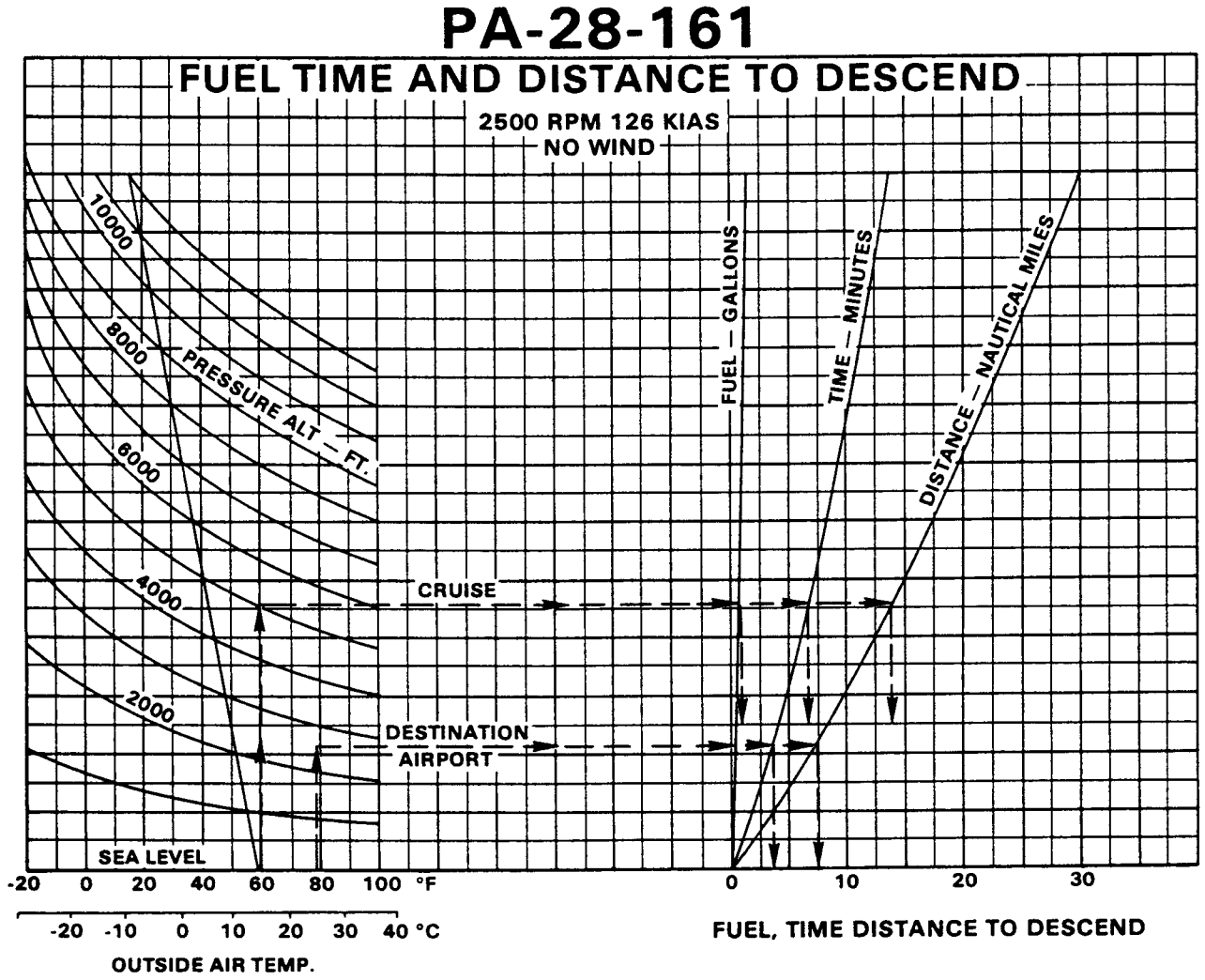
Figure 5-22



Example:
Cruise pressure altitude: 5000 ft.
Cruise power: 75% best economy mixture
Endurance w/45 min. reserve @ 55% power: 4.85 hrs.
Endurance w/no reserve: 5.45 hrs.

ENDURANCE

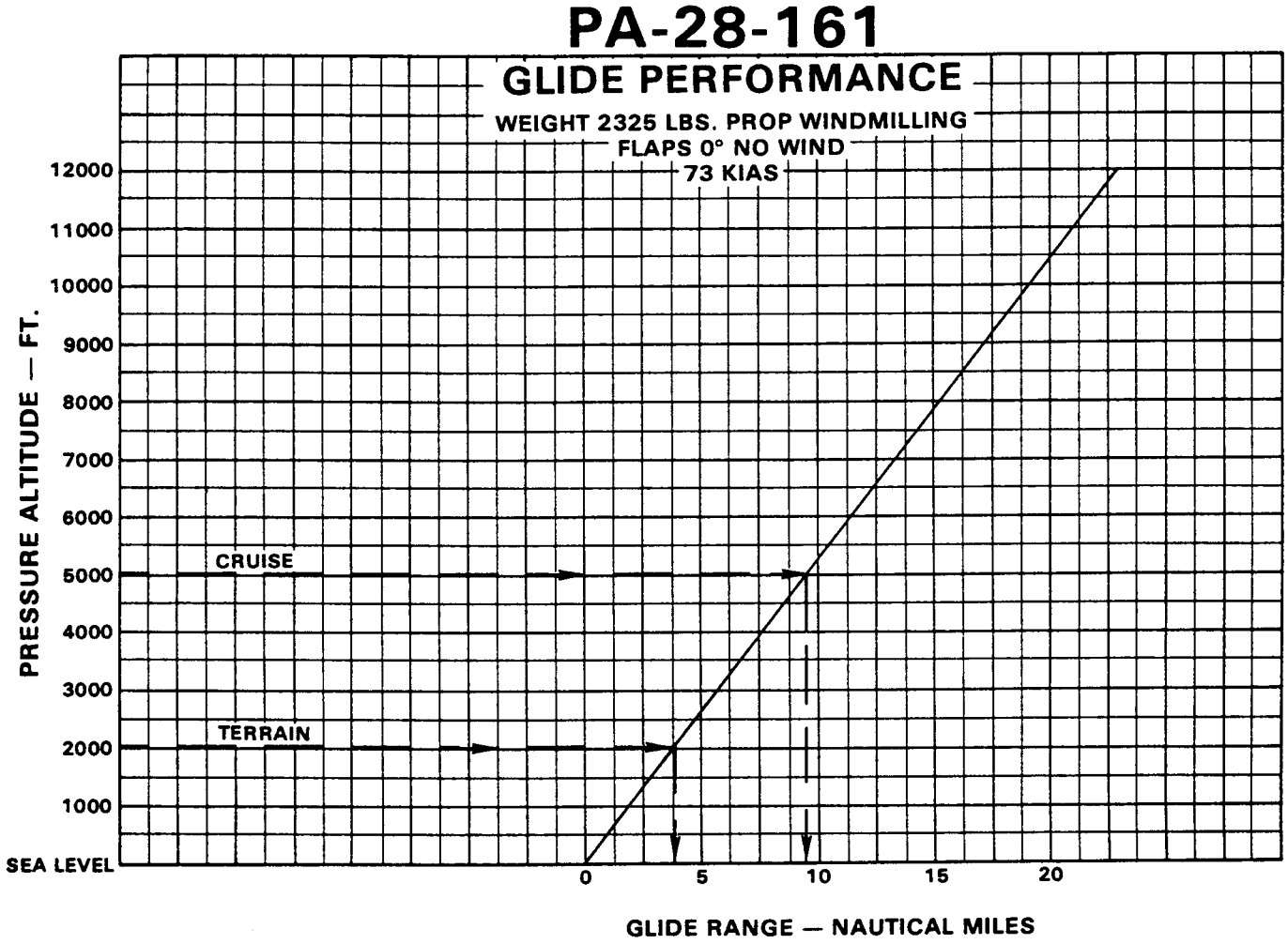
Figure 5-23



Example:
Destination airport pressure altitude: 2500 ft.
Destination airport temperature: 75°F (24°C)
Cruise pressure altitude: 5000 ft.
Cruise OAT: 60°F (16°C)
Time to descend (6.5 min. minus 3.5 min.): 3 min.
Distance to descend (14 miles minus 7.5 miles): 6.5 nautical miles
Fuel to descend: (1 gal. minus .5 gal.): .5 gal.

FUEL, TIME AND DISTANCE TO DESCEND

Figure 5-25

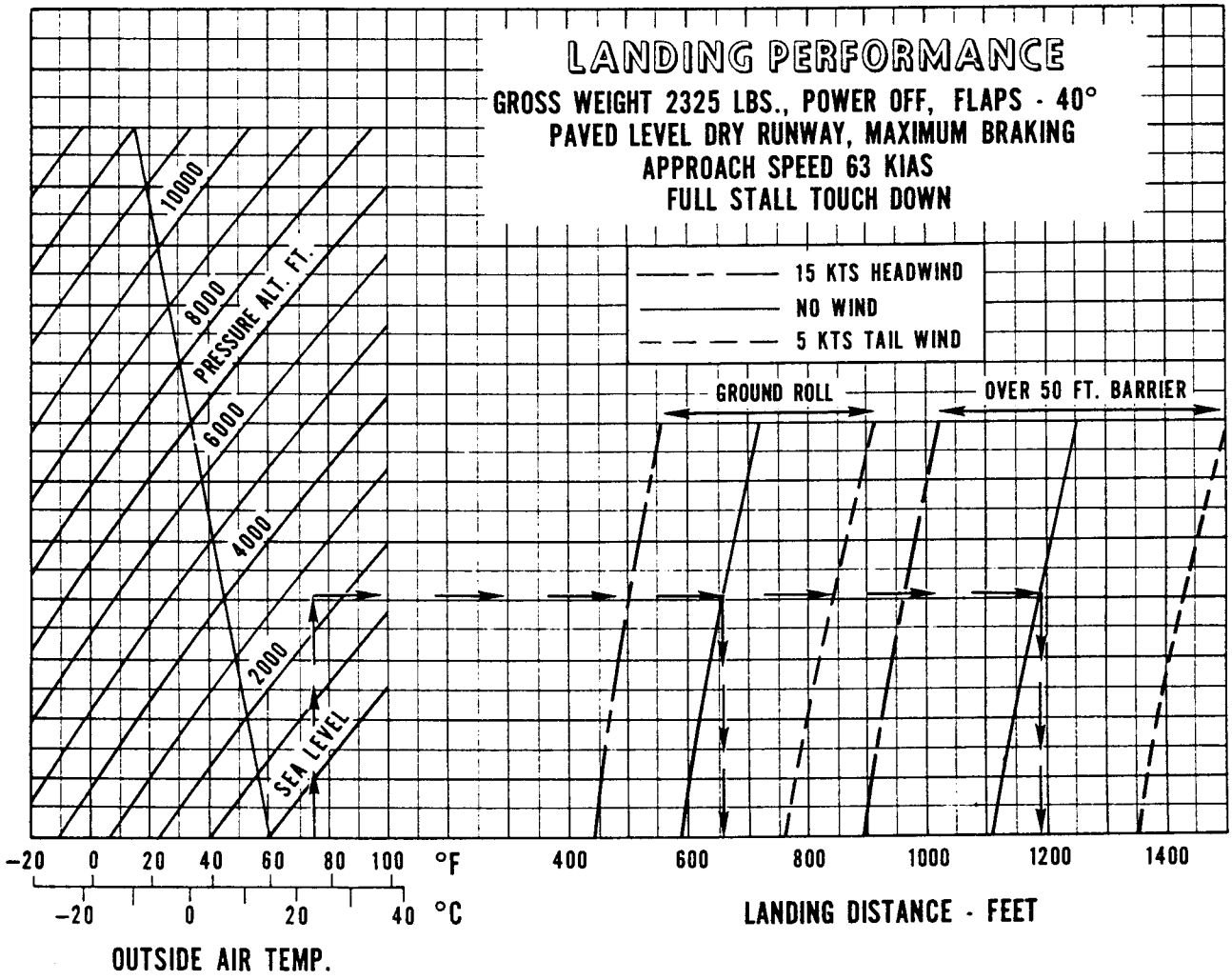


Example:
Cruise pressure altitude: 5000 ft.
Terrain pressure altitude: 2000 ft.
Glide distance (9.5 miles minus 3.8 miles): 5.7 nautical miles

GLIDE PERFORMANCE

Figure 5-27

PA-28-161



Example:

Destination airport pressure altitude: 2500 ft.
Destination airport temperature: 75°F
Destination airport wind: 0 KTS
Ground roll: 660 ft.
Distance over 50 ft. barrier: 1190 ft.

LANDING PERFORMANCE

Figure 5-29

TABLE OF CONTENTS

SECTION 6

WEIGHT AND BALANCE

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WT. & BALANCE and EQUIPMENT LIST REVISION

PIPER	PA-28-161	28-7916139	N405SB
MAKE	MODEL	S/N	REGISTRATION #

IN / OUT	ITEM	WEIGHT	ARM	MOMENT
4/21/2023	*AIRCRAFT OLD AS EQUIPPED*	*1421.19*	*89.04*	*126536.49*
	Remove			
	Vacuum DG	-2.00	68.00	-136.00
	#1 OBS Head	-2.00	68.00	-136.00
	SPS400 Intercom	-0.50	69.00	-34.50
	Vacuum system	-3.00	44.00	-132.00
	KMA24 Audio Panel	-1.70	68.00	-115.60
				0.00
	Installed			0.00
	PS800G Audio Panel	1.50	69.00	103.50
	G5 HIS	2.00	69.00	138.00
	GAD29 Magnetic Indicator	1.00	60.00	60.00
				0.00
				0.00
Ref:	Piper Maintenance Manual and Tech Measurements			0.00
				0.00
AIRCRAFT CURRENTLY AS EQUIPPED		**1416.49**	**89.15**	**126283.89**

GROSS WT.	**2325.00**
EMPTY WT.	**1416.49**
E.W.C.G.	**89.15**
MOMENT	**126283.89**
USEFUL LOAD	**908.51**
DATE	**April 21, 2023**

Neil P. Minschwaner

Name: Neil P. Minschwaner

Certificate #: 2257455

SBI AIRCRAFT SERVICES LACONIA
65 Aviation Drive - Laconia Airport
Gilford, NH 03246
(603) 528-6818 Tel - (603) 528-1814 Fax
www.skybright.com

WT. & BALANCE and EQUIPMENT LIST REVISION

PIPER	PA-28-161	28-7916139	N405SB
MAKE	MODEL	S/N	REGISTRATION #

IN / OUT	ITEM	WEIGHT	ARM	MOMENT
1/27/2021	*AIRCRAFT OLD AS EQUIPPED*	*1422.69*	*88.96*	*126557.49*
	Removed ALY-6521R Alternator	-12.40	14.00	-173.60
	Installed 10-1050-1 Alternator, S/N H-U120058	10.90	14.00	152.60
				0.00
				0.00
				0.00
				0.00
				0.00
				0.00
				0.00
				0.00
Ref:	Piper Maintenance Manual and Tech Measurements			0.00
				0.00
				0.00
				0.00
AIRCRAFT CURRENTLY AS EQUIPPED		**1421.19**	**89.04**	**126536.49**

GROSS WT.	**2325.00**
EMPTY WT.	**1421.19**
E.W.C.G.	**89.04**
MOMENT	**126536.49**
USEFUL LOAD	**903.81**
DATE	**January 27, 2021**

Name: Zachariah A. Rezucha

Certificate #: Pending

SBI AIRCRAFT SERVICES LACONIA
65 Aviation Drive - Laconia Airport
Gilford, NH 03246
(603) 528-6818 Tel - (603) 528-1814 Fax
www.skybright.com

Weight And Balance Data
And Equipment List

104 Grafton Drive Portsmouth New Hampshire 03801
Phone: 603 430 1111 Toll Free: 888 732 7324
fax: Maintenance and Avionics 603 430 1199, fax: Operations 603-776-0266
FAA Certified Repair Station #P49R490N

Aircraft Model : Piper PA-28-161
Registration: N405SB S/N 28-7916139

Revised Weight and Balance

Information taken from previous weight and balance dated		12/13/2016	
	1423.94	88.93	126630.14
	Weight	Arm	Moment
Equipment Removed			
RC Allen Gyro, Horizon P/N 102-0041-04	-2.25	59.40	-133.65
Equipment Installed			
G5 ADI P/N 011-03809-00	1.00	61.00	61.00
NEW Total	**1422.69**	**88.96**	**126557.49**

Superseded

New empty weight: **1422.69 lbs**
New empty weight CG: **88.96 ins. Aft of datum**
Max weight: **2325.00 lbs**
New useful load: **902.31 lbs**

All measurements are in inches and pounds
It is the pilots responsibility to ensure that the aircraft is properly loaded so that the entire flight is conducted within the Gross Weight Center of Gravity Chart in the Limitations section of the Flight Manual.

12/16/2020 Revised By: Alec Jordan

Weight And Balance Data
And Equipment List

104 Grafton Drive Portsmouth New Hampshire 03801
Phone: 603 430 1111 Toll Free: 888 732 7324
fax: Maintenance and Avionics 603 430 1199, fax: Operations 603-776-0266
FAA Certified Repair Station #P49R490N

Aircraft Model Piper PA-28-161 S/N 28-7916139
Registration N405SB

Revised Weight and Balance

Information taken from previous weight and balance Dated 8/15/2005	1427.84	88.83	126835.74
	Weight	Arm	Moment
Equipment Removed			
King KLN-89B GPS P/N 066-01148 w/ rack	-3.4	58.0	-197.20
King KA-92 GPS Antenna P/N 071-01553-0200	-0.3	100.0	-30.00
King KMA-20 Audio Panel P/N 066-1024-03 w/ rack	-2.2	60.0	-132.00
King KT-76A Transponder P/N 066-1062-00 w/ rack	-3.0	58.0	-174.00
King KI-208 VOR/LOC Indicator P/N 066-3036-00	-1.0	59.5	-59.50
Mac 1700 VTX Nav/Com P/N 82-01445-008 w/ rack	-7.6	56.5	-429.40
Equipment Installed			
Garmin GTN 650 GPS/Nav/Com P/N 011-02256-00 w/ rack	7.0	58.0	406.00
Garmin GTX 345 Transponder P/N 011-03302-00 w/ rack	3.1	58.5	181.35
Garmin GI 106B CDI P/N 013-00593-01	1.3	60.0	78.00
Garmin GA 35 GPS Antenna P/N 013-00235-00	0.5	100.0	50.00
BendixKing KMA-24 Audio Panel P/N 066-1055-03	1.7	59.5	101.15
New Total	**1423.94**	**88.93**	**126630.14**

Revised per Mac 12/14/2016 Jordan

New Empty Weight: **1423.94 lbs.**
New Empty Weight CG: **88.93 ins. aft of datum**
Max Weight: **2325 lbs.**
New useful load: **901.06 lbs.**

All measurements are in inches and pounds

It is the pilots responsibility to ensure that the aircraft is properly loaded so that the entire flight is conducted within the Gross Weight Center of Gravity Chart in the Limitations section of the Flight Manual.

12/13/2016 Revised By: Bryan Frobese

WT. & BALANCE and EQUIPMENT LIST REVISION

PIPER	PA-28-161	28-7916139	N405SB
MAKE	MODEL	S/N	REGISTRATION #

IN / OUT	ITEM	WEIGHT	ARM	MOMENT
10/4/1999	*AIRCRAFT OLD AS EQUIPPED*	*1436.90*	*88.81*	*127611.63*
OUT	King KR-86 ADF system	-6.70	91.60	-613.72
OUT	Northstar M1 Loran	-4.20	56.60	-237.72
OUT	Northstar Loran Antenna	-0.40	116.60	-46.64
OUT	Northstar Loran Coupling Unit	-0.28	122.90	-34.41
				0.00
IN	King KLN-89B GPS	2.25	57.60	129.60
IN	King KA-92 Antenna	0.27	100.00	27.00
				0.00
				0.00
				0.00
				0.00
				0.00
Ref:	*POH VB-880 Section 1.11(a) (Normal category)*			0.00
				0.00
	AIRCRAFT CURRENT AS EQUIPPED	**1427.84**	**88.83**	**126835.74**

Revised on 12-13-2016 PJB

GROSS WT.	**2325.00**
EMPTY WT.	**1427.84**
E.W.C.G.	**88.83**
MOMENT	**126835.74**
USEFUL LOAD	**897.16**
DATE	**August 15, 2005**

Name: Philip J. Baker

2613182 A&P

Certificate #:

A.E.R.O. Aircraft Services

Laconia Municipal Airport (KLCI)

60 Airport Road Gilford, NH 03246

(603) 524-8800 Tel - (603) 524-1214 Fax

REVISION

WEIGHT AND BALANCE AND EQUIPMENT LIST

DATE: 10-4-99

AIRCRAFT MAKE AND MODEL	REGISTRATION NUMBER	SERIAL NUMBER
PIPER PA-28-161	N405SB	28-7916139

-OUT +IN	ITEM	WEIGHT	ARM	MOMENT
	AIRCRAFT--OLD AS EQUIPPED	1447.2	88.39	127920.23
-	REMOVED ELECTROSYSTEMS ENGINE STARTER	-20.2	29.00	- 585.80
+	INSTALL SKYTEC ENGINE STARTER	+ 9.9	28.00	+ 277.20
		1436.9	88.81	127611.63

Superseded 8-15-05

GROSS WEIGHT: 2325. NORMAL CATEGORY

EMPTY WEIGHT: 1436.9

E.W.C.G.: 88.81

MOMENT: 127611.63

USEFUL LOAD: 888.1 NORMAL CATEGORY

NAME: [signature]

SIGNATURE: 10-4-99

SCOTT LANIA A&P 20520453

SKY BRIGHT, INC.
20 Airport Road, Laconia Airport
GILFORD, NEW HAMPSHIRE 03246
Airport (603) 528-6818 (24 Hours)

REVISION

WEIGHT AND BALANCE AND EQUIPMENT LIST

DATE: 1-10-98

AIRCRAFT MAKE AND MODEL	REGISTRATION NUMBER	SERIAL NUMBER
~~PIPER PA28-161~~	N405SB	28-7916139

-OUT +IN	ITEM	WEIGHT	ARM	MOMENT
	AIRCRAFT--OLD AS EQUIPPED	1536.1	88,96	136643.03
-	REMOVED AIRCONDITIONER SYSTEM	58.0	103.6	6008.8
-	REMOVED NOSE FAIRING	10.3	36.8	374
-	REMOVED MAIN GEAR FAIRING	20.6	113.6	2340
		1447.2	88.39	127920.23

SUPERSEDED
10-4-99

GROSS WEIGHT: 2325

EMPTY WEIGHT: 1447.2

E.W.C.G.: 88.39

MOMENT: 127920.23

USEFUL LOAD: 877.8

NAME: RICHARD MEYERS A&P 1537483

SIGNATURE: Richard Meyers

SKY BRIGHT, INC.
20 Airport Road, Laconia Airport
GILFORD, NEW HAMPSHIRE 03246
Airport (603) 528-6818 (24 Hours)

CRAIG AVIONICS

Date: March 8, 1996
A/C Model: PA-28-161
A/C Serial#: 28-7916139
A/C Reg.: N 3052C

REVISED WEIGHT AND BALANCE
Sky Bright Inc.

ITEM	WEIGHT	ARM	MOMENT
Old Aircraft Empty Weight Dated: 3/24/96	1539.9	88.75	136675.37
EQUIPMENT REMOVED:			
KX-175	7.5	56.6	42.5
KI-209	1.2	59.9	72.0
KN-75	1.6	184.3	295.0
EQUIPMENT INSTALLED:			
KX-155	5.3	57.6	305.28
KI-209	1.2	59.9	71.88
	1536.1		136643.03

SUPPLEMENT TO SEB W+B DATED 1-10-98

New Aircraft Empty Weight 1536.1 Lbs.
New Aircraft Empty Weight C.G. 88.96 Ins aft of datum
New Aircraft Useful Load 788.9 Lbs.

CRAIG R. DAVENPORT
CHIEF INSPECTOR
FAA REPAIR STATION
#RS5R182N

SECTION 6

WEIGHT AND BALANCE

6.1 GENERAL

In order to achieve the performance and flying characteristics which are designed into the airplane, it must be flown with the weight and center of gravity (C.G.) position within the approved operating range (envelope). Although the airplane offers flexibility of loading, it cannot be flown with the maximum number of adult passengers, full fuel tanks and maximum baggage. With the flexibility comes responsibility. The pilot must insure that the airplane is loaded within the loading envelope before he makes a takeoff.

Misloading carries consequences for any aircraft. An overloaded airplane will not take off, climb or cruise as well as a properly loaded one. The heavier the airplane is loaded, the less climb performance it will have.

Center of gravity is a determining factor in flight characteristics. If the C.G. is too far forward in any airplane, it may be difficult to rotate for takeoff or landing. If the C.G. is too far aft, the airplane may rotate prematurely on takeoff or tend to pitch up during climb. Longitudinal stability will be reduced. This can lead to inadvertent stalls and even spins; and spin recovery becomes more difficult as the center of gravity moves aft of the approved limit.

A properly loaded airplane, however, will perform as intended. Before the airplane is delivered, it is weighed, and a basic empty weight and C.G. location is computed (basic empty weight consists of the standard empty weight of the airplane plus the optional equipment). Using the basic empty weight and C.G. location, the pilot can easily determine the weight and C.G. position for the loaded airplane by computing the total weight and moment and then determining whether they are within the approved envelope.

The basic empty weight and C.G. location are recorded in the Weight and Balance Data Form (Figure 6-5) and the Weight and Balance Record (Figure 6-7). The current values should always be used. Whenever new equipment is added or any modification work is done, the mechanic responsible for the work is required to compute a new basic empty weight and C.G. position and to write these in the Aircraft Log Book and the Weight and Balance Record. The owner should make sure that it is done.

A weight and balance calculation is necessary in determining how much fuel or baggage can be boarded so as to keep within allowable limits. Check calculations prior to adding fuel to insure against improper loading.

The following pages are forms used in weighing an airplane in production and in computing basic empty weight, C.G. position, and useful load. Note that the useful load includes usable fuel, baggage, cargo and passengers. Following this is the method for computing takeoff weight and C.G.

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6.3 AIRPLANE WEIGHING PROCEDURE

At the time of delivery, Piper Aircraft Corporation provides each airplane with the basic empty weight and center of gravity location. This data is supplied by Figure 6-5.

The removal or addition of equipment or airplane modifications can affect the basic empty weight and center of gravity. The following is a weighing procedure to determine this basic empty weight and center of gravity location:

(a) Preparation

(1) Be certain that all items checked in the airplane equipment list are installed in the proper location in the airplane.

(2) Remove excessive dirt, grease, moisture, foreign items such as rags and tools from the airplane before weighing.

(3) Defuel airplane. Then open all fuel drains until all remaining fuel is drained. Operate engine on each tank until all undrainable fuel is used and engine stops. Then add the unusable fuel (2.0 gallons total, 1.0 gallons each wing).

CAUTION

Whenever the fuel system is completely drained and fuel is replenished, it will be necessary to run the engine for a minimum of three minutes at 1000 RPM on each tank to insure no air exists in the fuel supply lines.

(4) Fill with oil to full capacity.

(5) Place pilot and copilot seats in fourth (4th) notch, aft of forward position. Put flaps in the fully retracted position and all control surfaces in the neutral position. Tow bar should be in the proper location and all entrance and baggage doors closed.

(6) Weigh the airplane inside a closed building to prevent errors in scale readings due to wind.

(b) Leveling

(1) With airplane on scales, block main gear oleo pistons in the fully extended position.

(2) Level airplane (refer to Figure 6-3) deflating nose wheel tire, to center bubble on level.

(c) Weighing - Airplane Basic Empty Weight

(1) With the airplane level and brakes released, record the weight shown on each scale. Deduct the tare, if any, from each reading.

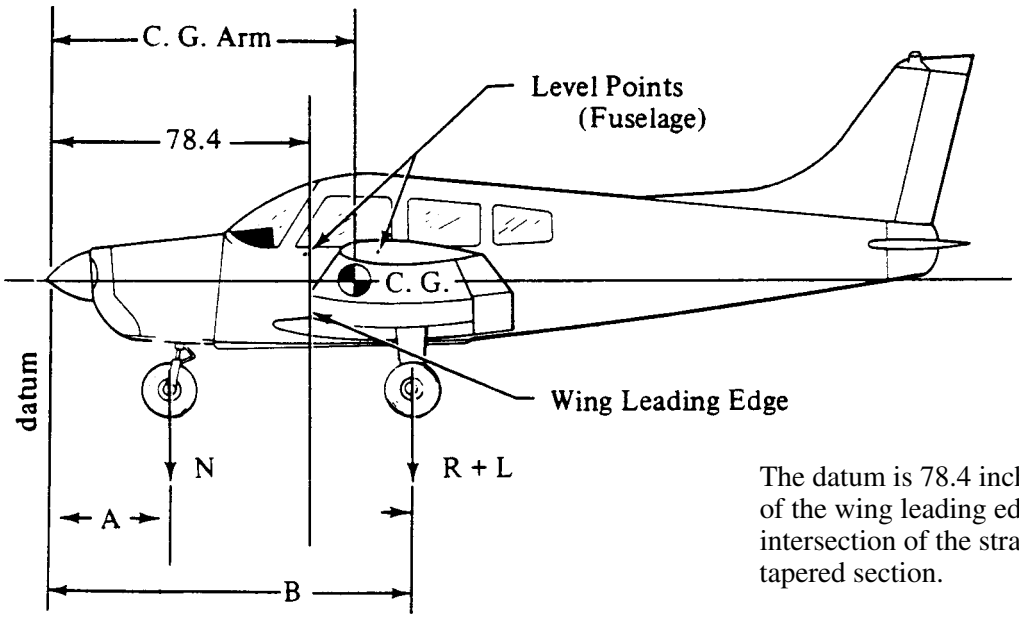
Scale Position and Symbol	Scale Reading	Tare	Net Weight
Nose Wheel (N)			
Right Main Wheel (R)			
Left Main Wheel (L)			
Basic Empty Weight, as Weighed (T)	—	—	

WEIGHING FORM

Figure 6-1

(d) Basic Empty Weight Center of Gravity

(1) The following geometry applies to the PA-28-161 airplane when it is level. Refer to Leveling paragraph 6.3 (b).



The datum is 78.4 inches ahead of the wing leading edge at the intersection of the straight and tapered section.

A = 30.9

B = 109.7

LEVELING DIAGRAM

Figure 6-3

(2) The basic empty weight center of gravity (as weighed including optional equipment, full oil and unusable fuel) can be determined by the following formula:

$$\text{C.G. Arm} = \frac{N\,(A) + (R+L)\,(B)}{T} \quad \text{inches}$$

Where: $T = N + R + L$

6.5 WEIGHT AND BALANCE DATA AND RECORD

The Basic Empty Weight, Center of Gravity Location and Useful Load listed in Figure 6-5 are for the airplane as delivered from the factory. These figures apply only to the specific airplane serial number and registration number shown.

The basic empty weight of the airplane as delivered from the factory has been entered in the Weight and Balance Record (Figure 6-7). This form is provided to present the current status of the airplane basic empty weight and a complete history of previous modifications. Any change to the permanently installed equipment or modification which affects weight or moment must be entered in the Weight and Balance Record.

MODEL PA-28-161 CHEROKEE WARRIOR II

Airplane Serial Number _____

Registration Number _____

Date _____

AIRPLANE BASIC EMPTY WEIGHT

Item		Weight (Lbs)	x	C.G. Arm (Inches Aft of Datum)	=	Moment (In-Lbs)
Standard Empty Weight*	Actual Computed					
Optional Equipment						
Basic Empty Weight						

*The standard empty weight includes full oil capacity and 2.0 gallons of unusable fuel.

AIRPLANE USEFUL LOAD - NORMAL CATEGORY OPERATION

(Gross Weight) - (Basic Empty Weight) = Useful Load

Normal Category: (2325 lbs) - (lbs) = lbs.

Utility Category: (2020 lbs) - (lbs) = lbs.

THIS BASIC EMPTY WEIGHT, C.G. AND USEFUL LOAD ARE FOR THE AIRPLANE AS LICENSED AT THE FACTORY. REFER TO APPROPRIATE AIRCRAFT RECORD WHEN ALTERATIONS HAVE BEEN MADE.

WEIGHT AND BALANCE DATA FORM

Figure 6-5

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PA-28-161			Serial Number	Registration Number					Page Number		
Date	Item No.		Description of Article or Modification	Weight Change						Running Basic Empty Weight	
				Added (+)			Removed (-)				
	In	Out		Wt. (Lb.)	Arm (In.)	Moment /100	Wt. (Lb.)	Arm (In.)	Moment /100	Wt. (Lb.)	Moment /100
			As Delivered								

WEIGHT AND BALANCE RECORD
Figure 6-7

PA-28-161	Serial Number			Registration Number				Page Number			
Date	Item No.		Description of Article or Modification	Weight Change						Running Basic Empty Weight	
				Added (+)			Removed (-)				
	In	Out		Wt. (Lb.)	Arm (In.)	Moment /100	Wt. (Lb.)	Arm (In.)	Moment /100	Wt. (Lb.)	Moment /100
			As Delivered								

WEIGHT AND BALANCE RECORD (cont)

Figure 6-7 (cont)

6.7 WEIGHT AND BALANCE DETERMINATION FOR FLIGHT

(a) Add the weight of all items to be loaded to the basic empty weight.

(b) Use the Loading Graph (Figure 6-13) to determine the moment of all items to be carried in the airplane.

(c) Add the moment of all items to be loaded to the basic empty weight moment.

(d) Divide the total moment by the total weight to determine the C.G. location.

(e) By using the figures of item (a) and item (d) (above), locate a point on the C.G. range and weight graph (Figure 6-15). If the point falls within the C.G. envelope, the loading meets the weight and balance requirements.

	Weight (Lbs)	Arm Aft Datum (Inches)	Moment (In-Lbs)
Basic Empty Weight			
Pilot and Front Passenger	340.0	80.5	27370
Passengers (Rear Seats)*	340.0	118.1	40154
Fuel (48 Gallon Maximum)		95.0	
Baggage* (200 Lbs. Maximum)		142.8	
Total Loaded Airplane			

The center of gravity (C.G.) of this sample loading problem is at inches aft of the datum line. Locate this point ( ) on the C.G. range and weight graph. Since this point falls within the weight - C.G. envelope, this loading meets the weight and balance requirements.

IT IS THE RESPONSIBILITY OF THE PILOT AND AIRCRAFT OWNER TO INSURE THAT THE AIRPLANE IS LOADED PROPERLY.

*Utility Category Operation - No baggage or aft passengers allowed.

SAMPLE LOADING PROBLEM (NORMAL CATEGORY)

Figure 6-9

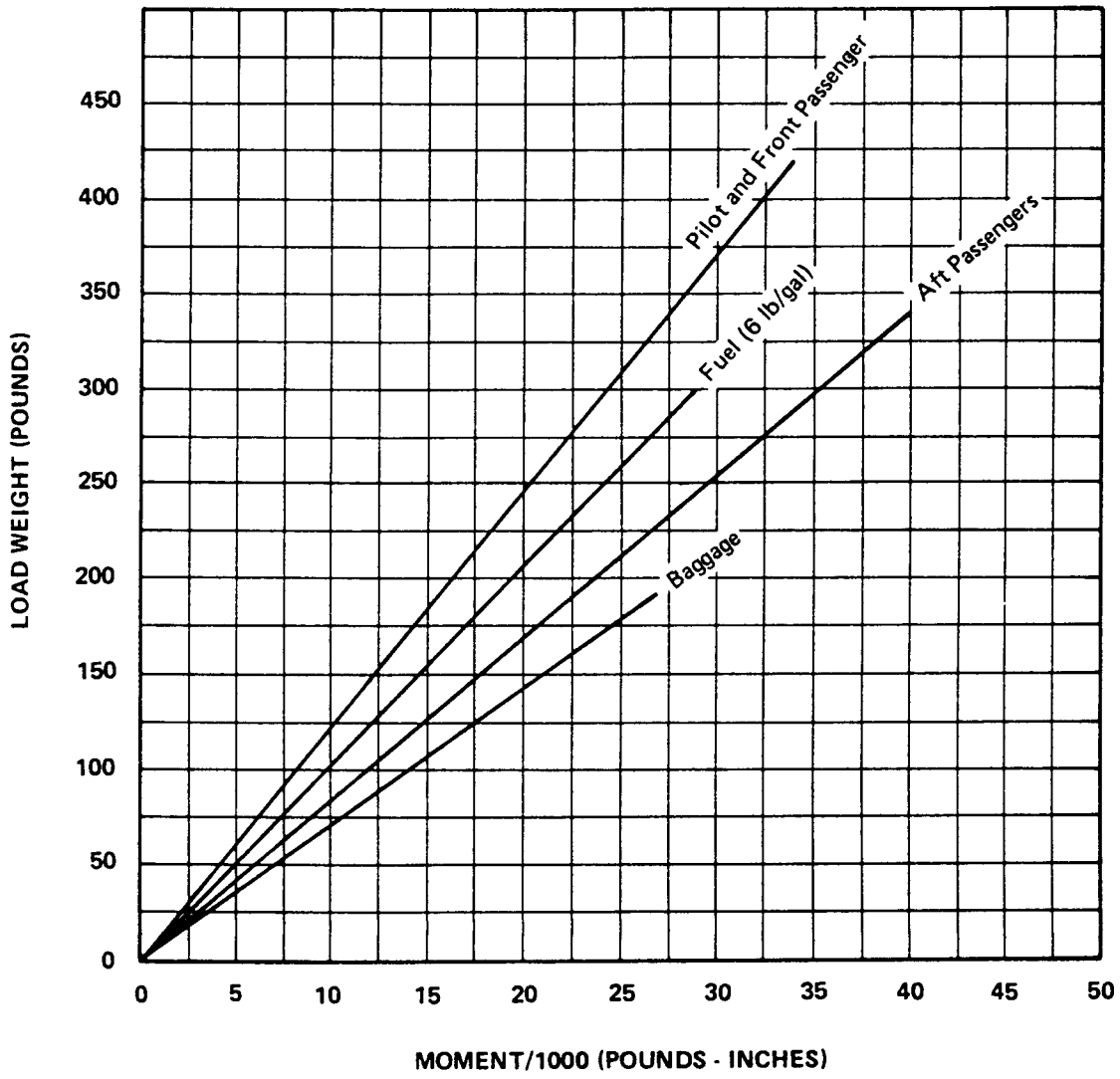
	Weight (Lbs)	Arm Aft Datum (Inches)	Moment (In-Lbs)
Basic Empty Weight			
Pilot and Front Passenger		80.5	
Passenger (Rear Seats)*		118.1	
Fuel (48 Gallon Maximum)		95.0	
Baggage* (200 Lbs. Maximum)		142.8	
Total Loaded Airplane			

Totals must be within approved weight and C.G. limits. It is the responsibility of the airplane owner and the pilot to insure that the airplane is loaded properly. The Basic Empty Weight C.G. is noted on the Weight and Balance Data Form (Figure 6-5). If the airplane has been altered, refer to the Weight and Balance Record for this information.

*Utility Category Operation - No baggage or aft passengers allowed.

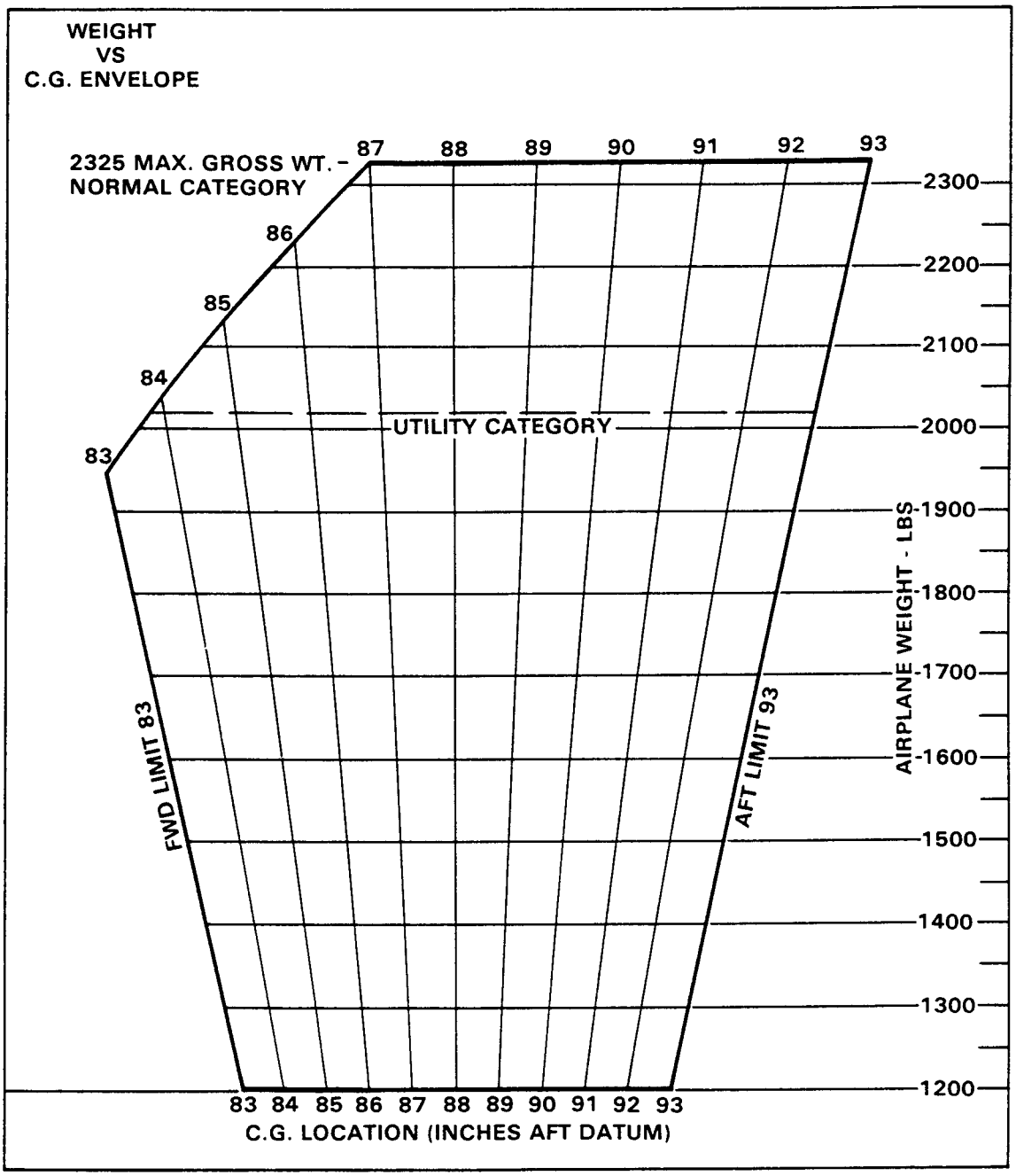
WEIGHT AND BALANCE LOADING FORM

Figure 6-11



LOADING GRAPH

Figure 6-13



C. G. RANGE AND WEIGHT

Figure 6-15

6.8 INSTRUCTIONS FOR USING THE WEIGHT AND BALANCE PLOTTER.

This plotter is provided to enable the pilot quickly and conveniently to:

(a) Determine the total weight and C.G. position.
(b) Decide how to change his load if his first loading is not within the allowable envelope.

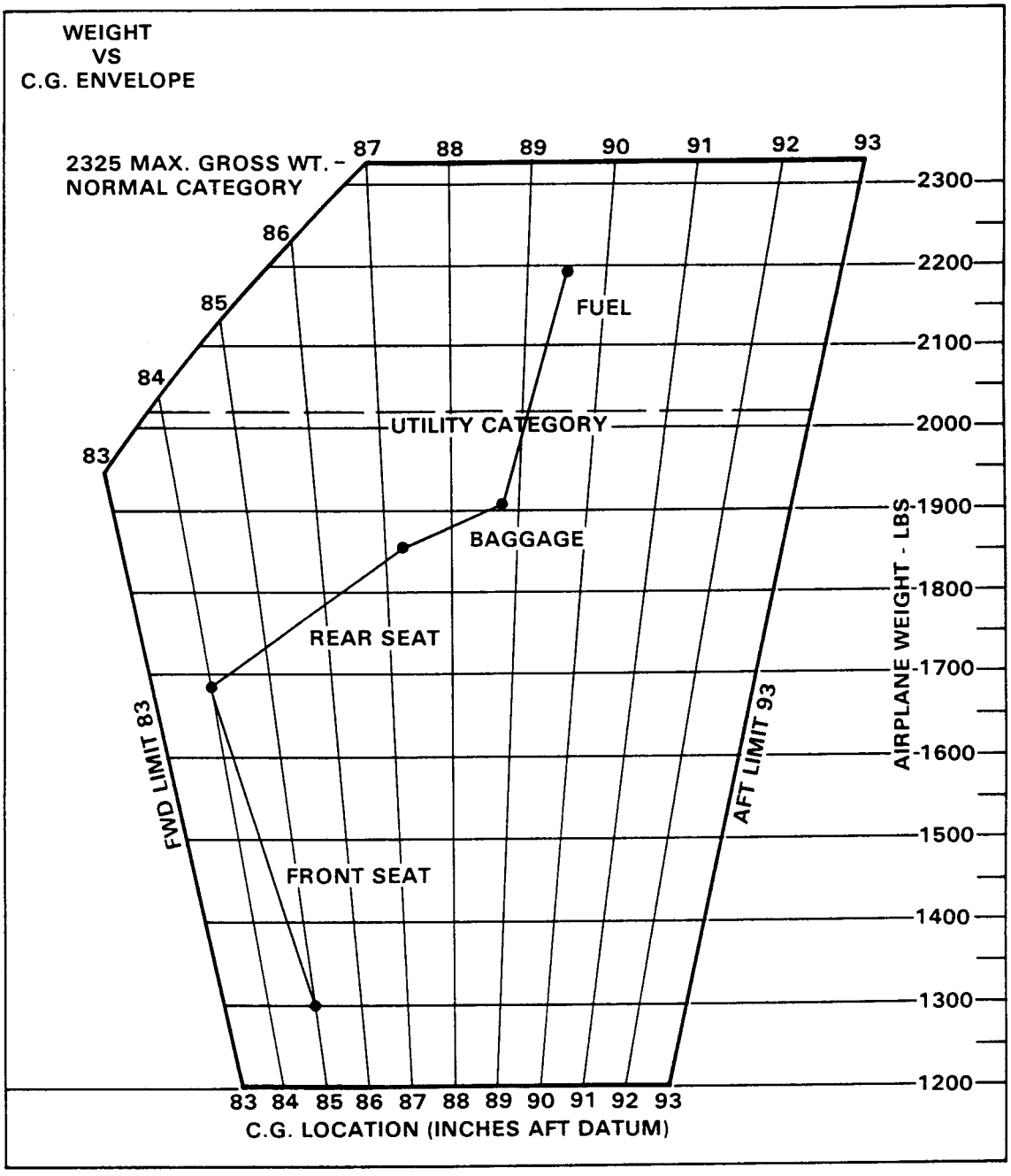
Heat can warp or ruin the plotter if it is left in the sunlight. Replacement plotters may be purchased from Piper dealers and distributors.

When the airplane is delivered, the basic weight and basic C.G. will be recorded on the computer. These should be changed any time the basic weight or C.G. location is changed.

The plotter enables the user to add weights and corresponding moments graphically. The effect of adding or disposing of useful load can easily be seen. The plotter does not cover the situation where cargo is loaded in locations other than on the seats or in the baggage compartments.

Brief instructions are given on the plotter itself. To use it, first plot a point on the grid to locate the basic weight and C.G. location. This can be put on more or less permanently because it will not change until the airplane is modified. Next, position the zero weight end of any one of the loading slots over this point. Using a pencil, draw a line along the slot to the weight which will be carried in that location. Then position the zero weight end of the next slot over the end of this line and draw another line representing the weight which will be located in this second position. When all the loads have been drawn in this manner, the final end of the segmented line locates the total load and the C.G. position of the airplane for takeoff. If this point is not within the allowable envelope it will be necessary to remove fuel, baggage or passengers and/or to rearrange baggage and passengers to get the final point to fall within the envelope.

Fuel burn-off does not significantly affect the center of gravity.



SAMPLE PROBLEM

SAMPLE PROBLEM

A sample problem will demonstrate the use of the weight and balance plotter.

Assume a basic weight and C.G. location of 1300 pounds at 85.00 inches respectively. We wish to carry a pilot and 3 passengers. Two men weighing 180 and 200 pounds will occupy the front seats, and two children weighting 80 and 100 pounds will ride in the rear. Two suitcases weighing 25 pounds and 20 pounds respectively, will be carried in the rear compartment. We wish to carry 48 gallons of fuel. Will we be within the safe envelope?

(a) Place a dot on the plotter grid at 1300 pounds and 85.00 inches to represent the basic airplane. (See illustration.)
(b) Slide the slotted plastic into position so that the dot is under the slot for the forward seats, at zero weight.
(c) Draw a line up the slot to the 380 pound position (180 + 200) and put a dot.
(d) Continue moving the plastic and plotting points to account for weight in the rear seats (80 + 100), baggage compartment (45), and fuel tanks (288).
(e) As can be seen from the illustration, the final dot shows the total weight to be 2193 pounds with the C.G. at 89.44. This is well within the envelope.

As fuel is burned off, the weight and C.G. will follow down the fuel line and stay within the envelope for landing.

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6.9 EQUIPMENT LIST

The following is a list of equipment which may be installed in the PA-28-161. It consists of those items used for defining the configuration of an airplane when the basic empty weight is established at the time of licensing. Only those standard items which are alternate standard items and those required to be listed by the certificating authority (FAA) are presented. Items marked with an "X" are those items which were installed on the airplane described below when licensed by the manufacturer.

Where the letter "A," "B," or "C" precedes an item, "A" denotes an item which is required equipment that must be installed in the aircraft; "B" denotes an item which is required equipment that must be installed in the aircraft unless replaced by an optional equivalent item; "C" denotes an optional item which replaces a required item of standard equipment. Where no letter precedes an item, that item is not required equipment.

Unless otherwise indicated, the installation certification basis for the equipment included in this list is the aircraft's approved type design.

PIPER AIRCRAFT CORPORATION PA-28-161 WARRIOR II

SERIAL NO. _____ REGISTRATION NO._____ DATE:_____

(a) Propeller and Propeller Accessories

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
1	A	Propeller, Sensenich 74DM6-0-60 Cert. Basis - TC P886		32.4	3.8	123
3		Spinner Dome and Bulkhead Piper Dwg. 35323 or 36850		2.9	3.8	11
4		Spinner Dome and Bulkhead Piper Dwg. 87325		3.3	3.8	13

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(b) Engine and Engine Accessories

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
5	A	Engine				
		a. Lycoming Model 0-320-D2A	_____	272.0	21.3	5794
		b. Lycoming Model 0-320-D3G	_____	268.0	21.3	5708
		Cert. Basis - TC 274				
7	B	Alternator 60 Amp				
		a. Prestolite No. ALY6422				
		Piper Dwg. 99981-0	_____	10.5	14.0	147
		b. Chrysler 3656624				
		Piper Dwg. 99945-0	_____	12.4	14.0	174
		c. Chrysler 4111810				
		Piper Dwg. 99945-3	_____	13.5	14.0	189
9	A	Engine Driven Fuel Pump				
		Lycoming Dwg. 75246				
		Cert. Basis - TC E274		1.7	36.3	62
10	A	Electric Fuel Pump				
		Bendix P/N 478360		1.8	36.8	66
11	A	Fuel Valve				
		Piper Dwg. 66945 or Allen				
		Aircraft Prod. Inc. No. 6S122	_____	0.4	61.9	25
12	A	Oil Coolers				
		Piper Dwg. 18622				
		Harrison No. C8526250		1.9	41.3	78
13	A	Air Filter				
		Piper Dwg. 35477		0.9	29.5	27
14	A	Starter				
		Prestolite MZ4218				
		Cert. Basis - TC E274		*17.0	14.5	247
15	A	Oil Filter				
		LW-13743 (Champion No. CH48110) or Lyc. No. 75528 (AC No. OF5578770)				
		Cert. Basis - TC E274		**2.5	35.3	89

*Included in engine weight.
**Includes adapter.

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(c) Landing Gear and Brakes

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
16	A	Two Main Wheel Assemblies a. Cleveland Aircraft Products Wheel Assy. No. 40-86 Brake Assy. No. 30-55 Cert. Basis - TSO C26a b. 6.00-6 Type III 4 Ply Rating Tires with Regular Tubes Cert. Basis - TSO C62	_____	32.3	109.6	3540
17	A	Nose Wheel Assembly a. Cleveland Aircraft Products Wheel Assy. No. 40-77A Cert. Basis - TSO C26a	_____	2.6	30.8	80
		b. McCauley Industrial Corp. Wheel Assy. No. D-30500 Cert. Basis - TSO C26b	_____	3.6	30.8	111
		c. 5.00-5 Type III 4 Ply Rating Tire with Regular Tube Cert. Basis - TSO C62	_____	5.8	30.8	179
18	A	Hand Brake Master Cylinder Piper Dwg. 65842 (Cleveland Aircraft Products P/N 10-22)	_____	0.6	60.9	37
19	A	Toe Brake Cylinders a. Cleveland Aircraft Products No 10-27	_____	0.7	53.0	37
		b. Gar-Kenyon Instrument No 17000	_____	0.4	53.0	21

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(d) Electrical Equipment

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
20	A	Voltage Regulator Piper Dwg. 68804-3		0.9	51.9	47
21	B	Battery Piper Dwg. 35544 (Rebat S-25)	_____	21.9	114.9	2516
22	A	Starter Relay Piper Dwg. 99130-2 RBM Controls P/N 111-111		1.0	45.8	46
23	A	Overvoltage Relay Piper Dwg. 35544 (Wico X16799)		0.5	55.4	28
24	A	Stall Warning Device Piper Dwg. 35544 (Safe Flight P/N C52207-4)		0.2	80.2	16
25	A	Stall Warning Horn Piper Dwg. 35544 (Safe Flight P/N 35214)	_____	0.2	58.8	12

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(e) Instruments

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
26	B	Altimeter Piper PS50008-2 or -3 Cert. Basis - TSO C10b	_____	1.1	60.9	67
27	B	Airspeed Indicator Piper PS50049-41S Cert. Basis - TSO C2b	_____	0.6	61.8	37
28	A	Compass Piper Dwg. 67462 Cert. Basis - TSO C7c		0.9	59.9	54
29	A	Tachometer Piper Dwg. 62177-3		0.7	61.2	43
30	A	Engine Cluster Piper Dwg. 95241-17		0.8	62.4	50

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(f) Miscellaneous

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
33	A	Front Seat Belts (2) Piper PS50039-4-2A American Safety Eqpt. Corp. 500576 Davis Acft. Prod. Inc. FDC-5900-120-5 (Black) Cert. Basis - TSO C22f		1.8	84.0	151
35	A	Aft Seat Belts (2) Piper PS50039-4-3 American Safety Eqpt. Corp. 449968 Davis Acft. Prod. Inc. FDC-5900-120-2 (Black) Cert. Basis - TSO C22f		1.6	123.0	197
36	B	Left Front Seat Piper Dwg. 79337-21	_____	15.5	84.0	1302
37	B	Right Front Seat Piper Dwg. 79337-2	_____	15.5	84.0	1302
38		Rear Seat Piper Dwg. 35131		27.0	124.1	3351
39	A	a. Shoulder Harness (2) Front Seats Only) Piper PS50039-4-20 Pacific Scientific P/N 110747-13	_____	1.4	119.5	167
	B	b. Shoulder Harness-Fixed (Front) (2) Piper Dwg. PS50039-4-23 American Safety Eqpt. Corp. 501385-407 Davis Acft. Prod. Inc. FDC-7275-16-4 (Black)	_____	1.1	119.5	131
40	A	Baggage Straps Piper Dwg. 66804 and 66805		1.3	142.8	186

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(g) Engine and Engine Accessories (Optional Equipment)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
45	Primer System Piper Dwg. 35327-0	_____	1.2	50.0	60
47	Carburetor Ice Detector Piper Dwg. 39684-2	_____	0.5	59.7	30

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(h) Propeller and Propeller Accessories (Optional Equipment)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)

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(i) Landing Gear and Brakes
(Optional Equipment)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
63	Nose Wheel Fairing Piper Dwg. 35513	_____	3.8	29.8	113
65	Main Wheel Fairings Piper Dwg. 65237	_____	7.6	113.6	863
67	Nose Wheel Fairing Piper Dwg. 37896-2	_____	10.3	36.3	374
69	Main Wheel Fairings Piper Dwg. 37885-2, -3	_____	20.6	113.6	2340
71	Nose Wheel Fairing Piper Dwg. 37896-2	_____	3.5	36.3	127
73	Main Wheel Fairings Piper Dwg. 79893-2, -3	_____	17.0	113.6	1931

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(j) Electrical Equipment
(Optional Equipment)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
79	Instrument Panel Lights Piper Dwg. 35544	_____	0.3	62.8	19
81	Instrument Light (2), Grimes 15-0083-7 or Whelen A300-W-14	_____	0.1	99.0	10
83	Cabin Light Piper Dwg. 95229	_____	0.3	99.0	30
85	Landing Light G.E. Model 4509	_____	0.5	13.1	7
87	Navigation Lights (2) Grimes Model A1285 (Red and Green)	_____	0.4	106.6	43
88	Navigation Light (Rear) (1) Grimes Model A2064 (White)	_____	0.2	281.0	56
89	Navigation Lights (Wing) (2) Red/White & Green/White Whelen Model A675	_____	0.5	106.6	53
90	Navigation Lights (Wing) (2) Red White & Green White with White Strobe (Wing) Whelen Model A600 Fin Strobe (A-470)	_____ _____	5.8 1.1	157.9 216.0	916 238
91	Navigation Lights (Wing) (2) Red White & Green White with Red Strobe (Wing) Fin Strobe (A-470)	_____ _____	5.8 1.1	157.9 216.0	916 238
92	Rotating Beacon Whelen Eng. Co. WRMI-12 Piper Dwg. 63892 or 63518	_____	1.5	263.4	395
93	Anti-Collision Light (Fin only) Piper Dwg. 99033-2 Includes power supply	_____	3.1	210.3	652

(j) Electrical Equipment
(Optional Equipment) (cont)

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
94		Anti-Collision Lights (Wing tips only) Cert. Basis - STC SA800 EA	_____	5.7	157.9	900
95		Anti-Collision Lights (Fin and Wing Tips) Piper Dwg. 99033-10	_____	6.1	172.8	1054
97		Heated Pitot Head Piper Dwg. 35493-2	_____	0.4	100.0	40
99		Piper Pitch Trim. Piper Dwg. 67496-3	_____	4.3	155.3	668
101	C	Battery 12V 35 A.H. Rebat R35 (Wt. 27.2 lbs.)	_____	*5.3	114.9	609
103		Auxiliary Power Receptacle Piper Dwg. 35298	_____	2.7	178.5	482
105		External Power Cable Piper Dwg. 62355-11	_____	4.6	142.8	657
107		Lighter, #200462, 12 Volt Universal	_____	0.2	62.9	13

*Weight and moment difference between standard and optional equipment.

(k) Instruments
(Optional Equipment)

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
113		Vacuum System Installation				
		a. With Airborne Model 211cc Pump	_____	4.5	39.1	176
		b. With Edo-Aire Model 1U128A Pump	_____	4.9	39.1	192
115		Attitude Gyro, Piper Dwg. 99002-2, -3, 4 or -8 Cert. Basis - TSO C4c	_____	2.2	59.4	131
117		Directional Gyro, Piper Dwg. 99003-2, -3, -4 or -7 Cert. Basis - TSO C5c	_____	2.6	59.7	155
119	C	Tru-Speed Indicator Piper PS50049-41T Cert. Basis - TSO C2b	_____	(same as standard equipment)		
121	C	Encoding Altimeter Piper PS50008-6 or -7 Cert. Basis - TSO C10b, C88	_____	*0.9	60.3	54
122		Altitude Digitizer (United Instruments P/N 5125-P3) Cert. Basis - TSO C88	_____	1.0	51.5	52
123		Vertical Speed Piper Dwg. 99010-2, -4 or -5 Cert. Basis - TSO C8b	_____	1.0	60.9	61
125		Alternate Static Source Piper Dwg. 35493	_____	0.4	61.0	24
127		Turn and Slip Indicator Piper PS50030-2 or -3 Cert. Basis - TSO C3b	_____	2.6	59.7	155

*Weight and moment difference between standard and optional equipment.

(k) Instruments
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
129	Engine Hour Meter Piper Dwg. 69889-0	_____	0.3	61.2	18
131	Clock	_____	0.4	62.4	25
132	Control Wheel Digital Clock Piper Dwg. 87347-3	_____	0.3	71.9	22
133	Air Temperature Gauge Piper Dwg. 99479-0 or -2	_____	0.2	72.6	15

(l) Autopilots
(Optional Equipment)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
141	AutoFlite II Piper Dwg. 99447 Cert. Basis - STC SA3066SW-D	_____	5.6	91.8	514
143	AutoControl IIIB	_____	9.6	77.6	745
	a. Omni Coupler 1C-388 Piper Dwg. 79221 Cert. Basis - STC SA3065SW-D	_____	1.0	59.3	59
145	AutoPilot - Century 21 Piper Dwg. 39726 Cert. Basis - STC SA3352SW	_____	12.0	69.0	828

(m) Radio Equipment
(Optional Equipment)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
151	Bendix AS-2015A-7 or -9 Audio Panel	_____	1.0	66.4	66
152	Bendix CN2013-1 Com/Nav Cert. Basis - TSO C34c, C35d, C36c, C37b, C38b, C40a	_____	7.5	61.4	461
153	Bendix CN2013-2 Com/Nav w/G.S. Receiver Cert. Basis - TSO C34c, C35d, C36c, C37b, C38b, C40a	_____	8.2	61.4	504
154	Bendix CN 2013-4 Com/Nav w/G.S. Receiver and M.B. Receiver	_____	8.5	61.4	522
155	Bendix ADF 2070 Cert. Basis - TSO C41c, C2a	_____	6.0*	105.0	630
156	Bendix TR2060 Transponder Cert. Basis - TSO C74c	_____	2.8*	63.6	178
157	Bendix CN2011 Dual Com/Nav Cert. Basis - TSO C34c, C35d C36c, C37b, C40a	_____	16.8	66.8	1122
158	Bendix IN2014B Indicator a. Single b Dual Cert. Basis - TSO C34c, C 36c, C40a, C66c	_____ _____	1.9 3.8	63.4 63.4	121 241
159	Bendix DME 2030 Cert. Basis - TSO C66a	_____	10.3*	185.0	1906

*Weight includes antenna and cable

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
161	Collins VHF-250 or VHF-251 Comm Transceiver				
	a. Single	_____	4.0	56.9	228
	b. Dual	_____	8.1	56.9	461
	Cert. Basis - TSO C37b, C38b				
162	Collins VIR-350 or VIR-351 Nav Receiver				
	a. Single	_____	3.9	57.4	224
	b. Dual	_____	7.9	57.4	453
	Cert. Basis - TSO C40a, C36c				
163	Collins IND-350 ( ) VOR/LOC Indicator				
	a. Single	_____	1.0	60.2	60
	b. Dual	_____	2.0	60.2	120
	Cert. Basis - TSO C40a, C36c				
164	Collins IND-351 ( ) VOR/LOC/GS Indicator				
	Cert. Basis - TSO C40a, C36c	_____	1.3	60.2	78
165	Collins GLS-350 Glide Slope Receiver				
	Cert. Basis - TSO C34c	_____	2.0	183.4	367
167	Collins DCE 400 Distance Computing Equipment				
	Cert. Basis - TSO C40a	_____	2.1	58.9	124
168	Collins RCR-650 ADF Receiver and Antenna and IND-650 Indicator				
	Cert. Basis - TSO C41c	_____	6.6	104.8	692
169	Collins RCR-650A ADF Receiver and Antenna and IND-650A Indicator				
	Cert. Basis - TSO C41c	_____	7.3	100.3	733

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
170	Collins AMR-350 Audio/Marker Panel Cert. Basis - TSO C35d, C50b	_____	*3.3	110.0	363
171	Collins DME-451 W/Ind. 451 450 Cert. Basis - TSO C66a	_____	8.0	174.9	1399
172	Collins TDR-950 Transponder Cert. Basis - TSO C74c	_____	**2.8	62.9	176
173	King KN 53 Nav/Receiver	_____	2.8	63.8	179
177	King KX 170 ( ) VHF Comm/Nav				
	a. Transceiver, Single	_____	7.5	56.6	425
	b. Transceiver, Dual	_____	15.0	56.6	849
174	King KN 53 Nav Receiver W GS Receiver				
	a. Single	_____	3.1	63.8	198
	b. Dual	_____	6.2	63.8	396
175	King KX 155 VHF Nav/Comm Transceiver				
	a. With Audio Amplifier	_____	5.0	58.1	291
	b. With Glide Slope Receiver	_____	5.3	58.1	308
	c. Without Glide Slope Receiver	_____	4.8	58.1	279
	Cert. Basis - TSO C37b, C38b, C40a, C36a				
176	King KX 165 VHF Nav/ Comm Transceiver				
	a. With Glide Slope Receiver	_____	5.7	58.0	331
	b. Without Glide Slope Receiver	_____	5.1	58.1	296
	Cert. Basis - TSO C37b, C38b C40a, C36a				

*Weight includes antenna and cable.
**Weight includes antenna.

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
178	King KX 175 ( ) VHF				
	a. Transceiver	_____	9.4	56.6	532
	b. King KN 72 VOR/LOC Converter	_____	1.3	183.6	239
	c. King KN 73 Glide Slope Receiver	_____	3.2	184.3	590
	d. KN 75 Glide Slope Receiver	_____	1.6	184.3	295
	e. King KN 77 VOR/LOC Converter	_____	3.6	183.6	661
	f. King KI-204 VOR/ILS Indicator	_____	1.7	60.5	103
	g. King KNI-520 VOR/ILS Indicator	_____	1.7	60.5	103
	Cert. Basis - TSO C36c, C37b, C38b, C40a				
179	King KX 175 ( ) VHF				
	a. Transceiver (2nd)	_____	8.6	56.6	487
	b. King KN 72 VOR/LOC Converter	_____	1.3	183.6	239
	c. King KN 77 VOR/LOC Converter	_____	4.2	183.6	771
	d. King KI-203 VOR/ILS Indicator	_____	1.6	60.5	97
	e. King KNI 520 VOR/ILS Indicator	_____	1.7	60.5	103
	Cert. Basis - TSO C36c, C37b, C38b, C40a				
180	King KY 196E Transceiver with RB 125 Power Booster				
	a. Single	_____	5.7	77.0	439
	b. Dual	_____	11.4	77.0	878
	Cert. Basis - TSO C37b, C38b				
181	King KY 197 Transceiver				
	a. Single	_____	4.2	58.7	246
	b. Dual	_____	8.4	58.7	492
	Cert. Basis - TSO C37B, C38B				
182	King KI 201 ( ) VOR/LOC Ind.				
	a. Single	_____	2.5	59.6	149
	b. Dual	_____	5.0	59.9	300
183	King KI 202 VOR/LOC Indicator				
	Cert. Basis - TSO C40a, C36c	_____	1.3	60.9	79

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
184	King KI 206 VOR/LOC Indicator Cert. Basis - TSO C40a, C36c	_____	1.3	60.9	79
185	King KI 208 VOR LOC Indicator				
	a. Single	_____	1.0	59.6	60
	b. Dual	_____	2.0	59.9	120
	Cert. Basis - TSO C34c, C36c, C40a				
186	King KI 209 VOR LOC GS Ind. Cert. Basis - TSO C34c C36c, c40a	_____	1.2	59.9	72
187	King KI 213 VOR LOC GS Ind.	_____	2.5	60.4	151
188	King KI 214 ( ) VOR LOC GS Ind.	_____	3.3	59.9	198
189	King KN 74 R-Nav	_____	4.7	56.6	266
191	King KN 61 DME	_____	12.5	179.0	2237
192	King KN 62A DME	_____	3.3	58.3	193
193	King KN 65A DME Cert. Basis - TSO C66a	_____	13.0	174.9	2274
194	King KRA-10 Radio Altimeter	_____	4.3	162.6	699
195	King KR 85 Digital ADF	_____	8.6	85.2	733
	a. Audio Amplifier Cert. Basis - TSO C41b	_____	0.8	51.0	41
196	King KR 85 ADF with KA 42B Loop and Sense Antenna	_____	9.5	85.2	809
	a. Audio Amplifier Cert. Basis - TSO C41b	_____	0.8	51.0	41
197	King KR 86 ADF				
	a. First	_____	6.7	91.6	614
	b. Second	_____	9.7	107.0	1038
	c. Audio Amplifier	_____	0.8	51.0	41

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
198	King KR 86 ADF with KA 42B Loop and Sense Antenna				
	a. First	_____	7.6	91.6	696
	b. Second	_____	10.6	107.0	1134
	c. Audio Amplifier	_____	0.8	51.0	41
199	King KR 87 ADF Receiver and Indicator	_____	4.0	59.0	236
	a. KA 44 Antenna	_____	2.8	147.4	413
	b. KA 44B Antenna	_____	3.6	150.6	542
	c. Audio Amplifier	_____	0.8	51.0	41
	Cert. Basis - TSO C41c				
200	King KMA 20 ( ) Audio Panel				
	Cert. Basis - TSO C35c, C50b	_____	*3.7	70.8	262
201	King KMA-24 Audio Panel	_____	1.7	65.3	111
	Cert. Basis - TSO C35d, C50b				
203	King KT 76 ( )/78 ( ) Transponder				
	Cert. Basis - TSO C74b	_____	*3.1	58.1	180
204	Narco Comm 10A VHF Transceiver	_____	3.9	57.4	224
205	Narco Comm 11A VHF Transceiver				
	a. Single	_____	3.6	57.4	207
	b. Dual	_____	7.1	57.4	408
207	Narco Comm 11B VHF Transceiver				
	a. Single	_____	3.9	57 4	224
	b. Dual	_____	7.8	57.4	448

*Weight includes antenna and cable.

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
209	Narco Comm 111 VHF Transceiver				
	a. Single	_____	3.0	57.4	172
	b. Dual	_____	6.0	57.4	344
	Cert. Basis - TSO C37b, C38b				
211	Narco Comm IIIB VHF Transceiver				
	a. Single	_____	3.9	57.4	224
	b. Dual	_____	7.8	57.4	448
	Cert. Basis - TSO C37b, C38b				
213	Narco Comm 120 VHF Transceiver				
	a. Single	_____	4.8	56.9	273
	b. Dual	_____	8.6	57.4	494
	Cert. Basis - TSO C37b, C38b				
215	Narco Nav 10 VHF Receiver	_____	1.9	58.6	111
217	Narco Nav 11 VHF Receiver				
	a. Single	_____	2.8	58.6	164
	b. Dual	_____	5.6	58.6	328
219	Narco Nav 12 VHF Receiver	_____	3.4	58.6	199
221	Narco Nav 14 VHF Receiver	_____	2.5	57.4	144
223	Narco Nav 111 Cert. Basis - TSO C36c, C40a, C66a	_____	2.5	58.6	147
225	Narco Nav 112 Receiver Cert. Basis - TSO C36c, C40a, C66c, C34c	_____	3.3	58.6	193
227	Narco Nav 114 VHF Receiver Cert. Basis - TSO C38b, C40a, C36c, C34c, C66a	_____	2.5	57.4	144

*Weight includes marker antenna and cable.

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
229	Narco Nav 121 VHF Receiver				
	a. Single	_____	3.1	58.4	181
	b. Dual	_____	6.2	58.4	362
	Cert. Basis - TSO C36c, C40c, C66a				
231	Narco Nav 122 VHF Receiver				
	a. Single	_____	*5.1	99.4	507
	b. Dual	_____	*8.6	82.9	713
	Cert. Basis - TSO C35d, C36c, C40c, C66a				
233	Narco Nav 122A VHF Receiver				
	a. Single	_____	* 5.2	98.5	512
	b. Dual	_____	* 8.8	82.2	723
	Cert. Basis - TSO C34c, C35d, C36c, C40c, C66a				
235	Narco Nav 124A VHF Receiver				
	a. Single	_____	* 6.2	92.3	572
	b. Dual	_____	*10.9	77.2	841
	Cert. Basis - TSO C35d, C36c, C40a, C66a				
237	Narco ID 124 VOR/LOC/GS Indicator				
	a. Single	_____	1.2	60.5	73
	b. Dual	_____	2.4	60.5	145
	Cert. Basis - TSO C34c, C35d, C36c, C40c				
239	Narco UGR-2A Glide Slope				
	a. Single	_____	4.2	154.0	647
	b. Dual	_____	8.4	220.0	1848
	Cert. Basis - TSO C34b				
241	Narco UGR-3 Glide Slope	_____	4.2	154.0	647
243	Narco MBT-12-R, Marker Beacon	_____	3.1	69.1	214
245	Narco CP-125 Audio Selector Panel	_____	2.2	60.2	132

*Weight includes marker antenna and cable.

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
247	Narco CP135 Audio Selector Panel Cert. Basis - TSO C50b	_____	2.2	55.0	121
249	Narco CP135M Audio Selector Panel Cert. Basis - TSO C50b, C35d	_____	* 3.7	114.3	423
251	Narco DME-190	_____	** 5.9	61.0	360
253	Narco DME-190 TSO Cert. Basis - TSO C66a	_____	** 5.9	60.9	359
255	Narco DME-195 Receiver and Indicator Cert. Basis - TSO C66a	_____	**13.2	154.5	2039
257	Narco ADF-140 a. Single b. Dual Cert. Basis - TSO C41c	_____ _____	6.0 ***17.9	91.2 107.6	547 1926
259	Narco ADF-141 a. Single b. Dual Cert. Basis - TSO C41c	_____ _____	6.0 ***17.9	91.2 107.6	547 1926
261	Narco AT50A Transponder Cert. Basis - TSO C74b a. Narco AR-500 Altitude Encoder Cert. Basis - TSO C88	_____ _____	** 3.0 1.0	57.3 51.5	172 52
263	Narco AT150 Transponder Cert. Basis - TSO C74c a. Narco AR-500 Altitude Encoder Cert. Basis - TSO C88	_____ _____	** 3.0 1.0	57.3 51.5	172 52

*Weight includes marker antenna and cable.
**Weight includes antenna and cable.
***Weight includes dual antenna and cable.

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
273	Antenna and Cable				
	a. Nav Receiving VRP-37 or AV12-PPR	_____	1.6	171.3	274
	b. #1 VHF Comm VTP-17	_____	0.7	125.7	88
	c. #2 VHF Comm VTP-17	_____	0.8	147.5	118
	d. Glide Slope (Single) GS401 or CI 104	_____	0.9	120.0	108
	e. Glide Slope (Dual) GS401 or CI 104	_____	2.8	154.0	431
	f. Single ADF Sense 99841	_____	0.4	150.0	60
	Piper Dwg. 99461				
275	Anti Static Antenna and Cable				
	a. #1 VHF Comm PS50040-18	_____	1.4	144.3	202
	b. #2 VHF Comm PS50040-18	_____	1.5	170.7	256
	c. Single ADF Sense 79160	_____	0.5	147.5	74
276	Marker Beacon Antenna Piper PS50040-15 King KA-23 or Narco VMA-15 or Commant CI-102	Included as part of marker beacon installation			
277	Marker Beacon Antenna Comant CI 102 Piper Dwg. 39737-4	_____	*1.2	175.0	210
278	Emergency Locator Transmitter (C.C.C. Model CIR-11-2)	_____	1.7	236.2	402
	a. Antenna and Coax	_____	0.2	224.4	45
	b. Shelf and Access Hole	_____	0.5	235.4	118
	Cert. Basis - TSO C91				
279	Emergency Locator Transmitter (Narco Model ELT-10)	_____	3.5	236.2	827
	a. Antenna and Coax	_____	0.3	224.4	67
	b. Shelf and Access Hole	_____	0.5	235.4	118
	Cert. Basis - TSO C91				
280	Microphone				
	a. Piper Dwg. 68856-10	_____	0.3	69.9	21
	b. Piper Dwg. 68856-11	_____	0.6	69.9	42
	c. Piper Dwg. 68856-12	_____	0.3	69.9	21

(m) Radio Equipment
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
281	Boom Microphone, Headset Piper Dwg. 37921-2	_____	0.3	80.5	24
283	Cabin Speaker Piper Dwg. 99220	_____	1.1	99.0	109
285	Headset Piper Dwg. 68856-10	_____	0.5	60.0	30

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(n) Miscellaneous
(Optional Equipment)

Item No.		Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
321		Zinc Chromate Finish Piper Dwg. 79700	_____	5.0	158.0	790
323		Stainless Steel Control Cables Piper Dwg. 79700	_____	_	_	_
325		Air Conditioner Piper Dwg. 99575-4	_____	68.3	103.6	7076
327		Overhead Vent System				
		a. Piper Dwg. 76304-9	_____	6.4	159.6	1022
		b. Piper Dwg. 76304-15	_____	5.7	148.9	849
329		Overhead Vent System with Ground Ventilating Blower				
		a. Piper Dwg. 76304-10	_____	14.9	172.2	2566
		b. Piper Dwg. 76304-16	_____	14.2	168.5	2393
331		Rear Seat Vents Piper Dwg. 68556	_____	2.5	98.0	245
333		Assist Step Piper Dwg. 65384	_____	1.8	156.0	281
335		Super Cabin Sound Proofing Piper Dwg. 79030-2	_____	18.1	86.8	1571
337	C	Adjustable Front Seat (Left) Piper Dwg. 79591-0/79591-2	_____	*6.6	80.3	530
339		Adjustable Front Seat (Right) Piper Dwg. 79591-1/79591-3	_____	*6.6	79.6	525
341		Headrests (2) Front Piper Dwg. 79337-18	_____	2.2	94.5	208
342		Shoulder Harness Inertia (Front) (2) Piper Dwg. PS50039-4-20 Pacific Scientific 1107447-13 (Black)	_____	1.3	119.5	155

*Weight and moment difference between standard and optional equipment.

(n) Miscellaneous
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
343	Inertia Safety Belts (Rear) (2) 0.8 lbs. each, Piper PS50039-4-14 Pacific Scientific 1107319-01 American Safety Eqpt. Corp. 500853-401 (Black)	_____	1.6	140.3	224
344	Shoulder Harness - Fixed (Rear) (2) Piper Dwg. PS50039-4-22 American Safety Eqpt. Corp. 501385-403 Davis Acft. Prod. Inc. FDC-7275-16-2 (Black)	_____	1.1	140.3	154
345	Shoulder Harness - Inertia (Rear) (2) Piper Dwg. PS50039-4-19 Pacific Scientific 1107447-01 (Black)	_____	1.3	140.3	187
346	Sun Visors Piper Dwg. 66991-0	_____	1.5	85.0	128
347	Assist Strap Piper Dwg. 79455	_____	0.2	109.5	22
349	Curtain and Rod Installation Piper Dwg. 67955-2	_____	4.2	124.0	521
351	Luxurious Interior Piper Dwg. 67952-4	_____	*14.5	98.3	1425
352	Deluxe Carpeting Piper Dwg. 66801	_____	*2.6	97.8	254
355	Fire Extinguisher a. Piper Dwg. 76167-2, Scott 42211-00	_____	4.6	71.0	327
	b. Piper Dwg. 37872-2, Graviner HA 1014-01	_____	5.6	57.9	324
357	Tow Bar Piper Dwg. 99458	_____	1.3	156.0	203

*Weight and moment difference between standard and optional equipment.

(n) Miscellaneous
(Optional Equipment) (cont)

Item No.	Item	Mark if Instl.	Weight (Pounds)	Arm (In.) Aft Datum	Moment (Lb-In.)
361	Locking Gas Cap Piper Dwg. 39830-2	_____	*0.1	94.1	9

*Weight and moment difference between standard and optional equipment.

TOTAL OPTIONAL EQUIPMENT	_____	_____	_____

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SECTION 7

DESCRIPTION AND OPERATION OF THE AIRPLANE AND ITS SYSTEMS

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SECTION 7

DESCRIPTION AND OPERATION OF THE AIRPLANE AND ITS SYSTEMS

7.1 THE AIRPLANE

The Cherokee Warrior II is a single-engine, fixed gear monoplane of all metal construction with low semi-tapered wings. It has four place seating and a two hundred pound baggage capacity.

7.3 AIRFRAME

The primary structure, with the exception of the steel tube engine mount, steel landing gear struts and isolated areas, is of aluminum alloy construction. Fiberglass and thermoplastic are used extensively in the extremities - the wing tips, the engine cowling, etc. - and in nonstructural components throughout the airplane.

The fuselage is a conventional semi-monocoque structure. On the right side of the airplane is a cabin door for entrance and exit and a baggage door to provide loading into the 24 cubic foot compartment.

The wing is a conventional semi-tapered design incorporating a laminar flow, NACA $65_2$415, airfoil section. The cantilever wings are attached to each side of the fuselage by insertion of the butt ends of the main spars into a spar box carry-through which is an integral part of the fuselage structure. The spar box carry-through structure, located under the rear seat, provides in effect a continuous main spar with splices at each side of the fuselage. There are also fore and aft attachments at the rear and at an auxiliary front spar.

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7.5 ENGINE AND PROPELLER

The PA-28- 161 is powered by a four cylinder, direct drive, horizontally opposed engine rated at 160 H P at 2700 RPM. It is equipped with a starter, a 60 amp 14 volt alternator, a shielded ignition, two magnetos, vacuum pump drive, a fuel pump, a wetted polyurethane foam induction air filter.

The engine compartment is accessible for inspection through top-hinged side panels on either side of the engine cowlings. The engine cowlings are cantilever structures attached at the fire wall The engine mounts are constructed of steel tubing, and dynafocal mounts are provided to reduce vibration.

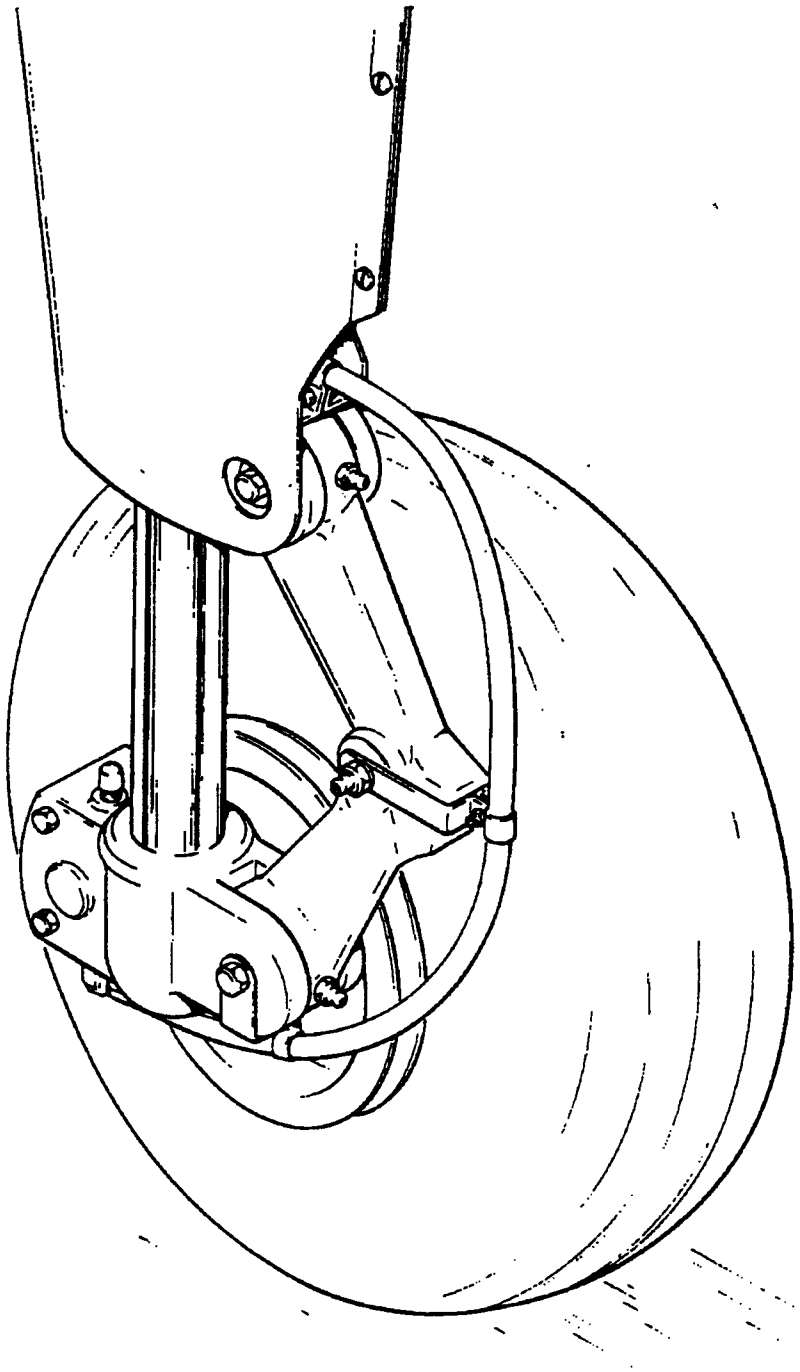
The exhaust system is constructed of stainless steel and incorporates dual mufflers with heater shrouds to supply heated air for the cabin, the defroster system and the carburetor deicing system.

An oil cooler is located on the left rear of the engine mounted to the engine baffling. Engine cooling air, which is picked up in the nose section of the engine cowling and carried through the baffling, is utilized on the left side for the oil cooler. A winterization plate is provided to restrict air during winter operation (refer to paragraph 8.29).

Engine air enters on either side of the propeller through openings in a nose cowling and is carried through the engine baffling around the engine and oil cooler. Air for the muffler shroud is also picked up from the nose cowling and carried through a duct to the shroud. Carburetor induction air enters a chin scoop on the lower right cowling and is passed through a wetted polyurethane filter to the carburetor air box. Heated air enters the carburetor air box through a hose connected to the heater shroud.

A fixed pitch propeller is installed as standard equipment. The propeller has a 74 inch diameter with a 58 or 60 inch pitch. The pitch is determined at 75% of the diameter. The propeller is made of an aluminum alloy construction.

The pilot should read and follow the procedures recommended in the Lycoming Operator's Manual for this engine in order to obtain maximum engine efficiency and time between engine overhauls.



MAIN WHEEL ASSEMBLY

Figure 7-1

7.7 LANDING GEAR

The fixed gear PA-28-161 is equipped with a Cleveland 5.00 x 5 wheel on the nose gear and a Cleveland 6.00 x 6 wheel on each main gear (Figure 7-1). Cleveland single disc hydraulic brake assemblies are provided on the main gear. The nose gear has a 5.00 x 5 four ply tire, while the main wheel assemblies have 6.00 x 6 four ply tires. At gross weight, the main gear tires require a pressure of 24 psi, and the nose gear tire requires a pressure of 30 psi.

The nose gear is steerable through a 30 degree arc each side of center by the use of the rudder pedals and toe brakes. A spring device is incorporated for rudder centering and to provide rudder trim. A bungee assembly on the nose gear steering mechanism reduces ground steering effort and dampens shocks and bumps during taxiing. The steering mechanism also incorporates a shimmy dampener.

The three struts are of the air-oil type with the normal static load extension being 3.25 inches for the nose gear and 4.50 inches for the main gear.

The brakes are actuated by toe brake pedals which are attached to the rudder pedals or by a hand lever and master cylinder located below and behind the center of the instrument sub panel. Hydraulic cylinders are located above each pedal and adjacent to the hand brake lever. The brake fluid reservoir is installed on the top left front face of the fire wall. The parking brake is incorporated in the master cylinder and is actuated by pulling back on the brake lever and depressing the knob attached to the left side of the handle. To release the parking brake, pull back on the brake lever to disengage the catch mechanism and allow the handle to swing forward (refer to Figure 7-5).

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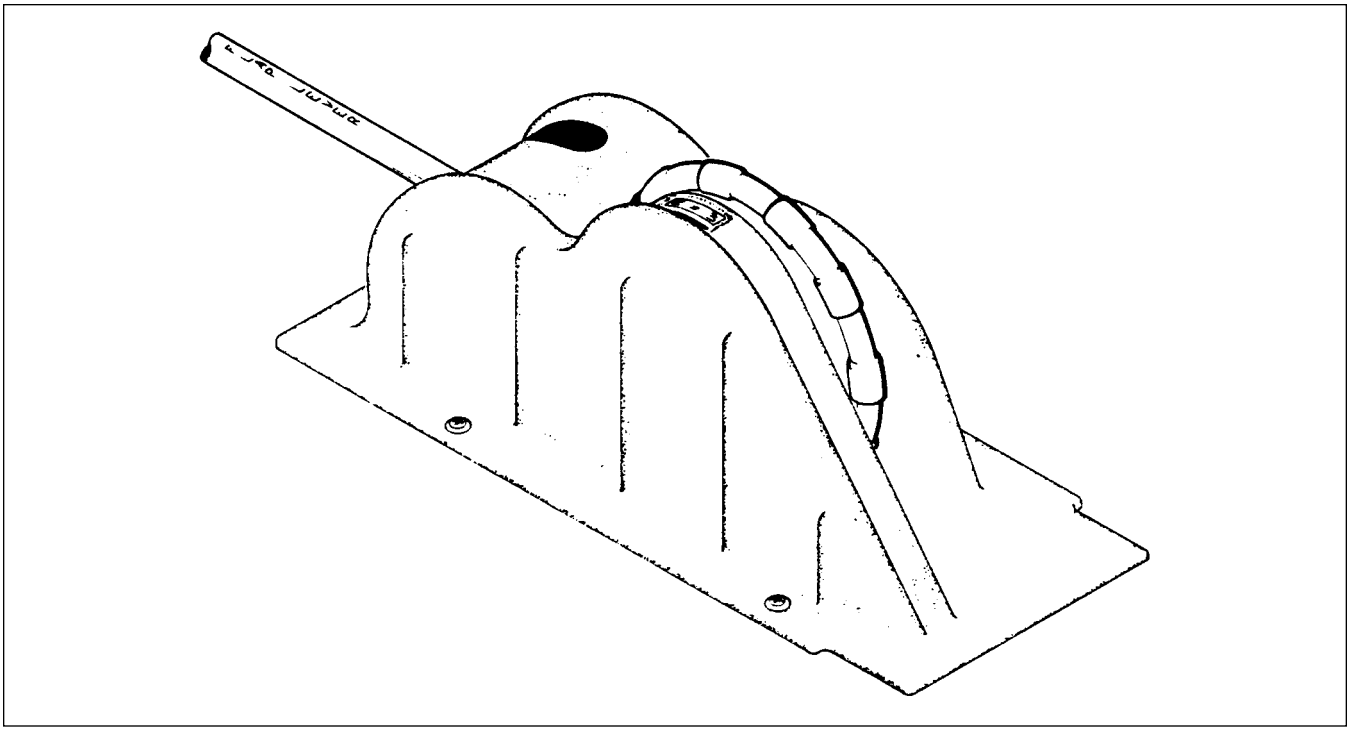


FLIGHT CONTROL CONSOLE

Figure 7-3

7.9 FLIGHT CONTROLS

Dual flight controls are provided as standard equipment. The flight controls actuate the control surfaces through a cable system.

The horizontal surface (stabilator) is of the flying tail design with a trim tab mounted on the trailing edge. This tab serves the dual function of providing trim control and pitch control forces. The trim tab is actuated by a trim control wheel located on the control console between the front seats (Figure 7-3). Forward rotation of the wheel gives nose down trim and aft rotation gives nose up trim.

The rudder is conventional in design and incorporates a rudder trim. The trim mechanism is a spring loaded recentering device. The trim control is located on the right side of the pedestal below the throttle quadrant (refer to Figure 7-5). Turning the trim control clockwise gives nose right trim and counterclockwise rotation gives nose left trim.

Manually controlled flaps are provided on the PA-28-161. The flaps are balanced and spring loaded to return to the retracted (up) position. A control handle, which is located between the two front seats on the control console (Figure 7-3), extends the flaps by the use of a control cable. To extend the flaps, the handle is pulled up to the desired flap setting of 10, 25 or 40 degrees. To retract, depress the button on the end of the handle and lower the control. When extending or retracting flaps, there is a pitch change in the airplane. This pitch change can be corrected either by stabilator trim or increased control wheel force. When the flaps are in the retracted (up) position the right flap, provided with an over-center lock mechanism, acts as a step.

NOTE

The right flap will support a load only in the fully retracted (up) position. When the flap is to be used as a step, make sure the flaps are in the retracted (up) position.

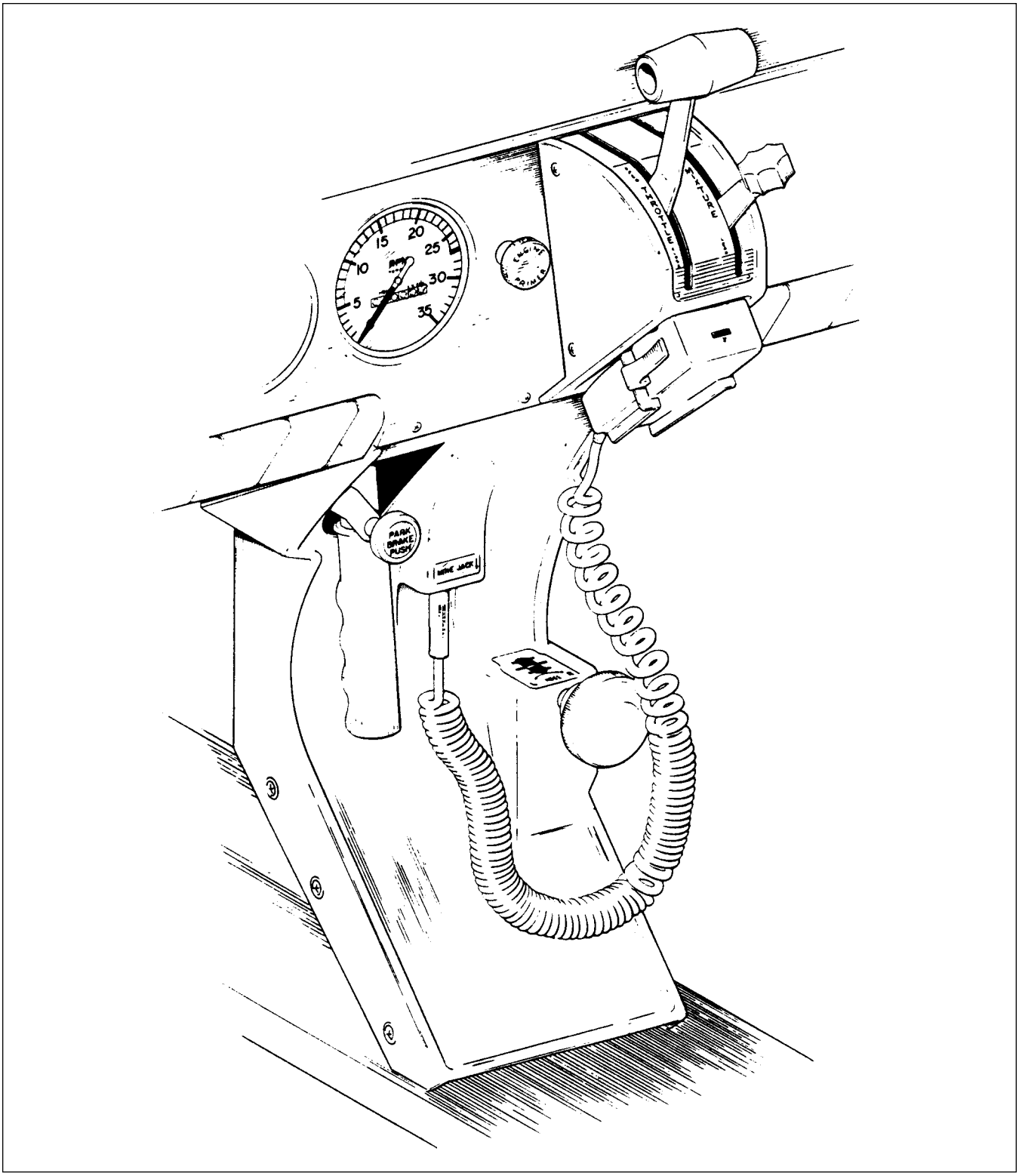
7.11 ENGINE CONTROLS

Engine controls consist of a throttle control and a mixture control lever. These controls are located on the control quadrant on the lower center of the instrument panel (Figure 7-5) where they are accessible to both the pilot and the copilot. The controls utilize teflon-lined control cables to reduce friction and binding.

The throttle lever is used to adjust engine RPM. The mixture control lever is used to adjust the air to fuel ratio. The engine is shut down by the placing of the mixture lever in the full lean position. For information on the leaning procedure, see the Avco-Lycoming Operator's Manual.

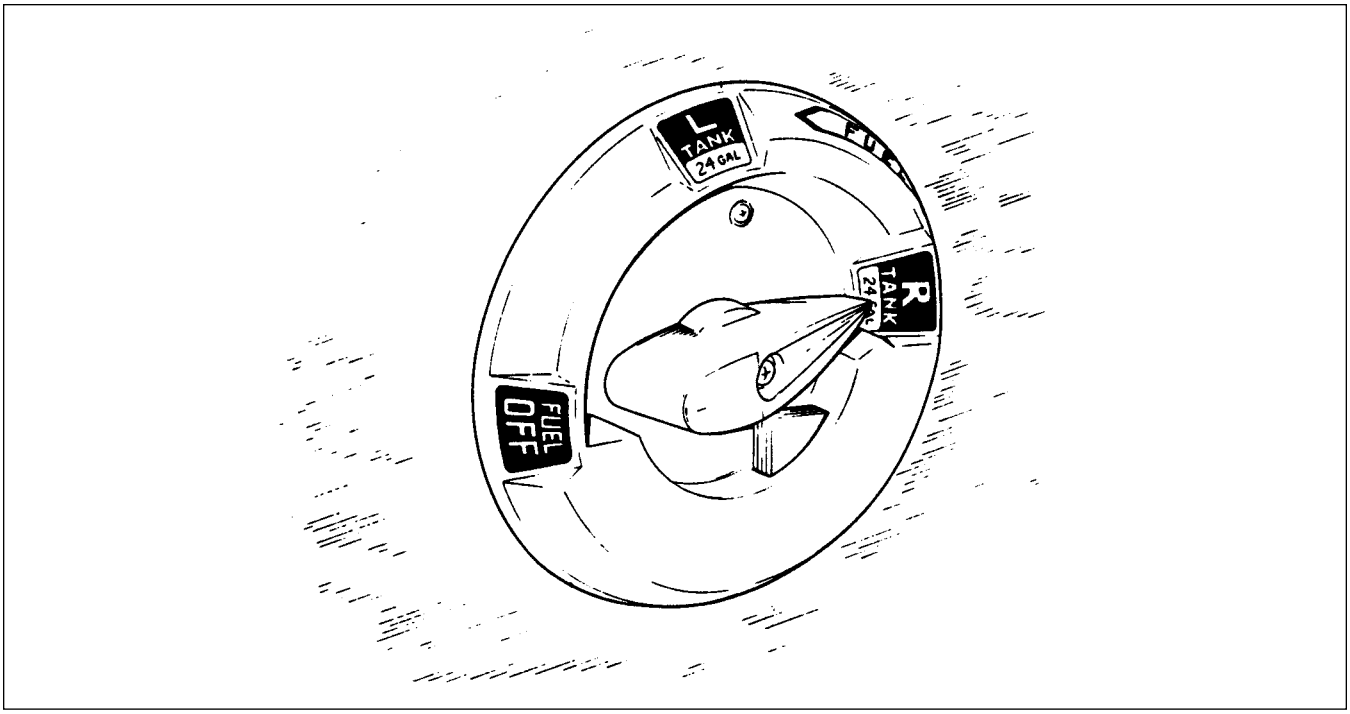
The friction adjustment lever on the right side of the control quadrant may be adjusted to increase or decrease the friction holding the throttle and mixture controls or to lock the controls in a selected position.

The carburetor heat control lever is located to the right of the control quadrant on the instrument panel. The control is placarded with two positions: "ON" (down), "OFF" (up).



CONTROL QUADRANT AND CONSOLE

Figure 7-5



FUEL SELECTOR

Figure 7-7

7.13 FUEL SYSTEM

Fuel is stored in two twenty-five gallon (24 gallons usable) fuel tanks, giving the airplane a total capacity of fifty U.S. gallons (48 gallons usable). Each tank is equipped with a filler neck indicator tab to aid in determining fuel remaining when the tanks are not full. Usable capacity to the bottom of the indicator tab is 17 gallons. The tanks are secured to the leading edge of each wing with screws and nut plates. This allows removal for service or inspection.

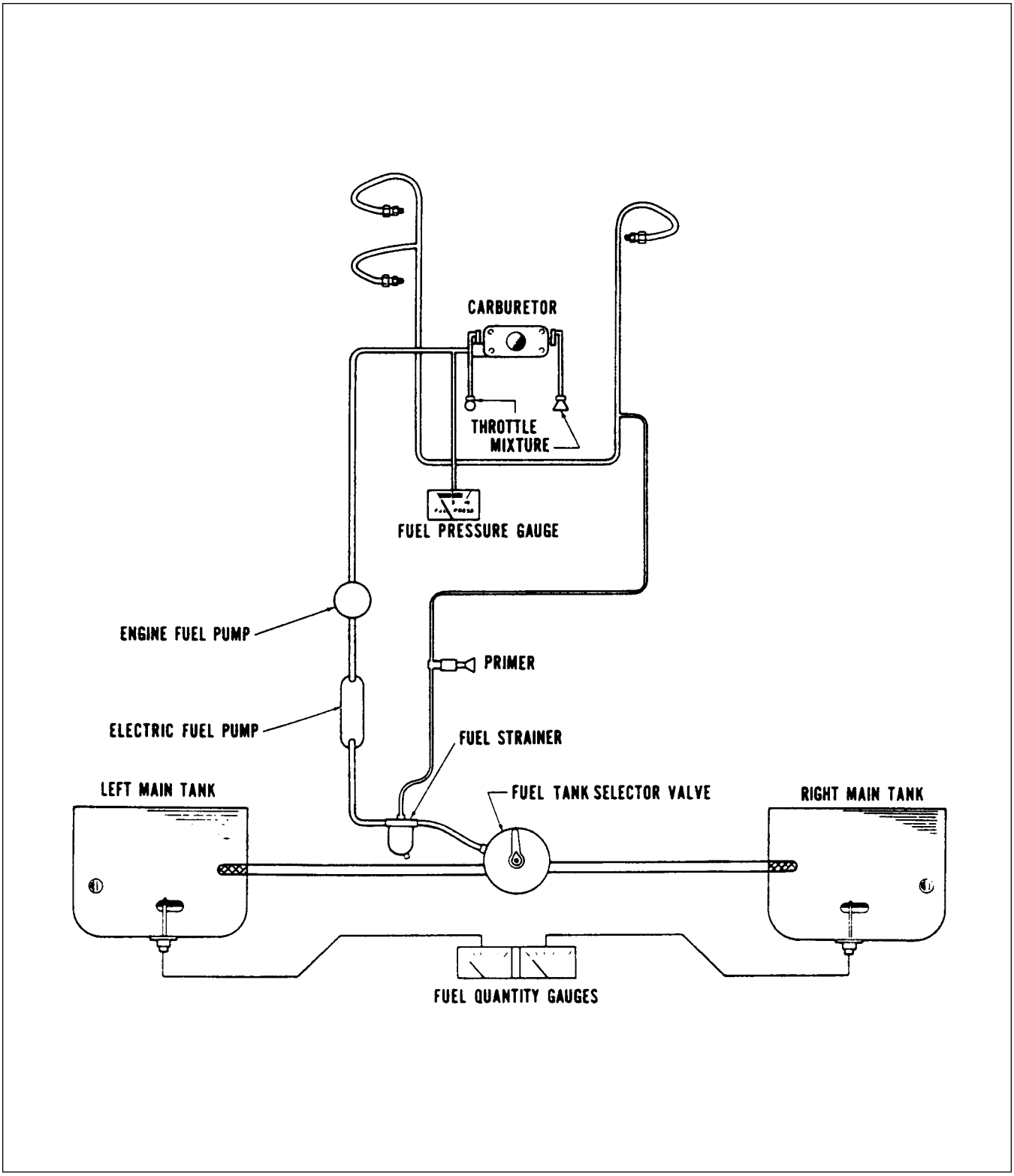
The fuel tank selector control (Figure 7-7) is located on the left side panel forward of the pilot's seat. The button on the selector cover must be depressed and held while the handle is moved to the OFF position. The button releases automatically when the handle is moved back to the ON position.

An auxiliary electric fuel pump is provided in case of the failure of the engine driven pump. The electric pump should be ON for all takeoffs and landings and when switching tanks. The fuel pump switch is located in the switch panel above the throttle quadrant.

The fuel drains should be opened daily prior to first flight to check for water or sediment. Each tank has an individual drain at the bottom, inboard rear corner.

A fuel strainer, located on the lower left front of the fire wall, has a drain which is accessible from outside the nose section. The strainer should also be drained before the first flight of the day. Refer to paragraph 8.21 for the complete fuel draining procedure.

Optional locking fuel caps are available for all fillers. A single key will fit fuel caps, cabin door and baggage door compartments.



FUEL SYSTEM SCHEMATIC

Figure 7-9

Fuel quantity and fuel pressure gauges are mounted in a gauge cluster located on the left side of the instrument panel to the right of the control wheel (refer to Figure 7-15).

An optional engine priming system is available to facilitate starting. The primer pump is located to the immediate left of the throttle quadrant (refer to Figure 7-5).

7.15 ELECTRICAL SYSTEM

The electrical system includes a 14-volt, 60 amp alternator, a 12-volt battery, a voltage regulator. an overvoltage relay and a master switch relay (Figure 7-11). The battery is mounted in a thermoplastic box immediately aft of the main spar on the right side of the fuselage below the rear passenger seat. The regulator and overvoltage relay are located on the forward left side of the fuselage behind the instrument panel.

Electrical switches are located on the right center instrument panel (refer to Figure 7-15) and the circuit breakers are located on the lower right instrument panel (refer to Figure 7-13). A rheostat switch on the left side of the switch panel controls the navigational lights and the radio lights. The similar switch on the right side controls and dims the panel lights.

Standard electrical accessories include a starter, electric fuel pump, stall warning indicator, cigar lighter, fuel gauge, ammeter, and annunciator panel.

The annunciator panel includes alternator and low oil pressure indicator lights. When the optional gyro system is installed, the annunciator panel also includes a low vacuum indicator light. The annunciator panel lights are provided only as a warning to the pilot that a system may not be operating properly, and that he should check and monitor the applicable system gauge to determine when or if any necessary action is required.

Optional electrical accessories include navigation lights, anti-collision light, landing light, instrument lighting. and cabin dome light. Circuits will handle the addition of communications and navigational equipment.

An optional light, mounted in the overhead panel, provides instrument and cockpit lighting for night flying. The light is controlled by a rheostat switch located adjacent to the light. A map light window in the lens is actuated by an adjacent switch.

WARNING

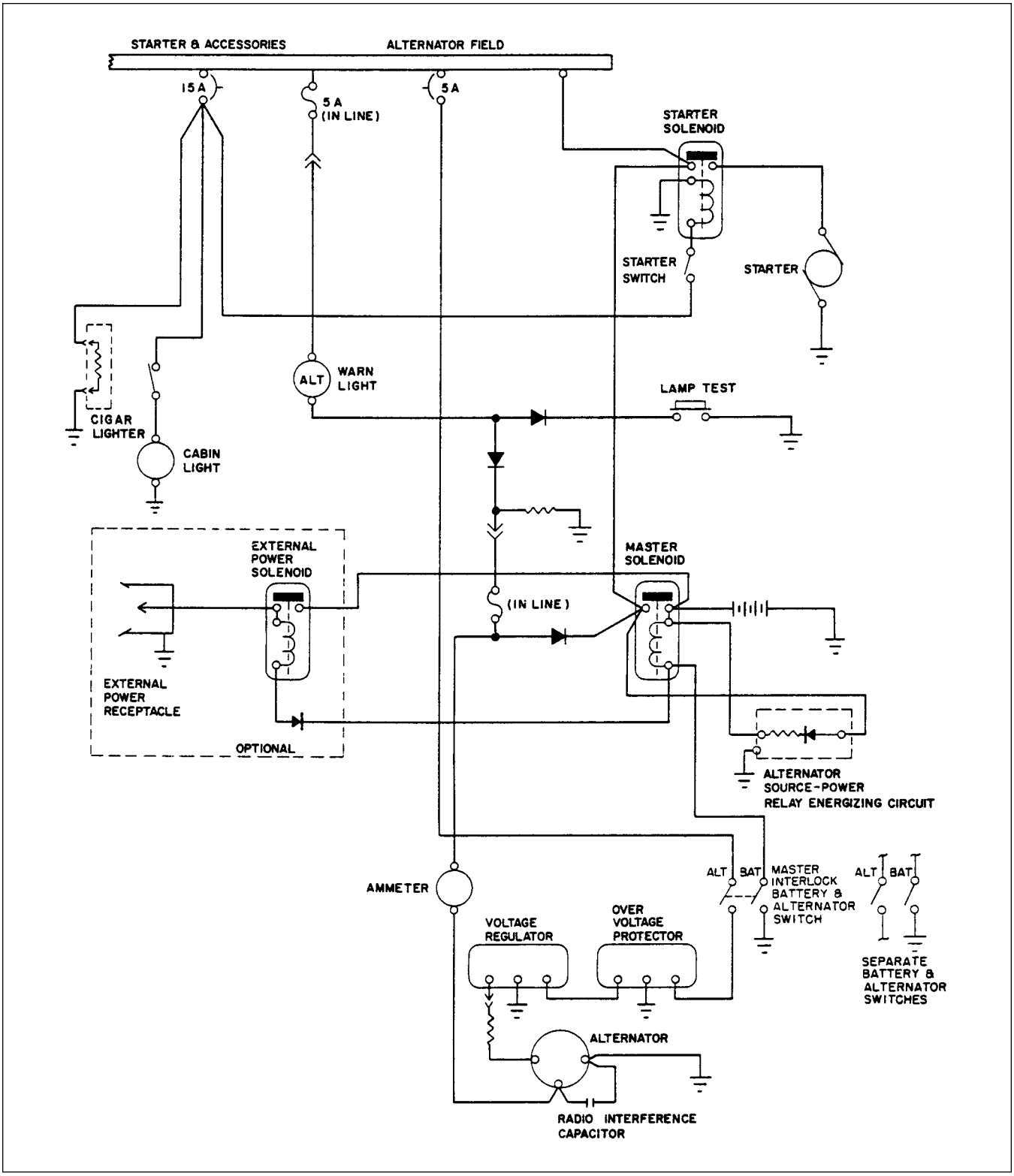
Anti-collision lights should not be operating when flying through cloud, fog or haze, since the reflected light can produce spatial disorientation. Strobe lights should not be used in close proximity to the ground such as during taxiing, takeoff or landing.

NOTE

On airplanes with interlocked BAT and ALT switches, the ALT switch is mechanically interlocked with the BAT switch. When ALT switch is turned ON, the BAT switch will also be turned ON. On airplanes with separate BAT and ALT switch operation, the switches may be positioned independently as desired.

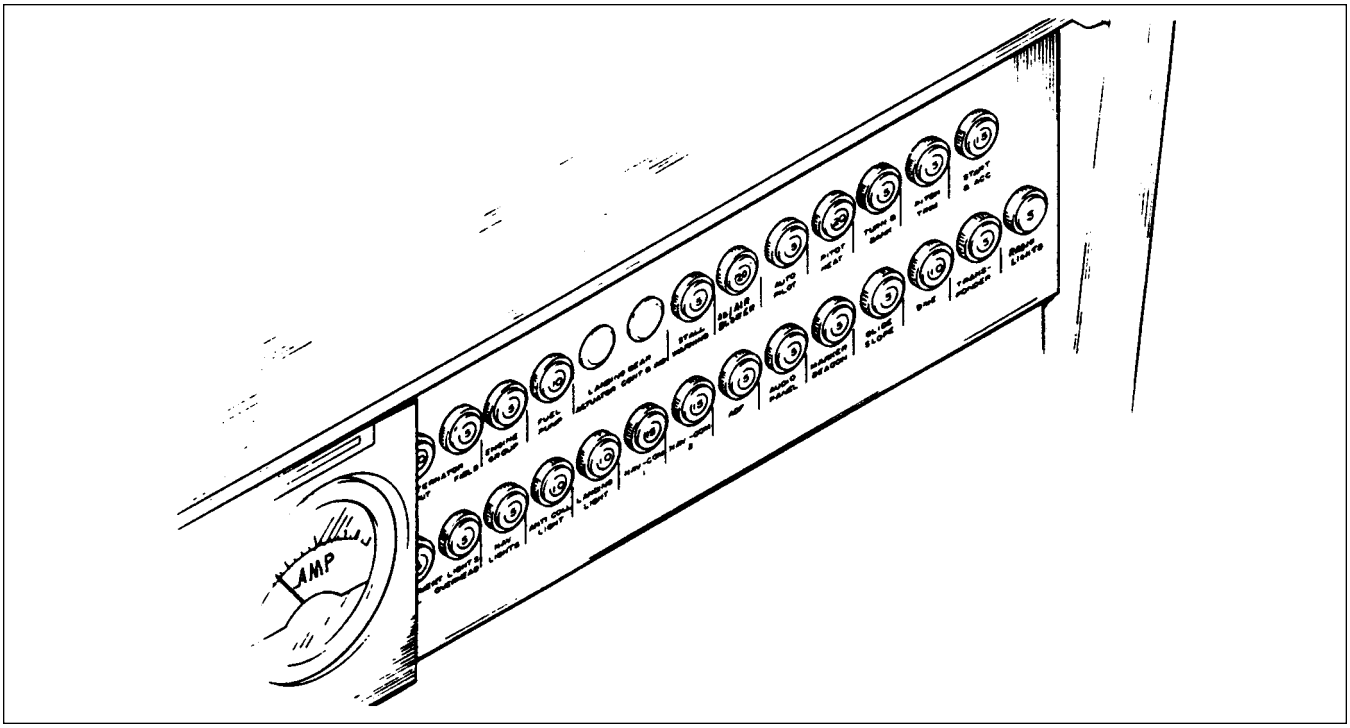
Unlike previous generator systems, the ammeter as installed does not show battery discharge; rather, it indicates the electrical load on the alternator in amperes. With all the electrical equipment off and the master switch on, the ammeter will indicate the charging rate of the battery. As each electrical unit is switched on, the ammeter will indicate the total ampere draw of all the units including the battery. For example, the average continuous load for night flight with radios on is about 30 amperes. This 30 ampere value plus approximately 2 amperes for a fully charged battery will appear continuously under these flight conditions. The amount of current shown on the ammeter will tell immediately if the alternator system is operating normally, as the amount of current shown should equal the total amperage drawn by the electrical equipment which is operating.

For abnormal and/or emergency operation and procedures, see Section 3.



ALTERNATOR AND STARTER SCHEMATIC

Figure 7-11



CIRCUIT BREAKER PANEL

Figure 7-13

7.17 VACUUM SYSTEM*

The vacuum system is designed to operate the air driven gyro instruments. This includes the directional and attitude gyros when installed. The system consists of an engine driven vacuum pump, a vacuum regulator, a filter and the necessary plumbing.

The vacuum pump is a dry type pump. A shear drive protects the pump from damage. If the drive shears, the gyros will become inoperative.

A vacuum gauge, mounted on the far right instrument panel provides a pilot check for the system during operation. A decrease in pressure in a system that remained constant over an extended period may indicate a dirty filter, dirty screens, possibly a sticky vacuum regulator or leak in the system (a low vacuum indicator light is provided in the annunciator panel). Zero pressure would indicate a sheared pump drive, defective pump, possibly a defective gauge or collapsed line. In the event of any gauge variation from the norm, the pilot should have a mechanic check the system to prevent possible damage to the system components or eventual failure of the system.

A vacuum regulator is provided in the system to protect the gyros. The valve is set so the normal vacuum reads 5.0 ± .1 inches of mercury, a setting which provides sufficient vacuum to operate all the gyros at their rated RPM. Higher settings will damage the gyros and with a low setting the gyros will be unreliable. The regulator is located behind the instrument panel. Vacuum pressure, even though set correctly, can read lower at very high altitude (above 12,000 ft), and at low engine RPM (usually on approach or during training maneuvers. This is normal and should not be considered a malfunction.

*Optional equipment

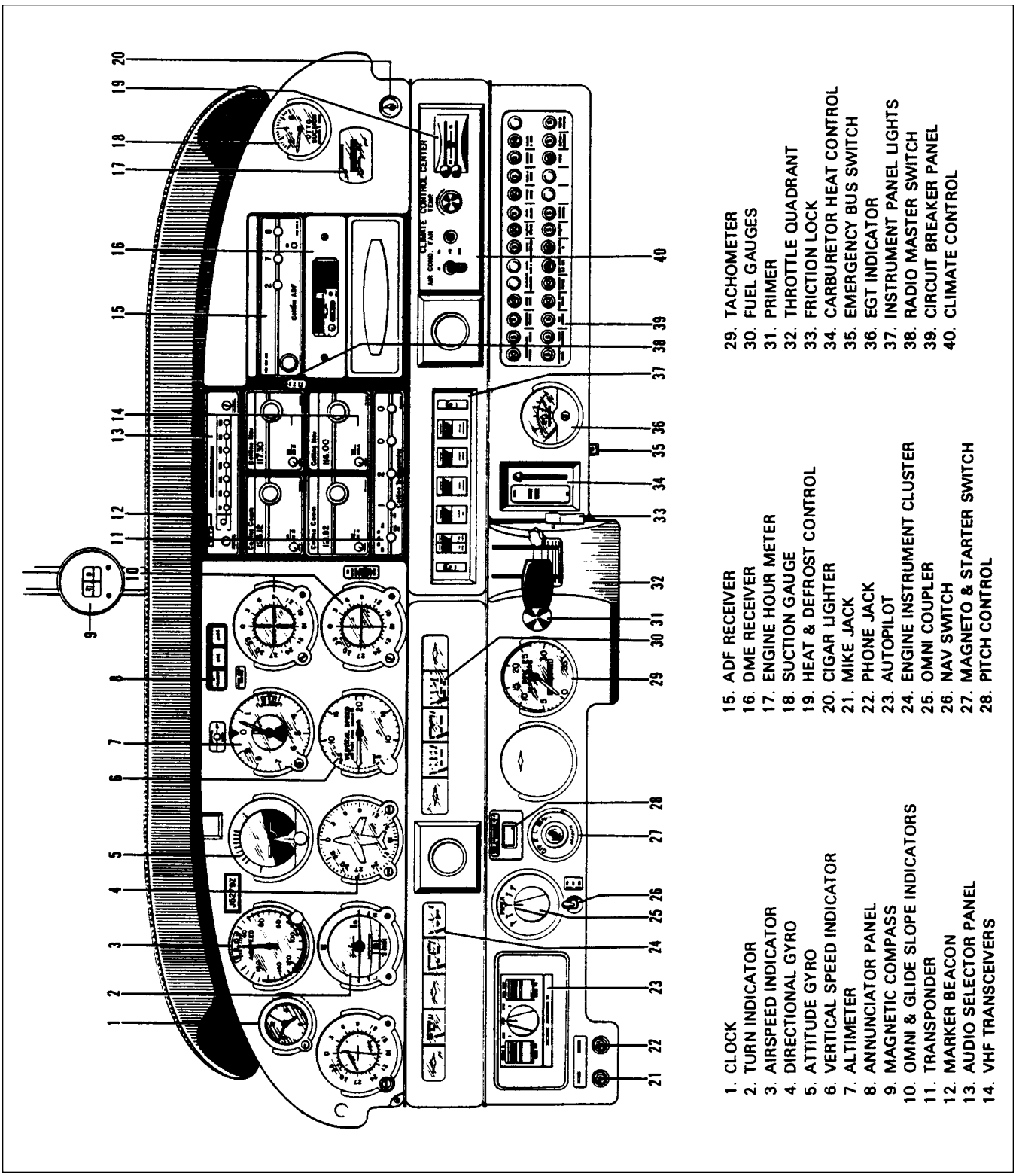
7.19 INSTRUMENT PANEL

The instrument panel (Figure 7-15) is designed to accommodate instruments and avionics equipment for VFR and IFR flights.

The radios and the circuit breakers are located on the upper and lower right panel respectively, and have circuits provided for the addition of optional radio equipment. An optional radio master switch is located near the top of the instrument panel between the radio stacks. It controls the power to all radios through the aircraft master switch. An emergency bus switch is also provided to provide auxiliary power to the avionics bus in event of a radio master switch circuit failure. The emergency bus switch is located behind the lower right shin guard left of the circuit breaker panel. An engine cluster is located to the right of the pilot control wheel and includes a fuel pressure gauge, a right and left main fuel quantity gauge, an oil temperature gauge and an oil pressure gauge.

Standard instruments include a compass, an airspeed indicator, a tachometer, an altimeter, an ammeter, an engine cluster, and an annunciator panel. The compass is mounted on the windshield bow in clear view of the pilot. The annunciator panel is mounted in the upper instrument panel to warn the pilot of a possible malfunction in the alternator, oil pressure, or vacuum systems.

Instrument options available for the panel includes a suction gauge, vertical speed indicator, attitude gyro, directional gyro, clock, tru-speed indicator and a turn and slip indicator or turn coordinator. The attitude gyro and directional gyro are vacuum operated through the use of a vacuum pump installed on the engine, while the turn and slip indicator is electrically operated. The vacuum suction gauge is on the far right of the instrument panel.



INSTRUMENT PANEL

Figure 7-15

7.21 PITOT-STATIC SYSTEM

The system supplies both pitot and static pressure for the airspeed indicator, altimeter, and the optional vertical speed indicator (Figure 7-17)

Pitot and static pressure are picked up by a pitot head installed on the bottom of the left wing and carried through pitot and static lines within the wing and fuselage to the gauges on the instrument panel.

An alternate static source is available as optional equipment. The control valve is located below the left side of the instrument panel. When the valve is set in the alternate position, the altimeter, vertical speed indicator and airspeed indicator will be using cabin air for static pressure. The storm window and cabin vents must be closed and the cabin heater and defroster must be on during alternate static source operation. The altimeter error is less than 50 feet unless otherwise placarded.

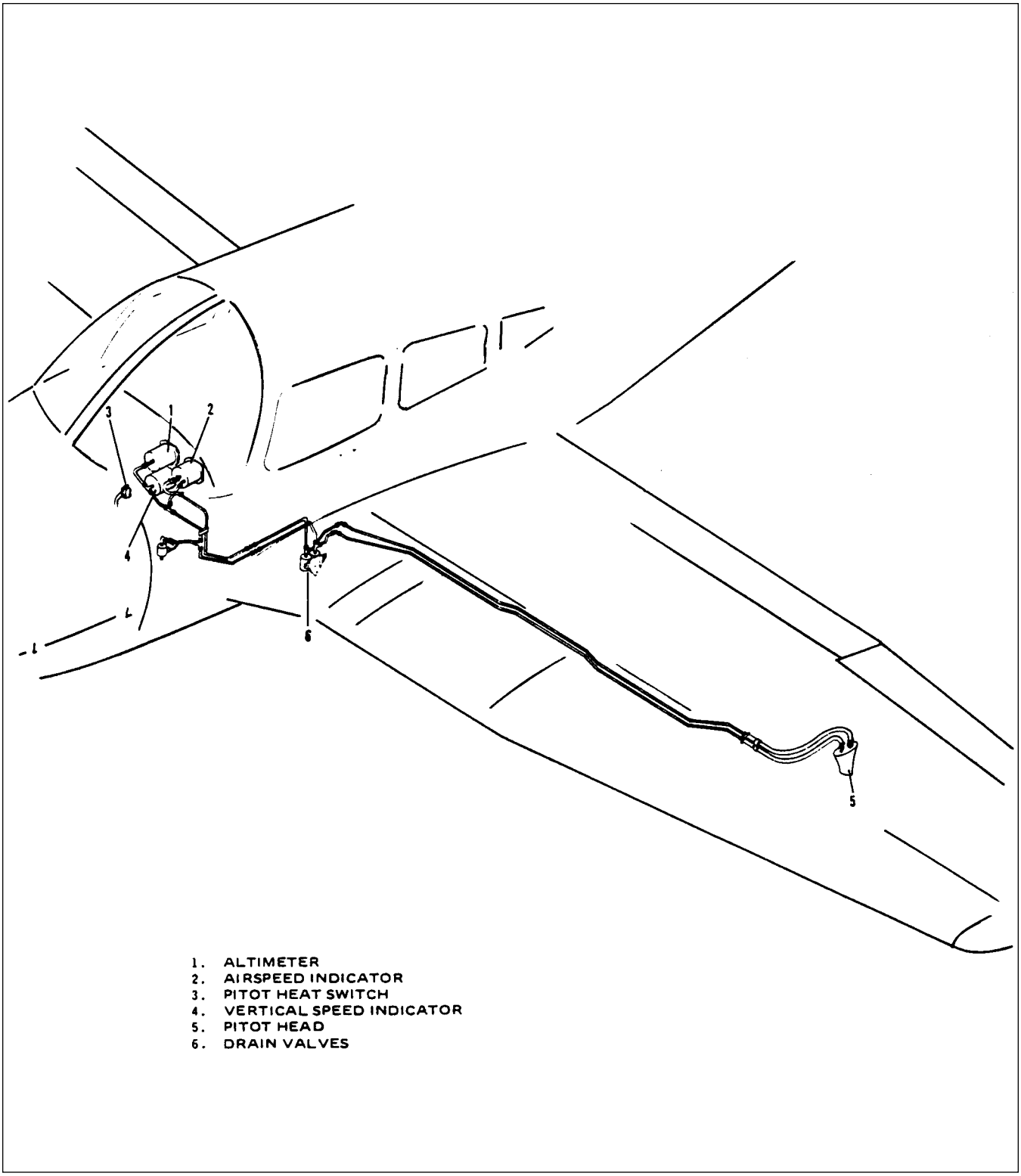
Both the pitot and static lines can be drained through separate drain valves located on the left lower side of the fuselage interior.

A heated pitot head, which alleviates problems with icing and heavy rain, is available as optional equipment. The switch for the heated pitot head is located on the electrical switch panel to the left of the right control wheel.

To prevent bugs and water from entering the pitot and static pressure holes, a cover should be placed over the pitot head. A partially or completely blocked pitot head will give erratic or zero readings on the instruments.

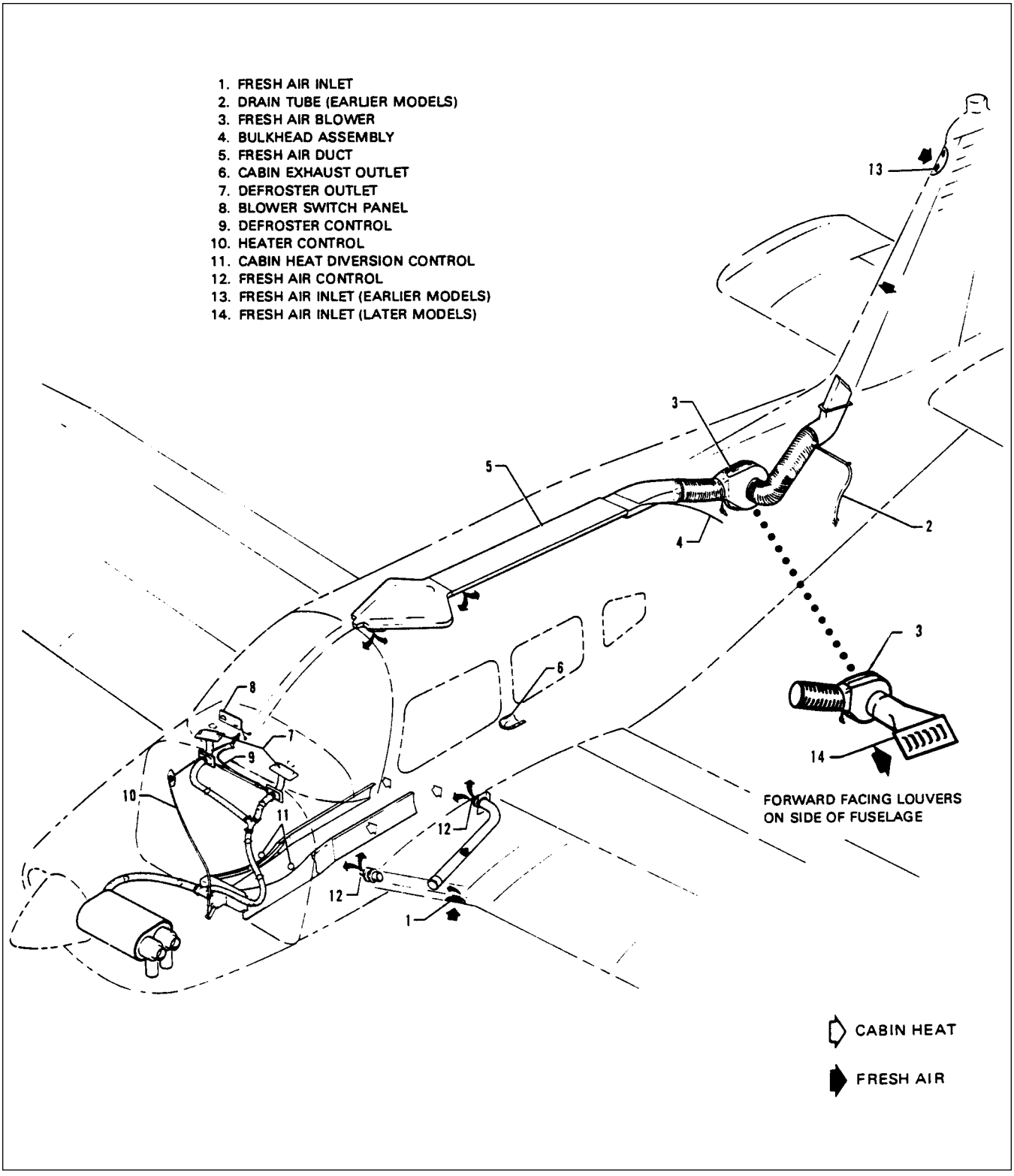
NOTE

During the preflight, check to make sure the pitot cover is removed.



PITOT-STATIC SYSTEM

Figure 7-17



HEATING AND VENTILATING SYSTEM

Figure 7-19

7.23 HEATING AND VENTILATING SYSTEM

Heat for the cabin interior and the defroster system is provided by a shroud attached to the muffler (Figure 7-19). The amount of heat can be regulated with the controls located on the far right side of the instrument panel.

The airflow between front and rear seats can be regulated by the heat diversion controls located on either side of the console atop the heat ducts.

CAUTION

When cabin heat is operated, heat duct surface becomes hot. This could result in burns if arms or legs are placed too close to heat duct outlets or surface.

Fresh air inlets are located in the leading edges of the wings near the fuselage. At each front seat location there is an adjustable fresh air outlet on the side of the cabin near the floor. Rear seat vents are optional. Cabin air is exhausted through an outlet located below the rear seat.

An optional overhead ventilating system with outlets over each seat is also available. An additional option to aid in fresh air circulation on models without air conditioning is a cabin air blower to force air through the overhead vent system. This blower is operated by a fan switch with four positions - “OFF,” “LOW,” “MED,” and “HIGH.” The switch is located on the right side of the instrument panel with the heater and defroster controls.

7.25 CABIN FEATURES

For ease of entry and exit and for pilot-passenger comfort, the front seats are adjustable fore and aft. The right front seat tilts forward to allow easy entry to the rear seats. The cabin interior includes a pilot storm window, ash trays and armrests on each front seat, two map pockets and pockets on the backs of the front seats.

The front seats can be equipped with optional headrests and optional vertical adjustment.

Seat belts are standard equipment for both front and rear seats. A single strap adjustable shoulder harness located above the side window, protects each front seat occupant. Optional shoulder straps for the rear seat occupants are available. The shoulder strap is routed over the shoulder adjacent to the window and attached to the lap belt in the general area of the occupant’s inboard hip. Adjust this fixed strap so that all controls are accessible while maintaining adequate restraint for the occupant. Shoulder harness with inertial reels are available for all four seats.

A check of the inertia reel mechanism is made by pulling sharply on the strap. The reel should lock in place under this test and prevent the strap from extending. For normal body movements. the strap will extend or retract as required.

7.27 BAGGAGE AREA

A 24 cubic foot baggage area, located behind the rear seat, is accessible from the cabin or loaded through a large 20 x 22 inch outside baggage door on the right side of the fuselage. Maximum capacity is 200 pounds. Tie-down straps are available and they should be used at all times.

NOTE

It is the pilot's responsibility to be sure when the baggage is loaded that the aircraft C.G. falls within the allowable C.G. range. (See Weight and Balance Section.)

7.29 STALL WARNING

An approaching stall is indicated by an audible alarm located behind the instrument panel. The indicator activates at between five and ten knots above stall speed.

7.31 FINISH

All exterior surfaces are primed with etching primer and finished with acrylic lacquer. To keep the finish attractive, economy size spray cans of touch-up paint are available from Piper Dealers.

An optional polyurethane finish is available.

7.33 PIPER EXTERNAL POWER*

An optional starting installation known as Piper External Power (PEP) is accessible through a receptacle located on the right side of the fuselage aft of the baggage door. An external battery can be connected to the socket, thus allowing the operator to crank the engine without having to gain access to the airplane's battery. Instructions on a placard located on the cover of the receptacle should be followed before using the external power. For instructions on the use of the PEP see; STARTING WITH EXTERNAL POWER SOURCE in Section 4 - Normal Operating Procedures.

7.35 EMERGENCY LOCATOR TRANSMITTER*

The Emergency Locator Transmitter (ELT) when installed, is located in the aft portion of the fuselage just below the stabilator leading edge and is accessible through a plate on the right side of the fuselage. This plate is attached with slotted-head nylon screws for ease of removal; these screws may be readily removed with a variety of common items such as a dime, a key, a knife blade, etc. If there are no tools available in an emergency the screw heads may be broken off by any means. The ELT is an emergency locator transmitter which meets the requirements of FAR 91.52.

A battery replacement date is marked on the transmitter to comply with FAA regulations, the battery must be replaced on or before this date. The battery must also be replaced if the transmitter has been used in an emergency situation or if the accumulated test time exceeds one hour, or if the unit has been inadvertently activated for an undetermined time period.

NOTE

If for any reason a test transmission is necessary, the test transmission should be conducted only in the first five minutes of any hour and limited to three audio sweeps. If the tests must be made at any other time, the tests should be coordinated with the nearest FAA tower or flight service station.

NARCO ELT 10 OPERATION

On the ELT unit itself is a three position switch placarded "ON," "OFF' and "ARM." The ARM position sets the ELT so that it will transmit after impact and will continue to transmit until its battery is drained. The ARM position is selected when the ELT is installed in the airplane and it should remain in that position.

To use the ELT as a portable unit in an emergency, remove the cover and unlatch the unit from its mounting base. The antenna cable is disconnected by a left quarter-turn of the knurled nut and a pull. A sharp tug on the two small wires will break them loose. Deploy the self-contained antenna by pulling the plastic tab marked "PULL FULLY TO EXTEND ANTENNA." Move the switch to ON to activate the transmitter.

In the event the transmitter is activated by an impact, it can only be turned off by moving the switch on the ELT unit to OFF. Normal operation can then be restored by pressing the small clear plastic reset button located on the top of the front face of the ELT and then moving the switch to ARM.

A pilot's remote switch located on the left side panel is provided to allow the transmitter to be turned on from inside the cabin. The pilot's remote switch is placarded "ON" and "ARMED." The switch is normally in the ARMED position. Moving the switch to ON will activate the transmitter. Moving the switch back to the ARMED position will turn off the transmitter only if the impact switch has not been activated.

*Optional equipment.

The ELT should be checked to make certain the unit has not been activated during the ground check. Check by selecting 121.50 MHz on an operating receiver. If there is an oscillating chirping sound, the ELT may have been activated and should be turned off immediately. This requires removal of the access cover and moving the switch to OFF, then press the reset button and return the switch to ARM. Recheck with the receiver to ascertain the transmitter is silent.

CCC CIR 11-2 OPERATION

On the unit itself is a three position selector switch placarded "OFF," "ARM" and "ON." The ARM position is provided to set the unit to the automatic position so that it will transmit only after impact and will continue to transmit until the battery is drained to depletion or until the switch is manually moved to the OFF position. The ARM position is selected when the transmitter is installed at the factory and the switch should remain in that position whenever the unit is installed in the airplane. The ON position is provided so the unit can be used as a portable transmitter or in the event the automatic feature was not triggered by impact or to periodically test the function of the transmitter.

Select the OFF position when changing the battery, when rearming the unit if it has been activated for any reason, or to discontinue transmission.

NOTE

If the switch has been placed in the ON position for any reason, the OFF position has to be selected before selecting ARM. If ARM is selected directly from the ON position, the unit will continue to transmit in the ARM position.

A pilot's remote switch, located on the left side panel, is provided to allow the transmitter to be controlled from inside the cabin. The pilot's remote switch is placarded "ON," "AUTO/ARM" and "OFF/RESET." The switch is normally left in the AUTO/ARM position. To turn the transmitter off, move the switch momentarily to the OFF/RESET position. The aircraft master switch must be ON to turn the transmitter OFF. To actuate the transmitter for tests or other reasons, move the switch upward to the ON position and leave it in that position as long as transmission is desired.

The unit is equipped with a portable antenna to allow the locator to be removed from the aircraft in case of an emergency and used as a portable signal transmitter.

The locator should be checked during the ground check to make certain the unit has not been accidentally activated. Check by tuning a radio receiver to 121.50 MHz. If there is an oscillating sound, the locator may have been activated and should be turned off immediately. Reset to the ARM position and check again to insure against outside interference.

7.37 AIR CONDITIONING*

The air conditioning system is a recirculating air system. The major items include: evaporator, condenser, compressor, blower, switches and temperature controls.

The evaporator is located behind the left rear side of the baggage compartment. This cools the air that is used for air conditioning.

The condenser is mounted on a retractable scoop located on the bottom of the fuselage and to the rear of the baggage compartment area. The scoop extends when the air conditioner is "ON" and retracts to a flush position when the system is "OFF."

The compressor is mounted on the forward right underside of the engine. It has an electric clutch which automatically engages or disengages the compressor to the belt drive system of the compressor.

An electrical blower is mounted on the aft side of the rear cabin panel. Air from the baggage area is drawn through the evaporator by the blower and distributed through an overhead duct to individual outlets located adjacent to each occupant.

The switches and temperature control are located on the lower right side of the instrument panel in the climate control center panel. The temperature control regulates the desired temperature of the cabin. Turn the control clockwise for increased cooling, counterclockwise for decreased cooling.

Located inboard of the temperature control is the fan speed switch and the air conditioning "ON-OFF" switch. The fan can be operated independently of the air conditioning. However, it must be on for air conditioner operation. Turning either switch off will disengage the compressor clutch and retract the condenser door. Cooling air should be felt within one minute after the air conditioner is turned on.

NOTE

If the system is not operating in 5 minutes, turn the system "OFF" until the fault is corrected.

The "FAN" switch allows operation of the fan with the air conditioner turned "OFF" to aid cabin air circulation if desired. A "LOW," "MED" or "HIGH" flow of air can be selected to the air conditioner outlets located in the overhead duct. The outlets can be adjusted or turned off by each occupant to regulate individual cooling effect.

The "DOOR OPEN" indicator light is located to the left of the radio stack in front of the pilot. The light illuminates whenever the condenser door is open and remains on until the door is closed.

A circuit breaker located on the circuit breaker panel protects the air conditioning electrical system.

Whenever the throttle is in the full throttle position, it actuates a micro switch which disengages the compressor and retracts the scoop. This is done to obtain maximum power and maximum rate of climb. The fan continues to operate and the air will remain cool for approximately one minute. When the throttle is retarded approximately 1/4 inch, the clutch will engage and the scoop will extend, again supplying cool, dry air.

*Optional equipment

7.39 CARBURETOR ICE DETECTION SYSTEM*

A carburetor ice detection system is available as optional equipment.

The system consists of a control box mounted on instrument panel, a probe sensor mounted in the carburetor and a red warning light to indicate the presence of ice in the carburetor. If ice is present apply full carburetor heat. Refer to 3.28. Carburetor Icing, in the emergency procedures. To adjust the system for critical ice detection, first turn on the airplanes master switch and then turn on the ice detection unit. Turn the sensitivity knob fully counter clockwise causing the carb ice light to come on. Now rotate the sensitivity knob back (clockwise) until the ice light just goes out. This establishes the critical setting.

WARNING

This instrument is approved as optional equipment only and Flight Operations should not be predicated on its use.

*Optional equipment

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AIRPLANE HANDLING, SERVICING AND MAINTENANCE

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SECTION 8

AIRPLANE HANDLING, SERVICING AND MAINTENANCE

8.1 GENERAL

This section provides general guidelines relating to the handling, servicing and maintenance of the Cherokee Warrior II. For complete maintenance instructions, refer to the PA-28 Service Manual.

WARNING

Inspection, maintenance and parts requirements for all non-PIPER approved STC installations are not included in this handbook. When a non-PIPER approved STC installation is incorporated on the airplane, those portions of the airplane affected by the installation must be inspected in accordance with the inspection program published by the owner of the STC. Since non-PIPER approved STC installations may change systems interface, operating characteristics and component loads or stresses on adjacent structures, PIPER provided inspection criteria may not be valid for airplanes with non-PIPER approved STC installations.

WARNING

Modifications must be approved in writing by PIPER prior to installation. Any and all other installations, whatsoever, of any kind will void this warranty in it's entirety.

WARNING

Use only genuine PIPER parts or PIPER approved parts obtained from PIPER approved sources, in connection with the maintenance and repair of PIPER airplanes.

Genuine PIPER parts are produced and inspected under rigorous procedures to insure airworthiness and suitability for use in PIPER airplane applications. Parts purchased from sources other than PIPER, even though identical in appearance, may not have had the required tests and inspections performed, may be different in fabrication techniques and materials, and may be dangerous when installed in an airplane.

Additionally, reworked or salvaged parts or those parts obtained from non-PIPER approved sources, may have service histories which are unknown or cannot be authenticated, may have been subjected to unacceptable stresses or temperatures or may have other hidden damage not discernible through routine visual or nondestructive testing. This may render the part, component or structural assembly, even though originally manufactured by PIPER, unsuitable and unsafe for airplane use.

PIPER expressly disclaims any responsibility for malfunctions, failures, damage or injury caused by use of non-PIPER approved parts.

8.1 GENERAL (continued)

Every owner should stay in close contact with an authorized Piper Service Center or Piper's Customer Services Department to obtain the latest information pertaining to their airplane, and to avail themselves of Piper's support systems.

Piper takes a continuing interest in having owners get the most efficient use from their airplane and keeping it in the best mechanical condition. Consequently, Piper, from time to time, issues service releases including Service Bulletins, Service Letters, Service Spares Letters, and others relating to the airplane.

Service Bulletins are of special importance and Piper considers compliance mandatory. These are sent directly to the latest FAA-registered owners in the United States (U.S.) and Piper Service Centers worldwide. Depending on the nature of the release, material and labor allowances may apply. This information is provided to all authorized Piper Service Centers.

Service Letters deal with product improvements and servicing techniques pertaining to the airplane. They are sent to Piper Service Centers and, if necessary, to the latest FAA-registered owners in the U.S. Owners should give careful attention to Service Letter information.

Service Spares Letters offer improved parts, kits, and optional equipment which were not available originally, and which may be of interest to the owner.

Piper offers a subscription service for Service Bulletins, Service Letters, and Service Spares Letters. This service is available to interested persons such as owners, pilots, and mechanics at a nominal fee, and may be obtained through an authorized Piper Service Center or Piper's Customer Services Department.

Maintenance manuals, parts catalogs, and revisions to both, are available from Piper Service Centers or Piper's Customer Services Department.

Any correspondence regarding the airplane should include the airplane model and serial number to insure proper response.

8.3 AIRPLANE INSPECTION PERIODS

WARNING

All inspection intervals, replacement time limits, overhaul time limits, the method of inspection, life limits, cycle limits, etc., recommended by PIPER are solely based on the use of new, remanufactured or overhauled PIPER approved parts. If parts are designed, manufactured, remanufactured, overhauled and/or approved by entities other than PIPER, then the data in PIPER'S maintenance/service manuals and parts catalogs are no longer applicable and the purchaser is warned not to rely on such data for non-PIPER parts. All inspection intervals, replacement time limits, overhaul time limits, the method of inspection, life limits, cycle limits, etc., for such non-PIPER parts must be obtained from the manufacturer and/or seller of such non-PIPER parts.

Piper has developed inspection items and required inspection intervals for the PA-28 (see the latest revision of the PA-28 Service and Inspection Manuals). The PA-28 Inspection Manual contains appropriate forms, and all inspection procedures should be complied with by a properly trained, knowledgeable, and qualified mechanic at a Piper Authorized Service Center or a reputable repair shop. Piper cannot accept responsibility for the continued airworthiness of any aircraft not maintained to these standards, and/or not brought into compliance with applicable Service Bulletins issued by Piper, instructions issued by the engine, propeller, or accessory manufacturers, or Airworthiness Directives issued by the FAA.

A Progressive Inspection, approved by the Federal Aviation Administration (FAA), is also available to the owner. This involves routine and detailed inspections to allow maximum utilization of the airplane. Maintenance inspection costs are reduced, and the maximum standard of continued airworthiness is maintained.

In addition, but in conjunction with the above, the FAA requires periodic inspections on all aircraft to keep the Airworthiness Certificate in effect. The owner is responsible for assuring compliance with these inspection requirements and for maintaining proper documentation in logbooks and/or maintenance records.

A spectrographic analysis of the engine oil is available from several sources. This inspection, if performed properly, provides a good check of the internal condition of the engine. To be accurate, induction air filters must be cleaned or changed regularly, and oil samples must be taken and sent in at regular intervals.

8.5 PREVENTIVE MAINTENANCE

The holder of a pilot certificate issued under Federal Aviation Regulations (FAR) Part 61 may perform certain preventive maintenance as defined in the FARs. This maintenance may be performed only on an aircraft which the pilot owns and operates, and which is not used in air carrier or air taxi/commercial operations service.

All other aircraft maintenance must be accomplished by a person or facility appropriately certificated by the Federal Aviation Administration (FAA) to perform that work.

Anytime maintenance is accomplished, an entry must be made in the appropriate aircraft maintenance records. The entry shall include:

(a) The date the work was accomplished.
(b) Description of the work.
(c) Number of hours on the aircraft.
(d) The certificate number of pilot performing the work.
(e) Signature of the individual doing the work.

8.7 AIRPLANE ALTERATIONS

If the owner desires to have his aircraft modified, he must obtain FAA approval for the alteration. Major alterations accomplished in accordance with Advisory Circular 43.13-2, when performed by an A & P mechanic, may be approved by the local FAA office. Major alterations to the basic airframe or systems not covered by AC 43.13-2 require a Supplemental Type Certificate.

The owner or pilot is required to ascertain that the following Aircraft Papers are in order and in the aircraft.

(a) To be displayed in the aircraft at all times:
  (1) Aircraft Airworthiness Certificate Form FAA-8100-2.
  (2) Aircraft Registration Certificate Form FAA-8050-3.
  (3) Aircraft Radio Station License if transmitters are installed.

(b) To be carried in the aircraft at all times:
  (1) Pilot's Operating Handbook.
  (2) Weight and Balance data plus a copy of the latest Repair and Alteration Form FAA-337, if applicable.
  (3) Aircraft equipment list.

Although the aircraft and engine logbooks are not required to be in the aircraft, they should be made available upon request. Logbooks should be complete and up to date. Good records will reduce maintenance cost by giving the mechanic information about what has or has not been accomplished.

8.9 GROUND HANDLING

(a) Towing

The airplane may be moved on the ground by the use of the nose wheel steering bar that is stowed below the forward ledge of the baggage compartment or by power equipment that will not damage or excessively strain the nose gear steering assembly. Towing lugs are incorporated as part of the nose gear fork.

CAUTION

When towing with power equipment, do not turn the nose gear beyond its steering radius in either direction, as this will result in damage to the nose gear and steering mechanism.

CAUTION

Do not tow the airplane when the controls are secured.

In the event towing lines are necessary, ropes should be attached to both main gear struts as high up on the tubes as possible. Lines should be long enough to clear the nose and/or tail by not less than fifteen feet, and a qualified person should ride in the pilot's seat to maintain control by use of the brakes.

(b) Taxiing

Before attempting to taxi the airplane, ground personnel should be instructed and approved by a qualified person authorized by the owner. Engine starting and shut-down procedures as well as taxi techniques should be covered. When it is ascertained that the propeller back blast and taxi areas are clear, power should be applied to start the taxi roll, and the following checks should be performed:

(1) Taxi a few feet forward and apply the brakes to determine their effectiveness.
(2) While taxiing, make slight turns to ascertain the effectiveness of the steering.
(3) Observe wing clearance when taxiing near buildings or other stationary objects. If possible, station an observer outside the airplane.
(4) When taxiing over uneven ground, avoid holes and ruts.
(5) Do not operate the engine at high RPM when running up or taxiing over ground containing loose stones, gravel, or any loose material that may cause damage to the propeller blades.

(c) Parking

When parking the airplane, be sure that it is sufficiently protected from adverse weather conditions and that it presents no danger to other aircraft. When parking the airplane for any length of time or overnight, it is suggested that it be moored securely.

(1) To park the airplane, head it into the wind if possible.
(2) Set the parking brake by pulling back on the brake lever and depressing the knob on the handle. To release the parking brake, pull back on the handle until the catch disengages; then allow the handle to swing forward.

CAUTION

Care should be taken when setting brakes that are overheated or during cold weather when accumulated moisture may freeze a brake.

(3) Aileron and stabilator controls should be secured with the front seat belt and chocks used to properly block the wheels.

(d) Mooring

The airplane should be moored for immovability, security and protection. The following procedures should be used for the proper mooring of the airplane:

(1) Head the airplane into the wind if possible.
(2) Retract the flaps.
(3) Immobilize the ailerons and stabilator by looping the seat belt through the control wheel and pulling it snug.
(4) Block the wheels.
(5) Secure tie-down ropes to the wing tie-down rings and to the tail skid at approximately 45 degree angles to the ground. When using rope of non-synthetic material, leave sufficient slack to avoid damage to the airplane should the ropes contract.

CAUTION

Use bowline knots, square knots or locked slip knots. Do not use plain slip knots.

NOTE

Additional preparations for high winds include using tie-down ropes from the landing gear forks and securing the rudder.

(6) Install a pitot head cover if available. Be sure to remove the pitot head cover before flight.
(7) Cabin and baggage doors should be locked when the airplane is unattended.

8.11 ENGINE AIR FILTER

The wet-type polyurethane foam air filter must be inspected at least once every fifty hours. Under extremely adverse operating conditions, it may be necessary to inspect the filter more frequently. The filter is disposable and inexpensive and a spare should be kept on hand for a rapid replacement.

(a) Removal Of Engine Air Filter

The filter is located in the lower right front of the engine compartment and may be removed by the following procedure:

(1) Open the right side of the engine cowling.
(2) Loosen each of the four quarter-turn fasteners securing the air filter cover.
(3) Separate the cover and remove the filter.
(4) Inspect the filter. If it is excessively dirty or shows any damage, replace it immediately.

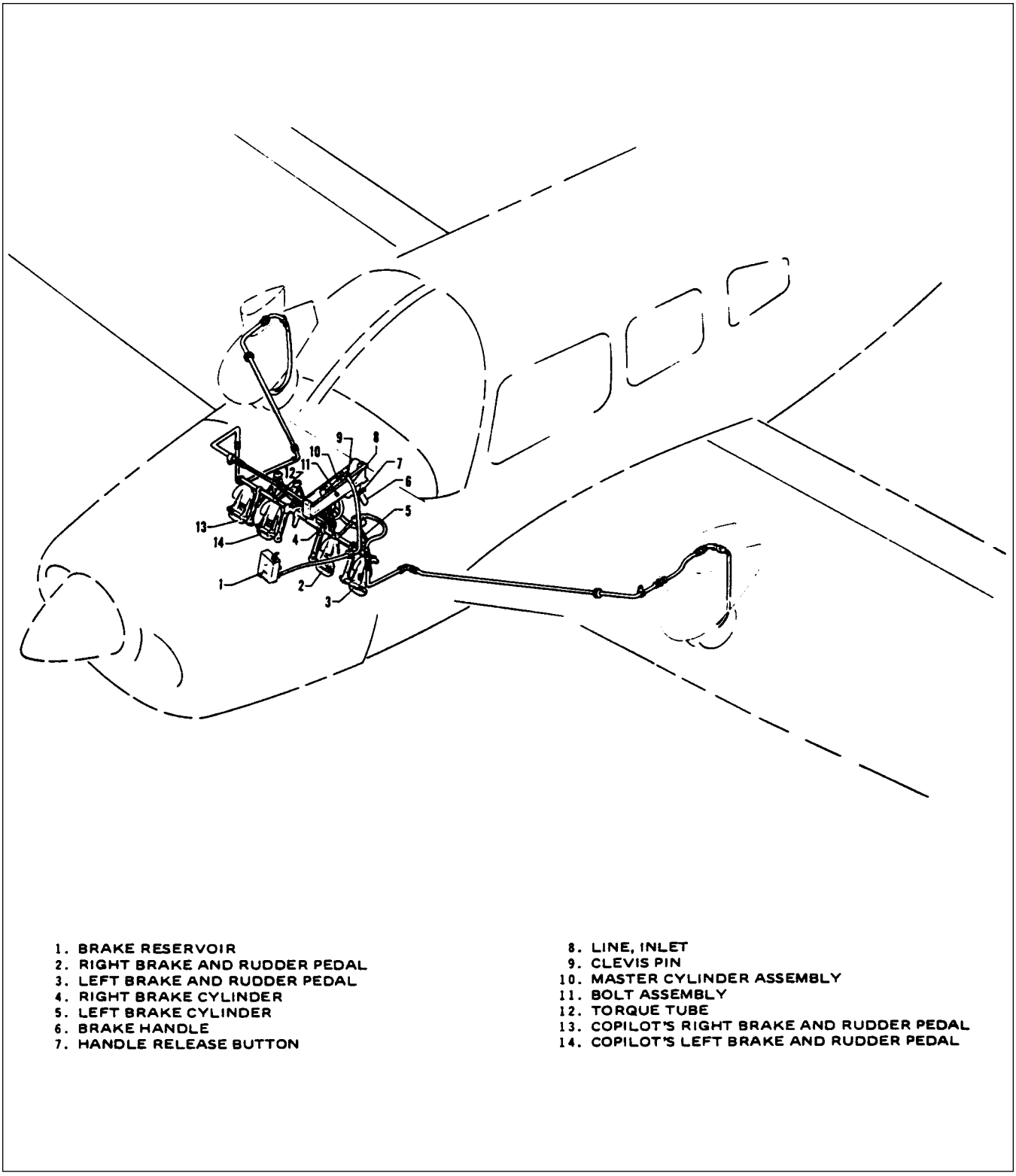
(b) Installation Of Engine Air Filter

When replacing the filter, install the filter in the reverse order of removal.

8.13 BRAKE SERVICE

The brake system is filled with MIL-H-5606 (petroleum base) hydraulic brake fluid. The fluid level should be checked periodically or at every 50 hour inspection and replenished when necessary. The brake reservoir is located on the fire wall in the engine compartment. If the entire system must be refilled, fill with fluid under pressure from the brake end of the system. This will eliminate air from the system.

No adjustment of the brake clearances is necessary. If after extended service brake blocks become excessively worn, they should be replaced with new segments.



BRAKE SYSTEM

Figure 8-1

8.15 LANDING GEAR SERVICE

The main landing gears use 6.00 x 6 wheels and the nose gear carries a 5.00 x 5 wheel. All three tires are four-ply rating, type III tires with tubes. (Refer to paragraph 8.23).

Wheels are removed by taking off the hub cap, cotter pin, axle nut, and the two bolts holding the brake segment in place. Mark tire and wheel for reinstallation; then dismount by deflating the tire, removing the three through-bolts from the wheel and separating the wheel halves.

Landing gear oleos should be serviced according to the instructions on the units. The main oleos should be extended under normal static load until 4.50 ± .25 inches of oleo piston tube is exposed, and the nose gear should show 3.25 ± .25 inches. Should the strut exposure be below that required, it should be determined whether air or oil is required by first raising the airplane on jacks; Depress the valve core to allow air to escape from the strut housing chamber. Remove the filler plug and slowly raise the strut to full compression. If the strut has sufficient fluid, it will be visible up to the bottom of the filler plug hole and will then require only proper inflation.

Should fluid be below the bottom of the filler plug hole, oil should be added. Replace the plug with valve core removed; attach a clear plastic hose to the valve stem of the filler plug and submerge the other end in a container of hydraulic fluid. Fully compress and extend the strut several times, thus drawing fluid from the container and expelling air from the strut chamber. To allow fluid to enter the bottom chamber of the main gear strut housing, the torque link assembly must be disconnected to let the strut be extended a minimum of 10 inches (the nose gear torque links need not be disconnected). Do not allow the strut to extend more than 12 inches. When air bubbles cease to flow through the hose, compress the strut fully and again check fluid level. Reinstall the valve core and filler plug, and the main gear torque links, if disconnected.

With fluid in the strut housing at the correct level, attach a strut pump to the air valve and with the airplane on the ground, inflate the oleo strut to the correct height.

In jacking the aircraft for landing gear or other service, two hydraulic jacks and a tail stand should be used. At least 250 pounds of ballast should be placed on the base of the tail stand before the airplane is jacked up. The hydraulic jacks should be placed under the jack points on the bottom of the wing and the airplane jacked up until the tail skid is at the right height to attach the tail stand. After the tail stand is attached and the ballast added, jacking may be continued until the airplane is at the height desired.

The steering arms from the rudder pedals to the nose wheel are adjusted at the nose wheel by turning the threaded rod end bearings in or out. Adjustment is normally accomplished at the forward end of the rods and should be done in such a way that the nose wheel is in line with the fore and aft axis of the plane when the rudder pedals and rudder are centered. Alignment of the nose wheel can be checked by pushing the airplane back and forth with the rudder centered to determine that the plane follows a perfectly straight line. The turning arc of the nose wheel is 30.0° ± 2° in either direction and is limited by stops on the bottom of the forging.

The rudder pedal arm stops should be carefully adjusted so that the pedal arms contact the stops just after the rudder hits its stops. This guarantees that the rudder will be allowed to move through its full travel.

8.17 PROPELLER SERVICE

The spinner and backing plate should be cleaned and inspected for cracks. Before each flight the propeller should be inspected for nicks, scratches, and corrosion. If found, they should be repaired as soon as possible by a rated mechanic, since a nick or scratch causes an area of increased stress which can lead to serious cracks or the loss of a propeller tip. The back face of the blades should be painted when necessary with flat black paint to retard glare. To prevent corrosion, the surface should be cleaned and waxed periodically.

8.19 OIL REQUIREMENTS

The oil capacity of engine is 8 quarts, and the minimum safe quantity is 2 quarts. It is recommended that engine oil be drained and renewed every 50 hours. The oil filter element should be changed every 50 hours of operation. The interval between oil and oil filter changes should not exceed a total of four (4) months. Under unfavorable dusty conditions, the oil and oil filter should be changed more frequently.

The following seasonal aviation oil grades and seasonal ambient temperature ranges are recommended.

Average Ambient Temperature	MIL-L-6082B SAE Grade	MIL-L-22851 Ashless Dispersant SAE Grades
All Temperatures	—	15W-50 or 20W-50
Above 80°F	60	60
Above 60°F	50	40 or 50
30°F to 90°F	40	40
0°F to 70°F	30	30, 40 or 20W-40
Below 10°F	20	30 or 20W-30

When operating temperatures overlap indicated ranges, use the lighter grade oil.

NOTE

Refer to the latest issue of Textron Lycoming Service Instruction 1014 (Lubricating Recommendations) for further information.

8.21 FUEL SYSTEM

(a) Servicing Fuel System

At every 50 hour inspection, the fuel screens in the strainer, in the electric fuel pumps, and at the carburetor inlet must be cleaned.

(b) Fuel Requirements (AVGAS ONLY)

The minimum aviation grade fuel for the PA-28-161 is 100. Since the use of lower grades can cause serious engine damage in a short period of time, the engine warranty is invalidated by the use of lower octanes.

Whenever 100 or 100LL grade fuel is not available, commercial grade 100/130 should be used. (See Fuel Grade Comparison Chart, Page 8-12.) Refer to the latest issue of Lycoming Service Instruction No. 1070 for additional information.

A summary of the current grades as well as the previous fuel designations is shown in the following chart:

FUEL GRADE COMPARISON CHART

Previous Commercial Fuel Grades (ASTM-D910)			Current Commercial Fuel Grades (ASTM-D910-75)			Current Military Fuel Grades (MIL-G-5572E) Amendment No. 3		
Grade	Color	Max. TEL ml/U.S. gal.	Grade	Color	Max. TEL ml/U.S. gal.	Grade	Color	Max. TEL ml/U.S. gal
80/87	red	0.5	80	red	0.5	80/87	red	0.5
91/98	blue	2.0	*100LL	blue	2.0	none	none	none
100/130	green	3.0	100	green	**3.0	100/130	green	**3.0
115/145	purple	4.6	none	none	none	115/145	purple	4.6

* - Grade 100LL fuel in some over seas countries is currently colored green and designated as 100L.

** - Commercial fuel grade 100 and grade 100/130 (both of which are colored green) having TEL content of up to 4 ml/U.S. gallon are approved for use in all engines certificated for use with grade 100/130 fuel.

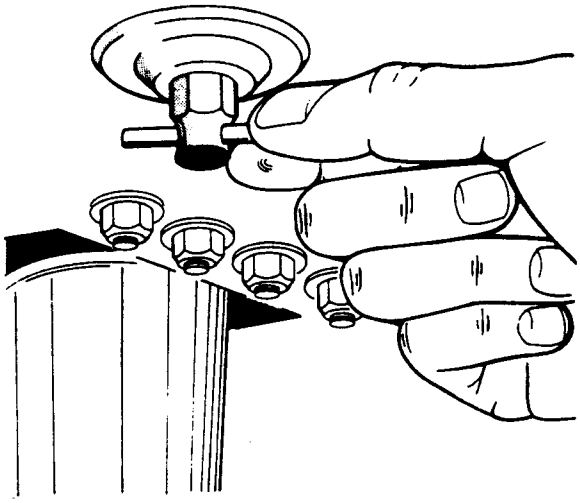
The operation of the aircraft is approved with an anti-icing additive in the fuel. When anti-icing additive is used it must meet the specification MIL-I-27686, must be uniformly blended with the fuel while refueling, must not exceed 0.15% by volume of the refueled quantity, and to ensure its effectiveness should be blended at not less than 0.10% by volume. One and one half liquid ozs. per ten gallon of fuel would fall within this range. A blender supplied by the additive manufacturer should be used. Except for the information contained in this section, the manufacturer's mixing or blending instructions should be carefully followed.

CAUTION

Assure that the additive is directed into the flowing fuel stream. The additive flow should start after and stop before the fuel flow. Do not permit the concentrated additive to come in contact with the aircraft painted surfaces or the interior surfaces of the tanks.

Some fuels have anti-icing additives pre-blended in the fuel at the refinery, so no further blending should be performed.

Fuel additive can not be used as a substitute for preflight draining of the fuel system drains.



FUEL DRAIN

Figure 8-3

(c) Filling Fuel Tanks

Observe all required precautions for handling gasoline. Fuel is stored in two twenty-five gallon (24 usable) tanks.

(d) Draining Fuel Strainer, Sumps and Lines

The fuel system sumps and strainer should be drained daily prior to the first flight and after refueling to avoid the accumulation of contaminants such as water or sediment. Each fuel tank is equipped with an individual quick drain located at the lower inboard rear corner of the tank. The fuel strainer is equipped with a quick drain located on the front lower corner of the firewall. Each of the fuel tank sumps should be drained first. Then the fuel strainer should be drained twice, once with the fuel selector valve on each tank. Each time fuel is drained, sufficient fuel should be allowed to flow to ensure removal of contaminants. This fuel should be collected in a suitable container, examined for contaminants. and then discarded.

CAUTION

When draining any amount of fuel, care should be taken to ensure that no fire hazard exists before starting the engine.

After draining, each quick drain should be checked to make sure it has closed completely and is not leaking.

(e) Draining Fuel System

The bulk of the fuel may be drained from the system by opening valve at the inboard end of each fuel tank. Push up on the arms of the drain valve and turn counterclockwise to hold the drain open. The remaining fuel in the system may be drained through the filter bowl. Any individual tank may be drained by closing the selector valve and then draining the desired tank.

8.23 TIRE INFLATION

For maximum service from the tires, keep them inflated to the proper pressures - 30 psi for the nose gear and 24 psi for the main gear. All wheels and tires are balanced before original installation, and the relationship of tire, tube and wheel should be maintained upon reinstallation. Unbalanced wheels can cause extreme vibration in the landing gear; therefore, in the installation of new components, it may be necessary to rebalance the wheels with the tires mounted. When checking tire pressure, examine the tires for wear, cuts, bruises, and slippage.

8.25 BATTERY SERVICE

Access to the 12-volt battery is obtained by raising the rear seat and removing the cover of the battery box. The plastic battery box has a drain tube which is normally closed off with a cap and which should be opened occasionally to drain off any accumulation of liquid.

The battery should be checked for proper fluid level. DO NOT fill the battery above the baffle plates. DO NOT fill the battery with acid - use only water. A hydrometer check will determine the percent of charge in the battery.

If the battery is not up to charge, recharge starting at a 4 amp rate and finishing with a 2 amp rate. Quick charges are not recommended.

8.27 CLEANING

(a) Cleaning Engine Compartment

Before cleaning the engine compartment, place a strip of tape on the magneto vents to prevent any solvent from entering these units.

(1) Place a large pan under the engine to catch waste.

(2) With the engine cowling removed, spray or brush the engine with solvent or a mixture of solvent and degreaser. In order to remove especially heavy dirt and grease deposits, it may be necessary to brush areas that were sprayed.

CAUTION

Do not spray solvent into the alternator, vacuum pump, starter, or air intakes.

(3) Allow the solvent to remain on the engine from five to ten minutes. Then rinse the engine clean with additional solvent and allow it to dry.

CAUTION

Do not operate the engine until excess solvent has evaporated or otherwise been removed.

(4) Remove the protective tape from the magnetos.
(5) Lubricate the controls, bearing surfaces, etc., in accordance with the Lubrication Chart.

(b) Cleaning Landing Gear

Before cleaning the landing gear, place a plastic cover or similar material over the wheel and brake assembly.

(1) Place a pan under the gear to catch waste.
(2) Spray or brush the gear area with solvent or a mixture of solvent and degreaser, as desired. Where heavy grease and dirt deposits have collected, it may be necessary to brush areas that were sprayed, in order to clean them.
(3) Allow the solvent to remain on the gear from five to ten minutes. Then rinse the gear with additional solvent and allow to dry.
(4) Remove the cover from the wheel and remove the catch pan.
(5) Lubricate the gear in accordance with the Lubrication Chart.

(c) Cleaning Exterior Surfaces

The airplane should be washed with a mild soap and water. Harsh abrasives or alkaline soaps or detergents could make scratches on painted or plastic surfaces or could cause corrosion of metal. Cover areas where cleaning solution could cause damage. To wash the airplane, use the following procedure:

(1) Flush away loose dirt with water.
(2) Apply cleaning solution with a soft cloth, a sponge or a soft bristle brush.
(3) To remove exhaust stains, allow the solution to remain on the surface longer.
(4) To remove stubborn oil and grease, use a cloth dampened with naphtha.
(5) Rinse all surfaces thoroughly.
(6) Any good automotive wax may be used to preserve painted surfaces. Soft cleaning cloths or a chamois should be used to prevent scratches when cleaning or polishing. A heavier coating of wax on the leading surfaces will reduce the abrasion problems in these areas.

(d) Cleaning Windshield and Windows

(1) Remove dirt, mud and other loose particles from exterior surfaces with clean water.
(2) Wash with mild soap and warm water or with aircraft plastic cleaner. Use a soft cloth or sponge in a straight back and forth motion. Do not rub harshly.
(3) Remove oil and grease with a cloth moistened with kerosene.

CAUTION

Do not use gasoline, alcohol, benzene, carbon tetrachoride, thinner, acetone, or window cleaning sprays.

(4) After cleaning plastic surfaces, apply a thin coat of hard polishing wax. Rub lightly with a soft cloth. Do not use a circular motion.
(5) A severe scratch or mar in plastic can be removed by rubbing out the scratch with jeweler's rouge. Smooth both sides and apply wax.

(e) Cleaning Headliner, Side Panels and Seats

(1) Clean headliner, side panels, and seats with a stiff bristle brush, and vacuum where necessary.
(2) Soiled upholstery, except leather, may be cleaned with a good upholstery cleaner suitable for the material. Carefully follow the manufacturer's instructions. Avoid soaking or harsh rubbing.

CAUTION

Solvent cleaners require adequate ventilation.

(3) Leather should be cleaned with saddle soap or a mild hand soap and water.

(f) Cleaning Carpets

To clean carpets, first remove loose dirt with a whisk broom or vacuum. For soiled spots and stubborn stains use a noninflammable dry cleaning fluid. Floor carpets may be removed and cleaned like any household carpet.

8.29 COLD WEATHER OPERATION

For cold weather operation a winterization plate is installed on the inlet opening of the oil cooler. This plate should be installed whenever the ambient temperature reaches 50°F or less. The plate should be removed and stored in the cockpit when the ambient temperature exceeds 50°F.

It is recommended that an optional Engine Breather Tube Winterization Kit be installed for cold weather operation. This kit is available through your Piper Dealer/Distributor.

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SUPPLEMENTS

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SECTION 9

SUPPLEMENTS

9.1 GENERAL

This section provides information in the form of Supplements which are necessary for efficient operation of the airplane when equipped with one or more of the various optional systems and equipment not provided with the standard airplane.

All of the Supplements provided by this section are "FAA Approved" and consecutively numbered as a permanent part of this Handbook. The information contained in each Supplement applies only when the related equipment is installed in the airplane.

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SUPPLEMENT 3

PIPER ELECTRIC PITCH TRIM

SECTION 1 - GENERAL

This supplement supplies information necessary for the operation of the airplane when the optional Piper Electric Pitch Trim is installed. The information contained within this supplement is to be used in conjunction with the complete handbook.

This supplement has been FAA Approved as a permanent part of this handbook and must remain in this handbook at all times when the optional Piper Electric Pitch Trim is installed.

SECTION 2 - LIMITATIONS

No changes of the basic limitations provided by Section 2 of this Pilot's Operating Handbook are necessary for this supplement.

SECTION 3

(a) In case of malfunction, disconnect electric pitch trim by advancing pitch trim switch on instrument panel to OFF position.
(b) In an emergency, electric pitch trim may be overpowered using manual pitch trim.
(c) In cruise configuration, malfunction results in 10° pitch change and 200 ft altitude variation.
(d) In approach configuration, a malfunction can result in a 5° pitch change and 50 ft altitude loss.

SECTION 4 - NORMAL PROCEDURES

The electric trim system may be turned ON or OFF by a switch located above the ignition switch. The pitch trim may be changed when the electric trim system is turned on either by moving the manual pitch trim control wheel or by operating the trim control switch on the pilot's control yoke.

SECTION 5 - PERFORMANCE

No changes of the basic performance provided by Section 5 of this Pilot's Operating Handbook are necessary for this supplement.

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SUPPLEMENT TO OWNER'S MANUAL, INSTALLATION AND PILOT'S GUIDE
FOR CIR-11 EMERGENCY LOCATOR TRANSMITTER

THE FOLLOWING INFORMATION REPLACES THE INFORMATION FOUND ON PAGES 5 THROUGH 8 OF THE CIR-11 OWNER'S MANUAL, PRODUCT SUPPORT PUBLICATION 950012, AND SHOULD BE AFFIXED TO THE MANUAL FOR FUTURE REFERENCE.

For maximum possible range, the CIR-11( ) should be located as high as possible above the surrounding terrain; for instance, in a tree or on a hill top. The antenna should be oriented vertically.

3. GENERAL INFORMATION

As in all radio transmissions you don't hear your broadcast going over the air, for the same reasons you will not hear your CIR-11( ) transmitting when the "G" switch is activated or when you set the ARM-ON-OFF switch to ON. If your aircraft receiver is functioning after a hard landing, the receiver may be tuned to 121.5 MHz and you will be able to hear your signal.

Under normal temperature conditions (16 to 27°C) the CIR-11( ) will continue to emit signals for over 48 hours, although not at full output power. For extended operation, intermittently turn the ELT off. This will extend the battery life.

The CIR-11( ) will operate in temperatures of -20°C to +70°C, however at the extremes some performance is lost.

When using the CIR-11( ) in temperatures below freezing, keep the transmitter inside your jacket with the antenna sticking outside. This prolongs the operating life.

If the aircraft is located in a valley, it is suggested to remove the CIR-11( ) and position it (antenna pointing skyward) in a clearing on top of a knoll, hill or other high place to achieve greater transmission range.

The CIR-11( ) is water resistant and is not affected if submerged for up to 20 hours. The shock of an emergency landing (ground or water) will activate the unit. Should a water landing be made, attempt to retrieve the CIR-11( ) as soon as possible, especially if the plane is submerged.

Add the CIR-11( ) to your preflight and post-flight check list:
Preflight - Assure the CIR-11( ) (Control Head and/or remote switch) is set to ARM.
Post-Flight - Assure the CIR-11( ) (Control Head and/or remote switch) is set to ARM.
Verify the ELT has not been inadvertently activated as follows: Tune a VHF receiver to 121.5 MHz. If an ELT tone is heard, it may be you or someone else. Check by waiting a moment or two to hear if it stops. If it doesn't stop, press in on the ARM-ON-OFF or remote RESET switch. If the tone ceases simultaneously with the pressing of the switch and starts again when the switch is released, the "G" switch could be bad. Set the switch to OFF and return the CIR-11( ) to the factory for service. The impact "G" switch can only be turned off by pressing the ARM-ON-OFF or remote RESET switch or setting the switch to OFF.

4. TYPE DESIGNATION

The CIR-11( ) is approved under FAA TSO-C91 for the following type categories:

Automatic Fixed Type . . . . . ELT (AP)

Automatic Portable Type . . . . . ELT (AP)

Personnel Type . . . . . ELT (P)

DESIGN FEATURES

More than 48 hours operational life

Antenna designed for optimum performance on any aircraft

Alkaline power supply

Fixed and portable antennas provided

Remote deactivation capability

Lightweight

Quick-release mounting bracket

Water resistant

6. SPECIFICATIONS

The CIR-11( ) meets all the requirements of:

FAA TSO-C91

FCC Part 87 (including 25 KHz occupied bandwidth)

Operating Frequencies:

121.5 and 243.0 MHz $\pm$ .005%

Operating Life:

48 Hours Minimum

Modulation Characteristics:

100% Amplitude Modulation (A9) Emission

Duty Cycle Greater Than 33-1/3%

2 to 4 sweeps per second

Peak Effective Radiated Power (PERP):

At least 125 mw on each frequency

Range:

Up to 200 miles

Operating Temperature:

-20°C to +55°C

Activation:

Automatic ($5.0 \pm \frac{2}{0}$ G) and/or manual

Power Source:

Collins Model BP60-13 Alkaline Battery Pack

Dimensions:

8-5/32 x 2-3/4 x 3

Weight:

3.5 Pounds

7. EQUIPMENT

Item	Description
1.	CIR-11( ) Emergency Locator Transmitter Assy.
	Complete with:
	Battery Pack BP60-13
	Portable Antenna ELT10-249
	Mounting Bracket

PS ENGINEERING® INCORPORATED
Sound Quality. Sound Engineering.
9800 Martel Road
Lenoir City, TN 37772
www.ps-engineering.com

PMA8000G

Audio Selector Panel with *IntelliAudio*®

Stereo IntelliVox® Intercom System with flightmate® and Bluetooth® Connectivity

Flying Never Sounded So Good!®

Pilot's Guide and Operation Manual

202-890-0912 Bluetooth® Rev. 9 January 2023

Applies to Serial Number PG0160`and above

Covered under one or more of the following Patent No. 4,941,187, 5,751,817, 5,903,227, 6,160,496, 6,493,450,

FAA-Approved: TSO C50c, C35d
EASA-Approved ETSO C50c, 2C35d

Index

Operation

PMA8000G controls

Power Switch (1) (EMG-Fail Safe Operation)

The power switch controls all audio selector panel functions, intercom and marker beacon receiver.

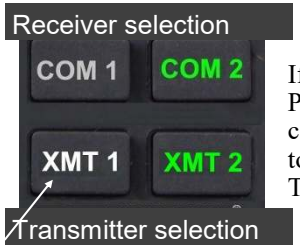
When the unit is turned off, either by pressing the volume control, *or* if the breaker is pulled removing power, the PMA8000G is in Fail-Safe mode. In this mode, the pilot's headset is connected to COM 1 for transmit and receive, and connected to unswitched input #1 for priority audio alerts. The fail safe audio will *only* be heard in the left ear of a stereo headset.

Communications Transmit (XMT) Selection (2)

To select COM 1 or COM 2 for transmit, press the XMT 1 or XMT 2 button on the bottom row. The bottom and top button indicators light, showing that you will transmit **and** receive on the selected radio.

The PMA8000G has an automatic selector system. Audio from the selected transceiver is automatically heard in the headsets and speaker (if selected). This guarantees that the pilot will *always* hear the audio from the transceiver selected for transmit.

In normal (not Split) modes, the PMA8000G gives priority to the pilot's radio Push-To-Talk (PTT). If the copilot is transmitting, and the pilot presses his PTT, the pilot's microphone will be heard over the selected com transmitter.

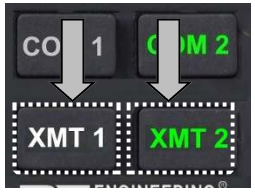


If the pilot is connected to a cell phone, the pilot PTT will switch the pilot microphone to the selected com transceiver, and allow aircraft communications to continue.

The copilot will also be able to transmit on the selected radio with his PTT as well.

Split Mode

To activate the **split** mode, push both the XMT 1 and XMT 2 (bottom) buttons at the same time. All four indicators will come on. In the **split** mode, the pilot is on COM 1, while the copilot is on COM 2. In **split** mode, the intercom between pilot and copilot is off, as is the green ICS indicator. Press the MUTE button to restore intercom to the CREW mode if desired.



Select either of the com XMT buttons to exit the **split** mode. It is not possible to have the pilot on COM 2 and copilot on COM 1 in **split** mode.

NOTE: Due to the nature of VHF communications signals, and the size constraints in general aviation aircraft, it is probable that there will be some bleed -over in the Split mode, particularly on adjacent frequencies. PS Engineering makes no warranty about the suitability of Split Mode in all aircraft conditions.

Communications Receive (RCV) Selection (3)

To listen to the other radio, press the upper COM 1 or COM 2 button. When a com is selected for receive, it will stay selected until manually deselected, even if you select, and then deselect its transmitter.

When switching from COM 1 to COM 2 while Com 2 was not previously selected, COM 1 audio will be switched off. In essence, switching the mic selector will not override prior selection of COM receiver audio.

Unless the audio panel is in "split" mode, the PMA8000G gives priority to the pilot's radio Push-To-Talk (PTT).

When in a Telephone mode, the pilot is connected to the cell phone, but still hears the COM radios selected. The radio PTT will switch the mic to the selected com, and allow continued aircraft communications to continue.

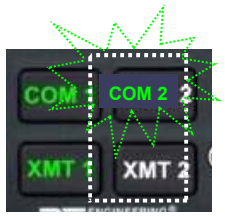
IntelliAudio® HRTF (11)

When IntelliAudio®, Head Related Transfer Function (**HRTF**) processing is active, COM 1 audio will be presented in the 10 o'clock position, and COM 2 will be presented from the 2 o'clock position. IntelliAudio® is toggled on and off by pushing the HRTF button. This function is **only** present for the *pilot and copilot* positions and requires properly wired stereo headsets.

Monitor Mode

The Monitor Mode allows the audio from a primary radio (selected for transmit) to mute the audio from a secondary radio (selected for listen only) when the primary radio is receiving a signal. This allows the crew to copy weather but still hear ATC clearly. Press the COM button on secondary radio until

"Monitor ON" is heard in the headset to activate the Monitor mode. The secondary COM button will blink in Monitor mode. Hold again to deactivate. NOTE: activating the monitor mode will turn on COM receive audio if not already selected.



Nav Audio Selector (4)

These buttons select the switched navigation receivers. The DME and ADF inputs (if present) are also shared with **AUX**. In SPLIT mode, only the pilot will hear selected navigation audio.

Cockpit Speaker (5)

This switch will place all selected radio audio on the cockpit speaker when this switch is selected. In "SPLIT" mode, the speaker carries the same audio as the pilot.

Depending on installation, important audio alerts such as radar altimeter or autopilot disconnect will come over the speaker even if it is not selected, while other unswitched inputs, will only be present if the **SPR** button is selected. Consult your professional avionics installer for these important configuration details.

Intercom Operation

IntelliVox® VOX-Squelch

IntelliVox® is PS Engineering's proprietary intercom squelch control. Through the use of digital processors, each microphone is monitored, and opens instantly when human speech is detected. This results in seamless conversations aboard the airplane for crew and passengers, without syllable clipping or fatigue-inducing noise.

No adjustment of the *IntelliVox*® squelch control is necessary. There is no field adjustment. The system is designed to block continuous tones, therefore people humming or whistling in monotone may be blocked after a few moments.

For consistent performance, any headset microphone **must** be placed within **¼-inch** of your lips, preferably against them. It is important to have the microphone element parallel to your mouth, and not twisted inside the cover.

Note: For optimum microphone performance, we recommend use of a Microphone Muff Kit from Oregon Aero (1-800-888-6910, www.oregonaero.com). This will not only optimize VOX performance, but will improve the overall clarity of all your communications.

You should also keep the microphone out of a direct wind. Moving your head through an air stream may cause the *IntelliVox*® to open momentarily. This is normal. The *IntelliVox*® is designed to work with normal aircraft

cabin noise levels (70 dB and above). Therefore, it may not always recognize speech and clip syllables in a quiet area, such as in the hangar, or without the engine running. This is also normal.

Intercom Volume Control (7)

The small volume control knob adjusts the loudness of the intercom for the pilot and copilot. It has no effect on selected radio levels, music input levels or passengers' volume level.

The larger, outer volume control knob controls intercom volume for the passengers. It has no effect on radio or music levels.

Mono Headsets in Stereo Installation

The pilot and copilot positions work with mono headsets. However, the IntelliAudio® feature will not function.

If a monaural headset is plugged in to a PMA8000G Stereo installation, one channel will be shorted and all passengers will lose one channel unless they switch to the “MONO” mode on the headset.

Intercom Modes (8)

The intercom has three modes. This button cycles through the intercom modes, from top to bottom and then back up. The description of the intercom mode function is valid only when the unit is not in the "**Split**" mode.

ISO: The pilot is isolated from the intercom and is connected only to the aircraft radio system. He will hear the aircraft radio reception (and sidetone during radio transmissions). Copilot will hear passengers’ intercom and entertainment, while passengers will hear copilot intercom and entertainment. Neither will hear aircraft radio receptions or pilot transmissions. The pilot can hear music if desired by pressing the MUSIC button.

ALL: All parties will hear the aircraft radio and intercom. Crew and passengers will hear selected entertainment. During any radio or intercom communications, the music volume automatically decreases. The music volume increases gradually back to the original level after communications have been completed.

CREW: Pilot and copilot are connected on one intercom channel and have exclusive access to the aircraft radios. They may also listen to Entertainment 1. Passengers can continue to communicate with themselves without interrupting the Crew and may listen to entertainment as configured.

In **SPLIT** mode, the intercom, and the indicator are off for the crew. Pressing the **Mute** button enables the activates the intercom and indicator in the **CRW** mode.

Remote ICS Mode switch

If a remote SWAP switch is installed, the intercom mode can be changed by a long press (more than 1 second) of the swap button.

Alternate Intercom Function

This function controls the distribution of aircraft radio within the intercom, as well as passenger intercom muting, **when in the "ALL" intercom mode**. In the "*standard intercom function*" mode, aircraft radios are distributed to all, when the intercom is in the ALL mode. In CREW mode, only the pilot and copilot positions will hear aircraft radios.

When "Alternate Intercom" function is toggled (hold ICS button for more than three seconds in ALL mode) a voice announces "*Alternate Intercom Function*," and the passengers will NOT hear aircraft radios, even in the **All** intercom mode. They will be able to converse with the crew. However, when the aircraft radio becomes active, the intercom audio from the passengers is muted, allowing the crew to focus on the radio. The passengers will still be able to talk to each other. Pressing holding the ICS button for three seconds again switches the intercom back to "*Standard Intercom Function*."

Alternate Intercom mode is reset at power up.

Marker Beacon Operation (9)
(Applies to 05-890-0912 Only)

The Marker Beacon Receiver uses visual and audio indicators to alert you when the aircraft passes over an ILS Marker Beacon transmitter.

The Blue, **O**uter Marker lamp has an associated 400-Hertz 'dash' tone. The lamp and tone will be keyed at a rate of two tones/flashes per second when the aircraft is in the range of the Outer Marker.

The Amber, **M**iddle Marker lamp is coupled with a 1300 Hertz tone, keyed alternately with short 'dot' and long 'dash' bursts at 95 combinations per minute.

The White, **I**nner marker lamp has a 3000 Hertz 'dot' tone, and will be keyed at a rate of six times per second.

Marker Beacon Receiver audio can be heard by a short press of the "**MKR**" push-button switch. The Audio (AUD) indicator will illuminate.

The **MKR** button is also used to set the receiver sensitivity and to test the indicator lamps mute the marker audio.

Short press - Controls Marker audio on/off. The top indicator (AUD) lights green when audio selected. Selection will be remembered during power cycles.

Medium press - This mutes the marker beacon audio. It will stay muted until the pilot reaches the next ILS marker indicator and then it will un-mute.

(Continued on page 11)

Quick reference PMA8000G Operation

This pull-out section covers advanced operation of the PMA8000G.

flightmate®

The flightmate® is an audio storage system with three features, radio playback, stored audio playback for a checklist or reminders.

flightmate® messages

The flightmate "scratchpad" recording offers 2 ½ minutes of recording time for checklists or other messages. To start recording, press and hold the flightmate "PLAY" button. A chime tone indicates the start of the recording. This recording can be played by pressing the PLAY button until the recording starts to playback.

Radio Playback

The radio recording portion is a continuous loop recorder, (last message received will be the first heard), the recorder has 45 seconds of recording time, or up to eight messages.

There are no buttons to press to start recording. The system automatically begins to record the instant the radio selected for transmit becomes active. Only the pilot and copilot will hear the playback audio.

To play back the last recorded message, you quickly press PLAY. You must wait for the message to stop playing before accessing the prior message. To cancel the playback, quickly press the PLAY button again. The next time the button is pressed, the next earlier message will be heard. If the radio becomes active while a message is playing, the message playback will stop. The new audio will not be stored.

Messages are lost when a different radio is selected for transmit.

The playback will stop whenever there is more incoming selected com audio, and the message can be replayed from the beginning. **Note:** an external playback button may also be installed in a convenient location.

Bluetooth Connection

1. Turn on the PMA8000G.
2. From your Bluetooth enabled device, search for other devices, and select the PMA8000G.
3. The PMA8000G should connect automatically. The Bluetooth symbol will light up.

The PMA8000G can be connected to one Bluetooth device at a time. It is able to pair with up to eight individual devices. When that number is exceeded, one device will be automatically un-paired to allow the new device. The device eliminated will be selected at random by the Bluetooth module. *Hint, if your old phone is not recognized by the PMA8000G, you may simply need to re-pair.*

Reset Bluetooth devices

If the Bluetooth stops responding to paired devices, it may be because of corrupted device data. Simply reset the Bluetooth system as follows:

1. Power off audio panel
2. Put the pilot's headset on
3. Hold down NAV1 and NAV2 buttons at the same time
4. Turn ON the audio panel, and continue to hold down the two buttons for at least 5 seconds or there is a "ding" sound in headset, if you do not hear this chime start over.
5. Audio panel re-initiation is complete
6. Remove PMA8000G device from your smart phone
7. Search for Bluetooth devices from your smart phone
8. Select PMA8000G
9. If password is required try 0000
10. Once paired stream music and place phone call

Split Mode

The Split mode puts the pilot on COM 1, while the copilot can use COM 2 independently. To enter the split mode, press both the XMT 1 and XMT 2 buttons at the same time. To exit, press the desired COM 1 or COM 2 XMT button.

When you activate the Split mode, the intercom is inhibited to avoid confusion with multiple conversations. The ICS indicator goes off when Split mode is activated. To reactivate the intercom, press the Mute button, which places the intercom in CREW mode.

Note: Split Mode does not turn off selected navaid audio to pilot. However, the copilot will only hear the selected com receiver and unswitched inputs.

Alternate Intercom mode

The Alternate Intercom mode allows the passengers and crew to converse, in ALL intercom mode, without distracting the crew from radio duties. The passenger microphones removed from the crew when the radio is active, and the passengers never hear aircraft radios. Press and hold the **ICS** button until the audio annunciation starts (about 1 second) to activate the function, and you will hear "Alternate Intercom Function." Hold the button again to exit, and you will hear "Standard Intercom Function."

Music Muting

Music source #1 (Bluetooth and Music 1 input) has three muting modes, which are announced in the headset as they are activated. These are: Radio Mute (aircraft radio mutes music), Mute on (both radio and intercom mutes music), and Mute off (nothing interrupts music). Press the Mute button to cycle through the modes in sequence.

Music #2 has either Mute On or our Karaoke™ mode, which the AUX button can be used to control this mute mode. A dedicated switch may also be installed.

Music in pilot isolate mode

The pilot can elect to listen to Music 1. While already in the ISO mode, press and hold the **ICS** button until the indicator blinks. The ICS indicator will then blink every few seconds to indicate this mode is active. Music muting will follow the modes listed above. This mode resets with power cycle.

Music Distribution

The Music (DISTR**) button** has three states; **Standard Music Distribution, Music 1 Pilot** and **Music 1 All Headsets**.

Music 1 (the Bluetooth device *or* the Music 1 input on the rear connector) can be distributed to all headsets depending on intercom mode.

In *Standard Music Distribution,* Music 1 is provided to the crew, and Music 2 is independently provided to the passengers.

In *Music 1, Pilot*, the pilot hears Music 1, while the copilot and passengers hear Music 2

(Continued from page 7)

The test mode is activated, all the marker lamps activate and light the T/M indicator next to the MKR button.

Long press – Controls marker low & high sense change. When HIGH is selected, the HIGH indicator will be lit in green. Selection will be remembered during power cycles. Because LOW is default mode, the indicator will be normally off.

Bluetooth® Telephone Mode

There is a ringtone in the headset when a call is received. The **PLAY** button flashes green, and you can answer by pressing the button. It remains green while a call is in progress.

The telephone mode serves as a full duplex interface for telephone systems such as cellular phones with Bluetooth® connectivity. When interfaced with an approved airborne telecommunications system, the PMA8000G can serve as an audio control and distribution center.

In **ALL** intercom mode, crew and passengers will be heard on the phone when they speak. All will hear selected audio. Com audio is automatically heard in the headsets.

In **CREW** mode, the pilot and copilot are connected to the telephone. The pilot and copilot will have transmit capability on the other selected transceiver Com 1 or 2, simply by using their respective PTT switch.

In **ISO** intercom mode, when the PMA8000G is in the **TEL** mode, the pilot position is in the "Phone Booth." Only the pilot will hear the telephone, and only he will be heard. He will also have access to Com 1 or 2, and will transmit on that radio using the PTT. All selected audio is provided. You can end the call by pressing **PLAY** again.

Music and Music Muting

The PMA8000G has two independent music inputs at the rear connector.

The PMA8000G also has the ability to receive streaming music from a Bluetooth-enabled device.

Music 1 will be heard by the pilot and copilot positions. Music 1 can also be distributed to the passengers using the MUSIC DISTR control. The Bluetooth streamed music in the PMA8000G is also handled as Music 1.

NOTE:

All music devices should be turned off for take off, landing, or any critical phase of flight. FAA Regulation 14 CFR 91.21 restricts the use of portable electronic devices.

§91.21 “(a) Except as provided in paragraph (b) of this section, no person may operate, nor may any operator or pilot in command of an aircraft allow the operation of, any portable electronic device on any of the following U.S.-registered civil aircraft. . .

“(b)(5) Any other portable electronic device that the operator of the aircraft has determined will not cause interference with the navigation or communication system of the aircraft on which it is to be used.”

You can refer to Advisory Circular 91.21-1C for more information, at www.faa.gov.

The front panel "Mute" button has **three** modes, and controls the Mute function for Music 1.

The SoftMute™ circuit will mute the music whenever there is conversation on the radio, the intercom, or both, depending on the “Mute” mode selected. When that conversation stops, the music returns to the previous level comfortably, over a second or so.

The mute mode functions are controlled through sequential pushes of the MUTE button, and include annunciations of the mode selected. The PMA8000G always defaults to *Mute On* when turned on.

Mute On- music **will** mute with *either* intercom *or* radio - MUTE button is lit green. Voice annunciation is "mute on."

Mute Off - “Karaoke” mode - music will not mute except during outgoing radio transmissions.- MUTE indicator is OFF. Annunciation is "mute off."

Radio Mute - *Radio communications* will mute music, but intercom will **not** mute music - MUTE indicator is OFF. Annunciation is "radio mute."

The passenger’s intercom also has a SoftMute™ circuit. If the passengers hear the radio, or talk on the intercom, the music will mute. If the audio panel is in CREW mode, then the radio reception will not affect the passenger music.

Passengers also have a **Mute Off** or “Karaoke” mode. If the passengers are listening to the music 1 input, their Mute Off is controlled by the front panel “Mute” button. If the passengers are listening to the music 2 input, their Karaoke Mode is activated by an external switch installed either in the panel, or connected to the AUX button logic output pin on the PMA8000G.

Music Distribution (DISTR)

The Music Distribution button allows you to either distribute the music 1 input to all intercom stations, all of the time, or have other distribution rules apply to your music inputs.

When “*Music one all headsets*” is selected (press and hold **MUSIC**), music 1 (from the rear connector input, Pins J2 23 & 24 only) or Bluetooth will be distributed to all headsets and is

independent of the intercom mode switch. Therefore, even in the CREW mode, the passengers will hear Music 1, even though they will not hear the intercom or radios.

This mode allows you to use a single in-flight entertainment source aboard, and to send it everywhere, even in crew mode. The music muting will be normal, and follow the selected mode of the crew or passengers.

Press and hold **MUSIC** again, this will select "*Music one pilot*". This will allow the Music 1 and/or Bluetooth to be distributed to the pilot's headsets while the copilot and passengers listen to Music #2.

Press and hold **MUSIC** again, and you will hear "*Standard Music Distribution.*" In this mode, Music 2 becomes active, and will always be presented to the passengers on the intercom. Music 1 is only available to the pilot and copilot. The intercom mode switch doesn't have any effect on the music distribution.

When the music distribution is "standard," Music 1 will always go to the pilot and copilot positions, and is never heard by the passengers. Music 2 is always heard by the passengers, and never heard by the pilot and copilot.

This mode is useful if your passengers have a different interest in entertainment or are watching a DVD, but do not want to be excluded from the intercom conversations.

The pilot and copilot will always hear **Music 1** through the unit rear connector. This is present in ALL and CREW intercom modes, and available to the copilot position in ISO mode.

Music 2 is provided to the passenger positions regardless of intercom mode, when the audio panel is in *Standard Music Distribution*. The pilot and copilot can never hear **Music 2**, under any condition.

If the passengers always want to hear the source in **Music 1**, input through the rear connector, regardless of the intercom mode (ISO/ALL/CRW), select "*Music 1 all headsets.*"

Pilot music on/off

A quick press if the MUSIC button will add music to the pilot headset, and the music button turns green. In this way, music is available to the pilot in any intercom mode, including Isolate.

flightmate®

flightmate® is an audio storage system with three features, radio playback, stored audio playback for a checklist or reminders.

flightmate® messages

The flightmate "scratchpad" recording offers 2 ½ minutes of recording time for checklists or other messages.

To start recording, press and hold the flightmate "PLAY" button.

A chime tone indicates the start of the recording. See Quick reference to store messages.

This recording can be played by pressing the PLAY button until the recording starts, and release.

The Radio Recording System is a continuous loop recorder, (last message received will be the first heard), the recorder has 45 seconds of recording time, or up to eight messages. There are no buttons to press to start recording. The system automatically begins to record the instant the radio selected for transmit becomes active. Only the pilot and copilot will hear the playback audio.

To play back the last recorded message, you press PLAY button quickly. You must wait for the message to stop playing before accessing the prior message. To cancel the playback, press play button again. The next time the button is pressed for one second, the next earlier message will be heard. If the radio becomes active while a message is playing, the message playback will stop. The new audio will not be stored. Press play to restart the message you were playing.

Messages are lost when a different radio is selected for transmit.

The playback will stop whenever there is more incoming selected com audio, and the message can be replayed from the beginning. **Note:** an external playback button may also be installed in a convenient location.

Backlighting

The lighting in buttons, indicators, and backlit text is controlled by the panel light dimmer. If a dimmer is not used, the input wiring should be connected for daytime viewing.

Public Address Function

To enter PA mode, press **SPR** button for more than one second. The **SPR** LED will blink while the audio panel is in PA mode. The copilot can use the selected com while the pilot will be heard over the speaker. To exit push **SPR** again. This mode is also reset when power is cycled.

Warranty & Service

In order for the factory warranty to be valid, the installations in a certified aircraft must be accomplished by an FAA-(or other ICAO agency) certified avionics shop and authorized PS Engineering dealer. If the unit is being installed by a non-certified individual in an experimental aircraft, a factory-made intercom harness must be used for the warranty to be valid.

PS Engineering, Inc. warrants this product to be free from defect in material and workmanship for a period of two (2) years from the date of retail sale by authorized PS Engineering dealer. During the first **twelve (12) months** of the three-year warranty period, PS Engineering, Inc., at its option, will send a replacement unit at our expense if the unit should be determined to be defective after consultation with a factory technician. For the remaining **twelve (12) months** of the three-year warranty period, PS Engineering will send a no-cost replacement unit at customer shipping expense.

All transportation charges for returning the defective units are the responsibility of the purchaser. All domestic transportation charges for returning the exchange or repaired unit to the purchaser will be borne by PS Engineering, Inc. The risk of loss or damage to the product is borne by the party making the shipment, unless the purchaser requests a specific method of shipment. In this case, the purchaser assumes the risk of loss.

This warranty is not transferable. Any implied warranties expire at the expiration date of this warranty. PS Engineering SHALL NOT BE LIABLE FOR INCIDENTAL OR CONSEQUENTIAL DAMAGES. This warranty does not cover a defect that has resulted from improper handling, storage or preservation, or unreasonable use or maintenance as determined by us. This warranty is void if there is any attempt to dissemble this product without factory authorization. This warranty gives you specific legal rights, and you may also have other rights, which may vary from state to state. Some states do not allow the exclusion of limitation of incidental or consequential damages, so the above limitation or exclusions may not apply to you.

All items repaired or replaced under this warranty are warranted for the remainder of the original warranty period. PS Engineering, Inc. reserves the rights to make modifications or improvements to the product without obligation to perform like modifications or improvements to previously manufactured products.

Factory Service

The units are covered by a two-year limited warranty. See warranty information. Call PS Engineering, Inc. at (865) 988-9800 before you return any unit. This will allow the service technician to provide any other suggestions for identifying the problem and recommend possible solutions.

After discussing the problem with the technician and you obtain a Return Authorization Number, ship product to:

PS Engineering, Inc. Attn: Service Department
9800 Martel Rd.

Lenoir City, TN 37772
(865) 988-9800 FAX (865) 988-6619 Email: contact@ps-engineering.com

Units that arrive without an RMA number, or telephone number for a responsible contact, will be returned un-repaired. PS Engineering is not responsible for items sent via US Mail.

Scan for warranty registration

Garmin International, Inc.
1200 E. 151st Street
Olathe, Kansas 66062 U.S.A.

FAA APPROVED

AIRPLANE FLIGHT MANUAL SUPPLEMENT
or
SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
for the
Garmin GTX 33X and GTX 3X5 Transponders with ADS-B
as installed in

_____
Make and Model Airplane

Registration Number: _____ Serial Number: _____

This document serves as an FAA Approved Airplane Flight Manual Supplement or Supplemental Airplane Flight Manual when the GTX 33X or GTX 3X5 with ADS-B is installed in accordance with Supplemental Type Certificate SA01714WI. This document must be incorporated into the FAA Approved Airplane Flight Manual or provided as an FAA Approved Supplemental Airplane Flight Manual.

The information contained herein supplements the FAA approved Airplane Flight Manual. For limitations, procedures, loading and performance information not contained in this document, refer to the FAA approved Airplane Flight Manual, markings, or placards.

FAA Approved By: JR Brownell

JR Brownell
ODA STC Unit Administrator
Garmin International, Inc.
ODA-240087-CE

Date: 6/16/2021

LOG OF REVISIONS				
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Revision Number	Date	Number	Description	FAA Approved
1	05/01/2013	All	Complete Supplement	*Robert Murray* Robert Murray ODA STC Unit Administrator Garmin International, Inc. ODA-240087-CE Date: *05/01/2013*
2	03/08/2016	All	New supplement format with GTX 3X5 added.	*Michael Warren* Michael Warren ODA STC Unit Administrator Garmin International, Inc. ODA-240087-CE Date: *03/08/2016*
3	12/07/2017	All	Updated SW versions and removed section 3.2.3. Updated section 2.2 Corrected PED FAR reference and additional minor corrections.	*Erik Frisk* Erik Frisk ODA STC Unit Administrator Garmin International, Inc. ODA-240087-CE Date: *12/21/2017*
4	09/09/2019	4, 6, 7, 9, 11, 13, 14, 18	Added GTX diversity units, updated SW versions, expanded allowed remote control panels, and incorporated other minor changes	*JR Brownell* JR Brownell ODA STC Unit Administrator Garmin International, Inc. ODA-240087-CE Date: *09/09/2019*
5	06/16/2021	10, 11, 14, 18	Updated GTX 3X5 Main software to version 2.60, added GI 275 as a control display and GPS 175/GNC 355 as a GPS source	See cover page 1

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Section 1. GENERAL

1.1 GTX 33X

The Garmin GTX 33X family consists of the GTX 330 ES and GTX 33 ES (Non-Diversity Mode S Transponders) and the GTX 330D ES and GTX 33D ES (Diversity Mode S Transponders). The ES option of any of the transponders provides ADS-B extended squitter functionality.

All Garmin GTX 33X transponders are a radio transmitter/receiver that operates on radar frequencies, receiving ground radar or TCAS interrogations at 1030 MHz and transmitting a coded response of pulses to ground-based radar on a frequency of 1090 MHz. Each unit is equipped with IDENT capability to initiate the SPI (special position identification) pulse for 18 seconds and will reply to ATCRBS Mode A, Mode C and Mode S All-Call interrogation. Interfaces to the GTX 33X are shown in the following block diagrams.

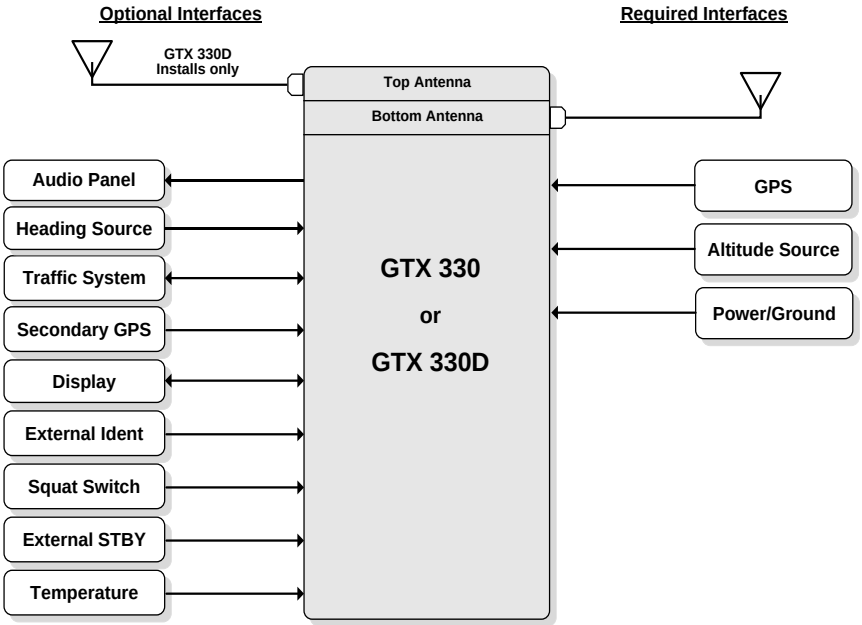

Figure 1 – GTX 330 or GTX 330D Interface Summary

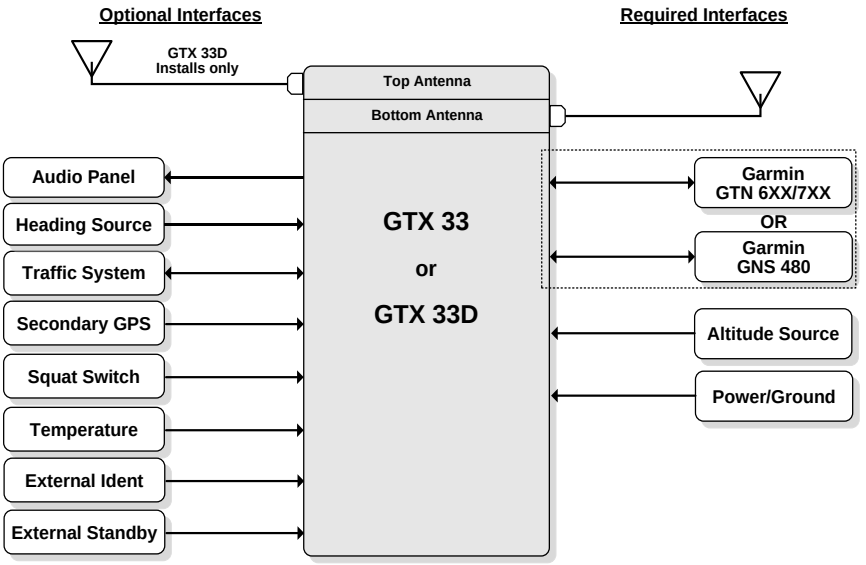

Figure 2 – GTX 33 or GTX 33D Interface Summary

The GTX 33X performs the following functions:

- Transmission of ADS-B out data on 1090 extended squitter (1090 MHz)
  - Integration of data from internal and external sources to transmit the following data per 14 CFR 91.227:
    - GPS Position, Altitude, and Position Integrity
    - Ground Track and/or Heading, Ground Speed, and Velocity Integrity
    - Air Ground Status
    - Flight ID, Call Sign, ICAO Registration Number
    - Capability and Status Information
    - Transponder Squawk Codes between 0000-7777.
    - Emergency Status
    - IDENT - initiates SPI (special position identification) pulse for 18 seconds
  - Pressure Altitude Broadcast Inhibit
- Reception of TIS-A traffic data from a ground station
- Provides TIS-A traffic alerting to the pilot via interfaced display and audio output

1.2 GTX 3X5

The Garmin GTX 3X5 family consists of the GTX 335, 335D, 335R, 335DR, 345, 345D, 345R, and 345DR transponders. The functional differences between each of these transponders are described in Table 1. Transponder models with a "D" designation are diversity capable and support both a top fuselage and bottom fuselage antenna.

Function	GTX 335/ 335D	GTX 335 w/GPS	GTX 335R/ 335DR	GTX 335R w/GPS	GTX 345/ 345D	GTX 345 w/GPS	GTX 345R/ 345DR	GTX 345R w/GPS
Panel mount	x	x			x	x		
Remote mount			x	x			x	x
Mode S	x	x	x	x	x	x	x	x
ADS-B (out)	x	x	x	x	x	x	x	x
ADS-B Traffic					x	x	x	x
FIS-B					x	x	x	x
Internal GPS		x		x		x		x
Bluetooth					x	x	x	x
Optional Garmin Altitude Encoder	x	x	x	x	x	x	x	x

Table 1 – GTX 3X5 Unit Configurations

Interfaces to the GTX 3X5 are shown in Figure 3.

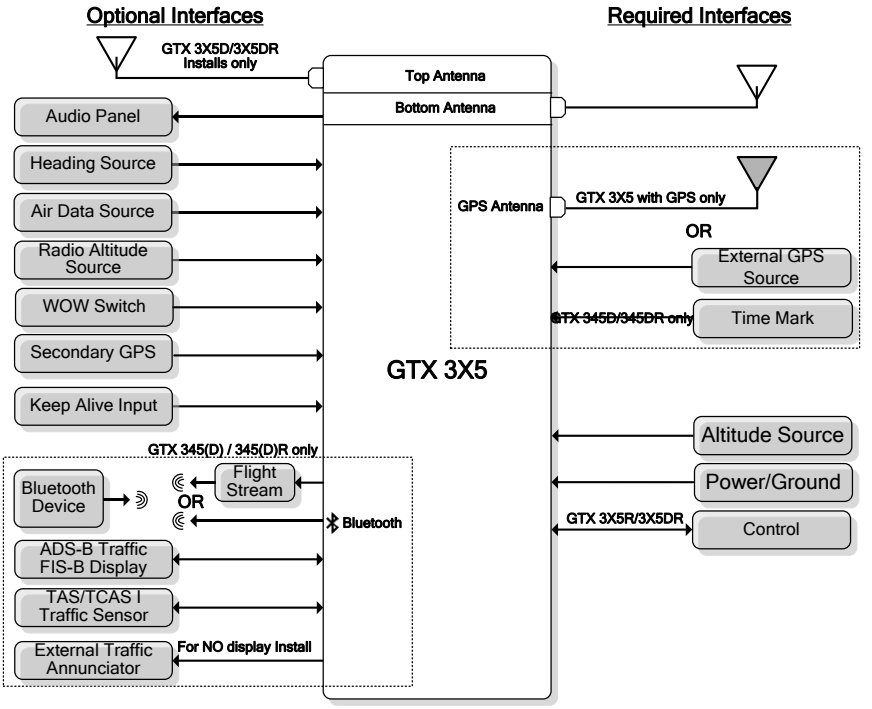

Figure 3 – GTX 3X5 Interface Summary

The GTX 3X5 performs the following functions:

- Transmission of ADS-B out data on 1090 extended squitter (1090 MHz)
  - Integration of data from internal and external sources to transmit the following data per 14 CFR 91.227:
    - GPS Position, Altitude, and Position Integrity
    - Ground Track and/or Heading, Ground Speed, and Velocity Integrity
    - Air Ground Status
    - Flight ID, Call Sign, ICAO Registration Number
    - Capability and Status Information
    - Transponder Squawk Codes between 0000-7777.
    - Emergency Status
    - IDENT - initiates SPI (special position identification) pulse for 18 seconds
  - Pressure Altitude Broadcast Inhibit

The GTX 335 performs the following additional functions:

- Reception of TIS-A traffic data from a ground station
- Provide TIS-A traffic alerting to the pilot via interfaced display and audio output.

The GTX 345 performs the following additional functions:

- Reception of ADS-B In data on 1090 MHz
  - ADS-B (Data directly from another transmitting aircraft)
  - ADS-R (Rebroadcast of ADS-B data from a ground station)
- Reception of ADS-B In data on UAT (978 MHz)
  - ADS-B (Data directly from another transmitting aircraft)
  - ADS-R (Rebroadcast of ADS-B data from a ground station)
  - TIS-B (Broadcast of secondary surveillance radar) (SSR) derived traffic information from a ground station.
  - FIS-B (Broadcast of aviation data from a ground station)
- Provide ADS-B traffic information and alerting to the pilot via an interfaced display
  - Correlation and consolidation of traffic data from multiple traffic sources
  - Aural and visual traffic alerting
- Provide FIS-B data to the pilot via an interfaced display
  - Graphical and textual weather products
    - NEXRAD
    - PIREPs
    - AIRMET/SIGMETs
    - METARs
    - TAFs
    - Winds Aloft
  - Aviation Data
    - TFRs
    - NOTAMs

1.3 Capabilities

The Garmin GTX 33X and GTX 3X5 as installed in this aircraft have been shown to meet the equipment requirements of 14 CFR § 91.227 when operating in accordance with sections 2.1 and 2.2 of this supplement.

1.4 Installation Configuration

This aircraft is equipped with a GTX 33X and/or GTX 3X5 with the following interfaces/ features:

Equipment Installed:

Transponder #1	Transponder #2 (if installed)
☐ GTX 330	☐ GTX 330
☐ GTX 330D	☐ GTX 330D
☐ GTX 33	☐ GTX 33
☐ GTX 33D	☐ GTX 33D
☐ GTX 335	☐ GTX 335
☐ GTX 335D	☐ GTX 335D
☐ GTX 335R	☐ GTX 335R
☐ GTX 335DR	☐ GTX 335DR
☐ GTX 345	☐ GTX 345
☐ GTX 345D	☐ GTX 345D
☐ GTX 345R	☐ GTX 345R
☐ GTX 345DR	☐ GTX 345DR

Interfaced GPS/SBAS Position Source(s):

GPS #1	GPS #2 (if installed)
☐ Internal	☐ Internal
☐ GTN 6XX/7XX Series	☐ GTN 6XX/7XX Series
☐ GNS 400W/500W Series	☐ GNS 400W/500W Series
☐ GNS 480	☐ GNS 480
☐ GIA 63W	☐ GIA 63W
☐ GDL 88 (GTX 330 only)	☐ GDL 88 (GTX 330 only)
☐ GPS 175/GNC 355	☐ GPS 175/GNC 355

Interfaced Pressure Altitude Source:

Pressure Altitude Source #1	Pressure Altitude Source #2 (if installed)
☐_____	☐_____
☐ Garmin Altitude Encoder	☐ Garmin Altitude Encoder

Interfaced Remote Control Display (Required for remotely mounted GTX variants):

Transponder #1 Remote Control Display	Transponder #2 Remote Control Display (if installed)
☐ GTN 6XX/7XX	☐ GTN 6XX/7XX
☐ GNS 480	☐ GNS 480
☐ G950/1000 Display	☐ G950/1000 Display
☐ GI 275	☐ GI 275
☐ Gables 7534 Controller	☐ Gables 7534 Controller
☐ Gables 7614 Controller	☐ Gables 7614 Controller
☐ CTL-92 Controller	☐ CTL-92 Controller
☐ CTL-92E Controller	☐ CTL-92E Controller

Interfaced Active Traffic System:

☐ None

☐ TCAD

☐ TAS/TCAS

NOTE

If the system includes all of the following components:

- GTX 345R or GTX 345DR,
- G950/1000 Display, and
- TCAD or TAS/TCAS

Then the aircraft is no longer equipped with a TSO compliant active TCAD, TAS or TCAS system. Any operational requirement to be equipped with such system is no longer met.

1.5 Definitions

The following terminology is used within this document:

ADS-B: Automatic Dependent Surveillance-Broadcast

AFM: Airplane Flight Manual

AFMS: Airplane Flight Manual Supplement

ATCRBS: Air Traffic Control Radar Beacon System

CFR: Code of Federal Regulations

ES: Extended Squitter

GNSS: Global Navigation Satellite System

GNS: Garmin Navigation System

GPS: Global Positioning System

GTX: Garmin Transponder

GTN: Garmin Touchscreen Navigator

ICAO: International Civil Aviation Organization

LRU: Line Replaceable Unit

PABI: Pressure Altitude Broadcast Inhibit

POH: Pilot Operating Handbook

SBAS: Satellite-Based Augmentation System

SW: Software

TCAS: Traffic Collision Avoidance System

TIS: Traffic Information Service

TX: Transmit

Section 2. LIMITATIONS

2.1 Minimum Equipment

The GTX 33X and GTX 3X5 must have the following system interfaces fully functional in order to be compliant with the requirements for 14 CFR 91.227 ADS-B Out operations:

Interfaced Equipment	Number Installed	Number Required
Uncorrected Pressure Altitude Source	1	1
GPS SBAS Position Source	1 or more	1
Remote Control Display (for remotely mounted transponders)	1 or more	1

Table 2 – Required Equipment

2.2 ADS-B Out

The GTX 33X and GTX 3X5 only comply with 14 CFR 91.227 for ADS-B Out when all required functions are operational. When the system is not operational, ADS-B Out transmit failure messages will be present on the remote control display interface, or the GTX 330 or GTX 3X5 panel display. If a Gables 7534 controller or Collins CTL-92/92E controller is being used the ADS-B equipment failure condition will be annunciated on the Gables or Collins display "Transponder Fail" while the ADS-B Out Position failure will be annunciated by the remotely installed "ADS-B POSN FAIL" Annunciator.

2.3 TIS Traffic Display with User Navigation Angle

Display of TIS traffic from a GTX 33/330 or GTX 335 is not permitted with an interfacing display configured for a navigation angle of "user".

2.4 Applicable System Software

This AFMS/AFM is applicable to the software versions shown in Table 3.

The Main GTX software version is displayed on the splash screen during start up for the GTX 330 and GTX 3X5 panel mounted units, and the External LRU or System page on the interfaced remote control display for remotely mounted GTX transponders.

Software Item	**Software Version** *(or later FAA Approved versions for this STC)*
GTX 33X Main SW Version	8.04
GTX 3X5 Main SW Version	2.60

Table 3 - Software Versions

2.5 Pressure Altitude Broadcast Inhibit (PABI)

Pressure Altitude Broadcast Inhibit shall only be enabled when requested by Air Traffic Control while operating within airspace requiring an ADS-B Out compliant transmitter. PABI is enabled by selecting the GTX to ON mode.

2.6 Datalinked Weather Display (GTX 345 Only)

Do not use datalink weather information for maneuvering in, near, or around areas of hazardous weather. Information provided by datalink weather products may not accurately depict current weather conditions.

Do not use the indicated datalink weather product age to determine the age of the weather information shown by the datalink weather product. Due to time delays inherent in gathering and processing weather data for datalink transmission, the weather information shown by the datalink weather product may be significantly older than the indicated weather product age.

Do not rely solely upon datalink services to provide Temporary Flight Restriction (TFR) or Notice to Airmen (NOTAM) information.

2.7 Portable Electronic Devices

This STC does not relieve the operator from complying with the requirements of 91.21 or any other operational regulation regarding portable electronic devices.

Section 3. EMERGENCY PROCEDURES

3.1 Emergency Procedures

No Change.

3.2 Abnormal Procedures

3.2.1 LOSS OF AIRCRAFT ELECTRICAL POWER GENERATION

XPDR Circuit Breaker..... **PULL**

Transponder and ADS-B Out functions will no longer be available.

NOTE

This guidance is supplementary to any guidance provided in the POH or AFM for the installed aircraft for loss of power generation.

3.2.2 LOSS OF GPS/SBAS POSITION DATA

When the GPS/SBAS receiver is inoperative or GPS position information is not available or invalid, the GTX will no longer be transmitting ADS-B Out data.

For GTX 330 installations:

NO ADSB annunciator illuminated:

Interfaced GPS position sources..... **VERIFY VALID POSITION**

For GTX 3X5 installations:

NO 1090ES TX annunciator illuminated:

Interfaced GPS position sources..... **VERIFY VALID POSITION**

For GTX 33 and GTX 3X5R installations:

Reference Display Device documentation for applicable annunciation:

Interfaced GPS position sources..... **VERIFY VALID POSITION**

Section 4. NORMAL PROCEDURES

The procedures described below are specific only to the panel mounted GTX 330 or GTX 3X5 transponders. Cockpit Reference Guides and Pilot Guides for interfaced remote control displays will provide additional operating information specific to the displays or other traffic systems.

ADS-B Out functionality resides within the GTX transponders thereby providing a single point of entry for Mode 3/A code, Flight ID, IDENT functionality and activating or deactivating emergency status for both transponder and ADS-B Out functions. Details on performing these procedures are located in the GTX 330/330D Pilot's Guide and GTX 3X5 Series Transponder Pilot's Guide.

4.1 Unit Power On

For GTX 330 installations:

GTX Mode..... **VERIFY ALT**
NO ADSB..... **CONSIDERED**

For GTX 3X5 installations:

GTX Mode..... **VERIFY ALT**
NO 1090ES TX ..... **CONSIDERED**

NOTE

The NO ADS-B or NO 1090ES TX Annunciation (or associated display annunciations) may illuminate as the unit powers on and begins to receive input from external systems, to include the SBAS position source.

4.2 Before Takeoff

For GTX 330 installations:

ADS-B TX.....**VERIFY ON**
NO ADSB ..... **EXTINGUISHED**

For GTX 3X5 installations:

1090ES TX CTL .....**VERIFY ON**
NO 1090ES TX ..... **EXTINGUISHED**

NOTE

The ADS-B TX or 1090ES TX CTL must be turned on and the NO ADS-B or NO 1090ES TX Annunciation (or associated display annunciations) must be **EXTINGUISHED** for the system to meet the requirements specified in 14 CFR 91.227. This system must be operational in certain airspaces after January 1, 2020 as specified by 14 CFR 91.225.

Section 5. PERFORMANCE

No change.

Section 6. WEIGHT AND BALANCE

See current weight and balance data.

Section 7. SYSTEM DESCRIPTION

The Garmin GTX 330 and GTX 3X5 Pilot's Guides, part numbers, and revisions listed below contain additional information regarding GTX system description, control, and function.

Title	Part Number	Revision
GTX 330 Pilot's Guide GTX 3X5 Pilot's Guide	190-00207-00 190-01499-00	Rev. G (or later) Rev. A (or later)

Pilot's Guides for interfaced displays, part numbers and revisions listed below, provide additional operating information for the Garmin GTX 33 and GTX 3X5R.

Title	Part Number	Revision
Garmin GTN 725/750 Pilot's Guide	190-01007-03	Rev. E (or later)
Garmin GTN 625/635/650 Pilot's Guide	190-01004-03	Rev. E (or later)
GNS 480 Pilot's Guide	190-00502-00	Rev. D (or later)
GTX 3X5 Series Transponder G1000 Pilot's Guide	190-01499-01	Rev. A (or later)
Garmin GI 275 Pilots's Guide	190-02246-01	Rev. F (or later)
Garmin GPS 175/GNC 355/GNX 375 Pilot's Guide	190-02488-01	Rev. B (or later)

7.1 GTX TIS Behavior

The TIS Standby/Operate controls for GTX 33/330 and GTX 335/335D units only function when the aircraft is airborne.

7.2 GTX 345R/345DR and G950/1000 No Bearing Traffic Alerts

No visual indication is provided for no bearing traffic alerts. Only an aural indication of the no bearing traffic alert is provided. If an aural alert for no bearing traffic has been previously issued, a "no bearing traffic clear" aural indication will be provided once all traffic alerts are resolved.

All aural alerts are inhibited below 500' AGL, therefore a "no bearing traffic clear" aural may not be heard in a landing or touch and go flight scenario.

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SAFETY TIPS

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SECTION 10

SAFETY TIPS

10.1 GENERAL

This section provides safety tips of particular value in the operation of the Cherokee Warrior II.

10.3 SAFETY TIPS

(a) Learn to trim for takeoff so that only a very light back pressure on the control wheel is required to lift the airplane off the ground.

(b) The best speed for takeoff is about 55 KIAS under normal conditions. Trying to pull the airplane off the ground at too low an airspeed decreases the controllability of the airplane in the event of engine failure.

(c) Flaps may be lowered at airspeeds up to 103 KIAS. To reduce flap operating loads, it is desirable to have the airplane at a slower speed before extending the flaps. The flap step will not support weight if the flaps are in any extended position. The flaps must be placed in the “UP” position before they will lock and support weight on the step.

(d) Before attempting to reset any circuit breaker, allow a two to five minute cooling off period.

(e) Before starting the engine, check that all radio switches, light switches and the pitot heat switch are in the off position so as not to create an overloaded condition when the starter is engaged.

(f) Anti-collision lights should not be operating when flying through cloud, fog or haze, since the reflected light can produce spatial disorientation. Strobe lights should not be used in close proximity to the ground such as during taxiing, takeoff or landing.

(g) The rudder pedals are suspended from a torque tube which extends across the fuselage. The pilot should become familiar with the proper positioning of his feet on the rudder pedals so as to avoid interference with the torque tube when moving the rudder pedals or operating the toe brakes.

(h) In an effort to avoid accidents, pilots should obtain and study the safety related information made available in FAA publications such as regulations, advisory circulars, Aviation News, AIM and safety aids.

(i) Prolonged slips or skids which result in excess of 2000 ft. of altitude loss, or other radical or extreme maneuvers which could cause uncovering of the fuel outlet must be avoided as fuel flow interruption may occur when tank being used is not full.

(j) Hand starting of the engine is not recommended, however, should hand starting of the engine be required, only experienced personnel should attempt this procedure. The magneto selector should be placed to "LEFT" during the starting procedures to reduce the probability of "kick back." Place the ignition switch to "BOTH" position after the engine has started.